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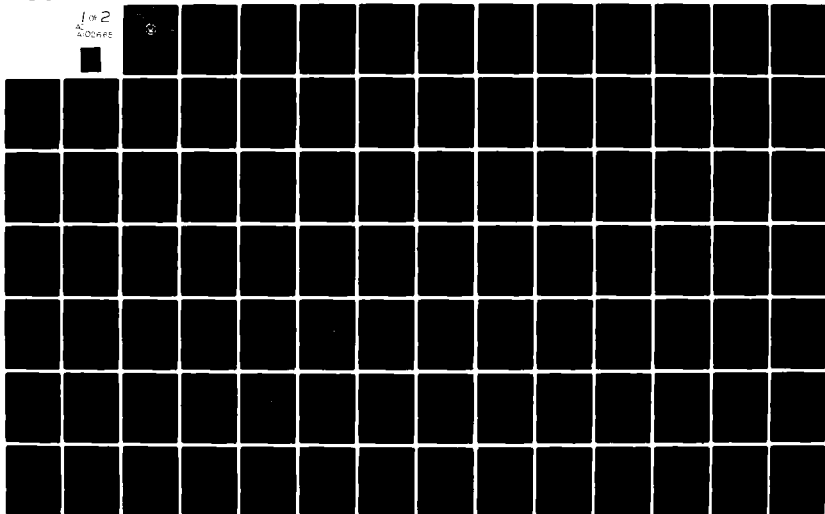
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# NAVAL POSTGRADUATE SCHOOL

Monterey, California



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## THESIS

AN ELASTIC-PLASTIC FINITE ELEMENT ANALYSIS  
OF NOTCHED ALUMINUM PANELS

by

Michael John Kaiser

March 1981

Thesis Advisor:

G. H. Lindsey

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Finite element, elastic and plastic analyses of various aluminum panels, containing holes and notches, were conducted for comparison with photoelastic experimental results. A FORTRAN IV program, ADINA (Automatic Dynamic Incremental Nonlinear Analysis), was used for both linear and nonlinear analyses. Mesh refinements were used for each panel and the monotonically convergent results were extrapolated using Richardson's method. Stresses were		

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Comparisons were made to the elastic, analytic series solution by Howland for a circular hole in a finite strip. The finite element results varied by less than one percent from Howland's solution. Handbook values for the elastic stress concentration factors of the geometries investigated differ from finite element results by less than one percent in all cases. The photoelastic works of Frocht were also compared where applicable. Stresses in the plastic range obtained from slip-line theory for a rigid-perfectly-plastic material show excellent correlation to a finite element analysis of such a material. Comparisons to elastic and plastic experimental data were made for the panels analyzed and show good correlation to finite element results.

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An Elastic-Plastic Finite Element Analysis  
of Notched Aluminum Panels

by

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Lieutenant Commander, United States Navy  
B.S., St. Cloud State University, 1969

Submitted in partial fulfillment of the  
requirements for the degree of

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## ABSTRACT

Finite element, elastic and plastic analyses of various aluminum panels, containing holes and notches, were conducted for comparison with photoelastic experimental results. A FORTRAN IV program, ADINA (Automatic Dynamic Incremental Nonlinear Analysis), was used for both linear and nonlinear analyses. Mesh refinements were used for each panel and the monotonically convergent results were extrapolated using Richardson's method. Stresses were locally smoothed from the Gauss integration points to the nodal points. Eight noded, isoparametric elements were used throughout. Modification to an ADINA preprocessor program, also coded in FORTRAN IV, was made for use with a VERSATEC plotter.

Comparisons were made to the elastic, analytic series solution by Howland for a circular hole in a finite strip. The finite element results varied by less than one percent from Howland's solution. Handbook values for the elastic stress concentration factors of the geometries investigated differ from finite element results by less than one percent in all cases. The photoelastic works of Frocht were also compared where applicable. Stresses in the plastic range obtained from slip-line theory for a rigid-perfectly-plastic material show excellent correlation to a finite element analysis of such a material. Comparisons to elastic and plastic experimental data were made for the panels analyzed and show good correlation to finite element results.

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## SYMBOLS AND ABBREVIATIONS

ADINA	Automatic Dynamic Incremental Nonlinear Analysis
b	Half width of strip
CPU	Central processor unit
E	Young's Modulus of Elasticity
$E_t$	Strain hardening tangent modulus
FEA	Finite element analysis
JCL	Job control language
$K_T$	Stress concentration factor referenced to reduced cross-section $\sigma/\sigma_n$
MVS	Multiple virtual storage
n	Ramberg-Osgood exponent
$O(h^m)$	Order of the discretization error
r	Radius of hole or notch
VM	Virtual machine
$\beta$	Ramberg-Osgood coefficient
$\epsilon$	General representation for strain
$\lambda$	Non-dimensional size parameter $\lambda=r/b$
$\nu$	Poisson's Ratio of transverse strain
$\sigma$	General representation for stress
$\sigma_\theta$	Principle stress in $\theta$ direction (hoop stress)
$\sigma_r$	Principle stress in radial direction
$\sigma_n$	Nominal stress in reduced cross-section
$\sigma_\infty$	Far-field stress
$\sigma_y$	Yield stress by 0.2% offset method

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## I. INTRODUCTION

Development of on-board fatigue monitoring systems for Naval aircraft have made it possible to record extensive structural loading data in flight. The strain gages used in such a system must be located away from stress concentration areas to prevent their fatigue; however, these areas are of the greatest interest in analyzing and predicting fatigue life of the structure. Understanding the relationship between nominal, far-field stresses and local stresses in critical areas thus becomes vitally important. Recent experimental investigations into the effect of uniform, far-field loads on stress concentration areas have been made at the Naval Postgraduate School (NPS) using photoelastic techniques [Refs. 1, 2 and 3]. These experiments involved loading 7075-T6 aluminum into both the elastic and plastic regions, as well as measurements of residual stresses resulting from plastic yielding.

Finite element analyses (FEA) of the aluminum panels used in the experiments of Stenstrom [Ref. 1] were conducted. The panels used in the experiments of Engle [Ref. 2] and Stuart [Ref. 3] have similar geometry. The finite element programs available at NPS were surveyed and ADINA [Ref. 4] was chosen for its proven ability to produce the nonlinear analyses required for plastic

yielding of aluminum. To provide increased accuracy, each panel was modeled using two meshes. The results obtained for the coarse and fine meshes were extrapolated to a final result using the Richardson extrapolation technique [Refs. 5 and 6].

Along with ADINA, a preprocessor program, PSAP1 [Ref. 7], was used to verify mesh connectivity prior to analysis by ADINA. PSAP1 provides a graphical output of the finite element mesh and was coded for the CALCOMP plotter installed at NPS prior to 1978. For this thesis PSAP1 was adapted for use with the VERSATEC plotter now installed at NPS.

The stress-strain material properties of the 7075-T6 aluminum actually used to make the panels had to be established to provide an accurate material model for use with ADINA. Material testing was conducted to establish the Young's Modulus ( $E$ ), Poisson's Ratio ( $\nu$ ), yield stress ( $\sigma_y$ ), strain hardening modulus ( $E_t$ ) and the Ramberg-Osgood coefficients  $\beta$  and  $n$ .

Comparisons were made to other works, in addition to the experiments conducted at NPS. The initial analysis involved a comparison of FEA to the results of Howland [Ref. 8], for a circular hole in a finite strip, to validate the methods used. A comparison of FEA to plane stress, slip-line theory, for rigid-perfectly-plastic material was also included as a validation for the plastic analyses.



## II. MATERIAL PROPERTY TESTING OF 7075-T6 ALUMINUM

The elastic and plastic material properties of the aluminum panels were established by tensile tests of uni-axial specimens made from the same mill run. The specimens were manufactured and tested according to current ASTM standards [Ref. 9]. MICRO-MEASUREMENTS, EA-13-125AD-120, precision strain gages with a temperature compensated bridge circuit were used on all specimens. Transverse gage sensitivity errors were corrected according to the manufacturer's recommendations [Ref. 10]. Critical cross-section measurements were made with a micrometer.

### A. TESTS FOR AXIAL LOADING

Initial tests of the two-gaged specimen, Fig. 1, in the MTS testing machine indicated a significant bending moment was being produced by the 30,000 lb GRIFF grips. To investigate this problem further, tests were conducted on both the MTS and RIEHLE test machines with a five-gaged specimen shown in Fig. 2. The results of these axial loading tests, shown in Table I, verified that the GRIFF grips on the MTS test machine do not give axial loading. An inspection of the gripped region on the specimen indicated that the jaws of the grip were not applying a uniformly distributed force and thereby induced a bending

moment by off-axis loading as shown in Fig. 2. The grips on the RIEHLE test machine gripped evenly and a uniform strain distribution resulted as seen in Table I.

## B. CHARACTERISTICS OF 7075-T6 ALUMINUM PANELS

The following characteristic properties were determined from the four specimens tested.

### 1. Young's Modulus (E)

Tests were conducted using the specimen shown in Fig. 3 on the MTS test machine with 10,000 lb INSTRON grips, which gripped the specimen evenly. The results of testing three specimens are shown in Tables II to IV. Linear regression in the elastic range of all the test results determined a Young's Modulus of  $10.12 \times 10^6$  psi, with a correlation coefficient of 0.9996.

### 2. Poisson's Ratio ( $\nu$ )

Tests were conducted using the specimen shown in Fig. 1 on the RIEHLE test machine with 10,000 lb RIEHLE grips. The results are tabulated in Table V. Linear regression of these results in the elastic region determined Poisson's Ratio to be 0.3256 with a correlation coefficient of 0.99996.

### 3. Yield Stress and Strain Hardening Modulus

These values, required for ADINA's bi-linear material model, were determined graphically using the data from Tables III and IV. Plastic region data in

Table II is not reliable because of excessive creep encountered during that test.

0.2% offset yield stress,  $\sigma_y = 76,000$  psi

strain hardening modulus,  $E_t = 566,000$  psi

The graphical fit of these values to the test data can be seen in Fig. 4.

#### 4. Ramberg-Osgood Coefficients

The Ramberg-Osgood equation for elastic-plastic stress-strain characterization is given by:

$$\epsilon = \frac{\sigma}{E} + \beta \left( \frac{\sigma}{E} \right)^n \quad (1)$$

where:

$\epsilon$  = strain

$\sigma$  = stress

$E$  = Young's modulus.

The  $\beta$  and  $n$  coefficients were determined graphically from the data of Table IV, by the method given by Rivello [Ref. 11]. The data in Table IV gave the following values which are the best fit to the combined test data

$$\beta = 1.479 \times 10^{43}$$

$$n = 21.58$$

The graphical fit of these values, in Eq. (1), with the test data is also shown in Fig. 4.

### III. MODIFICATION TO GRAPHICAL PREPROCESSOR

The use of a graphical preprocessor program, such as PSAP1, is vital in detecting mesh errors that may otherwise go unnoticed. Establishing the proper node locations and element geometry prior to analysis for a complex code such as ADINA is of utmost importance.

#### A. PSAP1 MODIFICATIONS

The program PSAP1 was originally coded in FORTRAN IV for use on the NPS IBM 360/370 installation with the CALCOMP Model 765 drum plotter. The CALCOMP system installed at NPS used the +Y axis as the unlimited plotting axis, see Fig. 5. The entire plotting logic in PSAP1 uses this orientation of axes to allow multiple plots in a continuous strip. With the VERSATEC Model 8222A electrostatic plotter now installed at NPS, the +X axis becomes the unlimited plotting axis, shown in Fig. 5. To avoid an extensive recoding of PSAP1 for use with the VERSATEC plotter, a simple coordinate transformation of the plot was made in a limited number of short subroutines. First, all installation dependent plotting calls used in PSAP1 were identified. These involved seven plotter functions for which new subroutines were coded.

<u>Function</u>	<u>New Subroutine</u>
Initialize Plotter	CALCMP
Move Plotter Pen	CALPLT
Letter on Plot	NOTATE
Number on Plot	CALNUM
Determine Current Pen Location	CALWH
Draw a Line	CALINE
Stop Plotter	PSTOP

The subroutines listed above merely rotates the plot to coincide with the VERSATEC axis orientation and retain all features originally in PSAP1. Since all plotter hardware code is now isolated in these seven subroutines, future adaptations to other plotting systems is simplified. To provide documentation of this update to PSAP1, a complete listing of the new program is provided in Appendix D.

#### B. USE OF PSAP1

Previous use of PSAP1 on the IBM 360 system necessitated use of a load module since PSAP1 took over one minute of CPU time to compile. With the new IBM 3033 system compilation requires eight seconds; however, use of a load module or disk stored source code is still recommended since PSAP1 contains roughly 2,500 lines of code. Appendix A contains sample JCL to use PSAP1 on the IBM 3033 MVS system. With minimal effort PSAP1 could also be set up

for use on the IBM VM/370 system. The user's manual for PSAP1 is in Ref. 7. In addition to a mesh plot, PSAP1 provides a listing of the node coordinates, element connectivity and several key input values used in execution of ADINA. This information provides a useful check of the input data.

#### IV. FINITE ELEMENT ANALYSIS (FEA)

##### A. DESCRIPTION OF MESHES USED

###### 1. Length to Width Ratios and Boundary Conditions

Initial finite element models of specimens had length to width ratios near one, as used by Garske [Ref. 12] for his FEA, but they did not provide the desired uniform distribution at the loading boundary. Specimen length to width ratios of 3-5 were used by Armen, Pifko and Levine [Ref. 13] in their FEA and by Stenstrom [Ref. 1] in his photoelastic experiments. The criteria established to determine uniform boundary stress distribution was uniformity in nodal displacements along the loaded edge as discussed by Segerlind [Ref. 14]. In the models used for FEA in this thesis, nodal displacements were uniform to within 0.1%, and the resulting stress distribution was uniform axially to within 0.1% at the panel ends. In all cases two-dimensional, eight noded, isoparametric elements were used. These higher order elements cannot be loaded in an "intuitive" manner as discussed by Zienkiewicz [Ref. 15, p. 223]. Figure 6 shows the nodal loading required to obtain a uniform surface load.

## 2. Element Meshes

Two meshes were developed for each panel analyzed. A reasonable effort was made to keep element corner angles as close to  $90^\circ$  as possible to reduce the effect of element distortion discussed by Hopkins and Gifford [Ref. 16]. All meshes modeled a quarter of the actual panel by using the two axes of symmetry as is common practice in FEA. The step from course to fine element meshes was made so that each element in the course mesh was subdivided into four smaller elements of the same type. Such a mesh subdivision can be expected to give monotonic convergence of results, Cook [Ref. 17], and allow extrapolation to results of an infinitely fine mesh. Figures 7 through 13 illustrate the element meshes used in this analysis as plotted by PSAP1.

## B. COMPUTATIONAL PROCEDURES

### 1. Using ADINA

Once the mesh has been developed, input data is prepared in accordance with the ADINA user's manual [Ref. 4]. This same set of data is then used as input for PSAP1 to check for errors and provide a graphical display of the element mesh. After preprocessing by PSAP1, the data is entered into ADINA for analysis. In the case of linear analysis, two types of stress output may be specified, nodal point or Gauss integration point. Nodal point output



can be computed for up to eight node point stresses for each element. Since 2x2 Gauss integration was used, four Gauss point stresses were computed for each element. The 2x2 Gauss integration is recognized as the most efficient integration order for this type of analysis [Ref. 15, p. 284]. The linear analysis used an isotropic linear elastic material model (MODEL "1" in Ref. 4) which required input of E and  $\nu$  material properties. The nonlinear analysis allows only Gauss point stress outputs and uses a bilinear elastic-plastic material model, with von Mises yield condition and isotropic strain hardening (MODEL "8" in Ref. 4).

For static analyses ADINA uses a time function method to apply loads in steps. Linear analysis loading was accomplished in a single step to a nominal value of 3,000 lbs load. Nonlinear analysis loads were applied in ten steps to a maximum value, matching the experimental loads, and then unloaded to zero in ten steps to obtain residual stresses. The stress output from ADINA is a listing of nodal or Gauss point stresses for each element. Since the only area of interest in this analysis was the distribution of stresses along the reduced cross-section, no large post-processing program was developed or used. All final computations using ADINA output data were accomplished on a HEWLETT-PACKARD 9830A calculator, using short programs coded in BASIC. If more extensive stress

distribution information were desired, some form of automated post-processing would be necessary to reduce the computational workload. At a minimum, nodal stress outputs by ADINA must be averaged to obtain unique values of stress at nodes shared by more than one element.

## 2. Richardson Extrapolation

The use of course and fine meshes allows extrapolation to an infinitely fine mesh as discussed earlier. Richardson extrapolation [Ref. 5] was used in this analysis where:

$$\sigma_{\text{extrap}} = \frac{\sigma_C (h_F)^m - \sigma_F (h_C)^m}{h_F^m - h_C^m} \quad (2)$$

where

$\sigma_{\text{extrap}}$  = extrapolated solution

$\sigma_C$  = solution obtained with  $h=h_C$

$\sigma_F$  = solution obtained with  $h=h_F$

$h_C$  = linear dimension of course element

$h_F$  = linear dimension of fine element

$m$  = 2 (for this analysis)

The exponent  $m$  is determined by the order of the discretization error  $O(h^m)$ . Since  $h$  represents the length of an element the element area is represented  $h^2$ . In a two dimensional problem such as this  $O(h^m)$  is of the order of  $h^2$ , the area of an element. In the mesh refinement scheme

used  $h_F = \frac{1}{2} h_C$  or  $\frac{h_F}{h_C} = \frac{1}{2}$ . Equation (2) can be rewritten

$$\sigma_{\text{extrap}} = \frac{\sigma_C \left(\frac{h_F}{h_C}\right)^2 - \sigma_F \left(\frac{h_C}{h_C}\right)^2}{\left(\frac{h_F}{h_C}\right)^2 - \left(\frac{h_C}{h_C}\right)^2} \quad (3)$$

thus

$$\sigma_{\text{extrap}} = \frac{\sigma_F - \frac{1}{4} \sigma_C}{\frac{3}{4}} \quad (4)$$

Equation (4) then becomes the relation to obtain extrapolated stresses from coarse and fine mesh results in a two dimensional analysis. Better extrapolations can be obtained by using three or more refined meshes, but, the computational effort increases significantly.

### 3. Optimal Stress Locations and Local Smoothing

It is generally accepted that the most accurate sampling points for stresses are the Gauss integration points within the element [Ref. 15, p. 281, and Ref 18]. In this analysis, the nodal points are of the greatest interest; thus a technique of local smoothing must be applied to the integration point stresses to obtain nodal stresses as reported by Hinton and Campbell [Ref. 19]. The formulation of this local smoothing technique for ADINA elements is developed in Appendix B. The nodal values obtained must then be averaged if shared by two or more elements.

#### 4. Computational Times

Because of the extremely large size of ADINA (about 17,000 lines of code in the NPS version) and the out of core solver, it does not adapt well to time sharing systems. Using the IBM 360 system at NPS, ADINA required 31 user defined overlays to create a manageable load module in about 30 minutes of CPU time. With the new IBM 3033 MVS system at NPS, ADINA is compiled without overlays in about one minute. When using a load module, the program execution took less than 2 minutes CPU time on the IBM 3033.

In addition to ADINA, the preprocessor (PSAP1) and post-processing techniques involve considerable time and effort. Figure 14 is a flow chart of the computational procedure used in this analysis. An example of the JCL to use ADINA in load module form on the IBM 3033 MVS system with use of the mass storage facilities is shown in Appendix C.

## V. RESULTS OF ANALYSIS

### A. CIRCULAR HOLES IN LINEAR MATERIAL

The FEA results for a circular hole in a finite width strip were used to validate the elastic computational procedure discussed earlier. The results of Howland [Ref. 8] were compared to both the Gauss point smoothed results and the nodal output results in Fig. 15. The stress concentration factor  $\sigma/\sigma_\infty$  is referenced to the far-field stress. The smoothed results give the best match to the results of Howland at the edge of the hole, and the only significant variation between the two FEA methods occurs within the first 0.25 inches from the edge. In order to obtain the 0.25 inch stress value for the coarse mesh, in the Gauss point smoothed result, a midside node value had to be obtained by the averaging method discussed in Appendix B. The linear distribution of smoothed stresses along the sides of the element, [Ref. 19], appears to produce a less accurate result in this area of extreme stress gradient, when compared to ADINA's nodal output result. This tendency was noted in all cases; however, the peak stress values from smoothed results consistently gave better correlation with other investigators [Ref. 20].

A circular hole with  $\lambda=0.25$  was also analyzed and compared to the experimental results of Stenstrom [Ref. 1]

along with an interpolation of Howland's results. The  $\sigma_\theta$  experimental data correlates well with the FEA results; however, the  $\sigma_r$  experimental data shows significant variation between 0.125 and 0.375 inches from the edge of the hole, as seen in Fig. 16.

#### B. OPPOSITE U NOTCHES IN LINEAR MATERIAL

The results for linear analysis are presented in non-dimensional stress concentration form; however, the normalizing stress changes. For U notches  $K_T$  is the stress concentration factor referenced to the theoretical, nominal stress ( $\sigma_n$ ) in the reduced cross-section where  $\sigma_n = \text{Load/Area of Reduced Cross-Section}$ .

##### 1. Shallow Notch Panel

The FEA results plotted with the experimental data of Stenstrom are shown in Fig. 17. Once again both FEA results are shown and the variation for the two methods occur within 0.25 inches from the notch edge; however, there was less variation than was seen in the circular hole analyses.

For this panel the experimental data appear to be uniformly below the FEA results for  $\sigma_\theta$ . The  $\sigma_r$  data shows significant variation at the 0.625 inch point but follows the proper trend within 0.5 inches from the notch edge. According to data collected by Peterson [Ref. 20], this notch geometry should yield a maximum  $K_T = 2.74$ . The

Gauss point smoothed results matched this value exactly. Stuart [Ref. 3] reported a  $K_T = 2.69$  for this same notch geometry, with a standard deviation of 0.187 in the 14 samples he measured by photoelastic methods. Early photoelastic work by Frocht [Ref. 21] determined  $K_T = 2.7$  for this notch geometry. The FEA results appear to be in good agreement with other investigators, for this notch geometry.

## 2. Deep Notch Panel

Results of the FEA for a deep notch were plotted with Stenstrom's experimental data in Fig. 18. The results of the two FEA methods again diverge within 0.5 inches from the notch edge. FEA stress values at the 0.25 inch point have spread farther apart in this case since the stress gradient is very severe at that point. The experimental data correlates well for both  $\sigma_\theta$  and  $\sigma_r$ ; however, the maximum experimental  $\sigma_\theta$  is considerably lower with a  $K_T = 3.83$ . Results reported by Stuart for this notch geometry was a  $K_T = 4.05$  with a standard deviation of 0.219 for 14 specimens measured by photoelastic methods. Frocht [Ref. 22] reported a photoelastic  $K_T = 3.9$  for this notch geometry, but concluded that the result was 5-10% low, giving a corrected range of  $K_T$  from 4.1 to 4.3. The Gauss smoothed FEA result gave a  $K_T = 4.24$  which compares well to an empirical relation given by Peterson [Ref. 20] for  $r/d < 0.25$ .

$$K_T = \left(1 - \frac{2t}{D}\right) (0.78 + 2.243\sqrt{t/r}) \left[ 0.993 + 0.18 \frac{2t}{D} - 1.06 \left(\frac{2t}{D}\right)^2 + 1.71 \left(\frac{2t}{D}\right)^3 \right] \quad (5)$$

where

$t$  = notch depth (3.9375)

$r$  = notch radius (0.625)

$d$  = minimum width (15.625)

$D$  = maximum width (23.5)

Inserting the values above into Eq. (5) gives a  $K_T = 4.26$ , which is 1/2% above the FEA result. Other FEA results reported by Armen, Pifko and Levin [Ref. 13] and Griffis [Ref. 23], using linear strain triangle (LST) elements, produced  $K_T$  values within 5% of those produced by use of Eq. (5). The rectangular elements used in this analysis are known to give better results than LST elements as noted by Clough [Ref. 24]. It is clear that the FEA results obtained for this notch are in good agreement with other works.

#### C. OPPOSITE U NOTCHES IN NONLINEAR MATERIAL

The analysis for loading into the plastic region of the 7075-T6 aluminum was made using the bilinear material model discussed earlier. The loads used were selected to match those used in the experiments of Stenstrom; thus allowing direct comparison. The strains obtained in those experiments were used to solve for stresses by use of the Prandtl-Reuss plastic flow equations.



### 1. Shallow Notch Panel

The results of FEA for the three load cases, 60,000, 65,000 and 70,000 lbs are presented along with the experimental results in Figs. 19 through 21. The  $\sigma_\theta$  results compare well although no trend for peak  $\sigma_\theta$  stress away from the notch edge is shown in the experimental data. In all cases the FEA determined the peak  $\sigma_\theta$  stress to occur near the yield boundary, and the gradient of the  $\sigma_\theta$  stress to fall off dramatically in the plastic zone. This characteristic behavior of the  $\sigma_\theta$  stress was reported by other investigators [Refs. 13 and 23] using FEA on 2024-T3 aluminum. Plane elastic-plastic stress distributions reported by Frocht [Ref. 25] show similar trends. The experimental data also shows a marked change in the gradient of  $\sigma_\theta$  stress within the plastic region. The growth of this plastic region is approximated using the FEA results for this notch in Figs. 22 through 24.

Experimental data for the  $\sigma_r$  stress distribution matches the FEA results closely except at the notch edge where the measured  $\sigma_r$  does not go to zero as it should. The characteristic peak value of  $\sigma_r$  near the plastic boundary as seen in the FEA results is also shown by the test data.

The FEA residual stress computed upon unloading from the three load cases are shown in Figs. 25 through 30. The characteristic distributions of the  $\sigma_\theta$  residual stress

agrees with those reported by others [Refs. 25, 26 and 27]. The experimental residual stress distributions reported by Stenstrom show similar trends but significant variations when compared to the FEA results.

## 2. Deep Notch Panel

Three load cases were computed to plastic loading levels; however, only limited experimental results are available for this notch as seen in Fig. 31. The 30,000 lb load is just at the onset of yield in the notch root area. Limited residual experimental data [Ref. 3] was available for comparison in Figs. 32 and 33 which are plots of the residual  $\sigma_{\theta}$  and  $\sigma_r$  stress distributions as a result of the three loading cases. The 30,000 lb load has caused yielding in a small region at the root of the notch as seen in Fig. 34. Figures 35 and 36 illustrate that the plastic zone does not grow to the extent it did in the shallow notch. The stress gradients are very severe in the deep notch and as a result the 0.25 inch sampling points used in the FEA may be useful in only showing gross trends close to the notch edge. The trends appear to be much the same as in the shallow notch, with the peak  $\sigma_{\theta}$  and  $\sigma_r$  stresses occurring near the yield boundary. The yield boundaries shown in Fig. 31 are approximations based on qualitative analysis of the finite element results. The  $\sigma_{\theta}$  experimental data for the 30,000 lb load case correlates well with the peak stress, again

appearing low as it did in the linear analysis. The residual  $\sigma_\theta$  stress distribution in Fig. 32 follow much the same trends as seen in the shallow notch but with much higher gradients within the first 0.5 inches from the notch. The FEA results for the residual  $\sigma_r$  stresses shown in Fig. 33 indicate a limitation in the element size used since the  $\sigma_r$  value of the notch edge does not return to zero as it should. Because of this problem the data may be questionable for showing proper trends in the first 0.5 inch from the notch edge.

### 3. Rigid-Perfectly Plastic Panel

A stress distribution for the theoretical material used in slip-line theory was desired. By using ADINA's bilinear material model such a material could be approximated reasonably well. For this analysis a Young's Modulus (E) of  $10^{26}$  psi was used to model perfect rigidity. The strain hardening modulus ( $E_t$ ) was set to zero to model perfect plasticity. Poisson's ratio ( $\nu$ ) was 0.4999999, as close to 0.5 as the computer would allow. The  $E = 10^{26}$  also represents a computer limit in approaching an infinitely large E. Figure 37 illustrates the results obtained and compares them to a slip-line solution. The results are normalized to the arbitrary 73,000 psi yield stress used in the analysis. The  $\sigma_\theta$  values obtained agree exactly with slip-line theory. Conversely the  $\sigma_r$  results do not reflect the same values as slip-line theory, but do show

a similar trend. The growth of the plastic zone obtained is shown in Figs. 38 through 40.

## VI. CONCLUSIONS AND RECOMMENDATIONS

The results obtained from the FEA have proven useful in determining the validity of experimental data gathered by photoelastic techniques. The FEA results varied by less than 1% when compared to published analytical results and handbook values. Experimental data from Stenstrom's photoelastic work correlated well with the FEA results. The primary exception was the residual stress experimental values, which varied significantly from the FEA results. The variation may be in transforming the residual strains measured photoelastically into stresses for comparison with the FEA results, since ADINA only provides a stress output. Limitations of ADINA's bilinear material model, initially considered severe, do not appear to have hampered this investigation. A possible exception is the residual analyses where the transition region from elastic to plastic strains becomes especially important. The Gauss point smoothed results gave the best correlations at the edge singularities in all cases; however, due to the limitations noted at 0.25 inches from the edge, the results at that specific point may not be as accurate for this method. The nodal output results gave consistently higher stress values at the edge singularities. Because

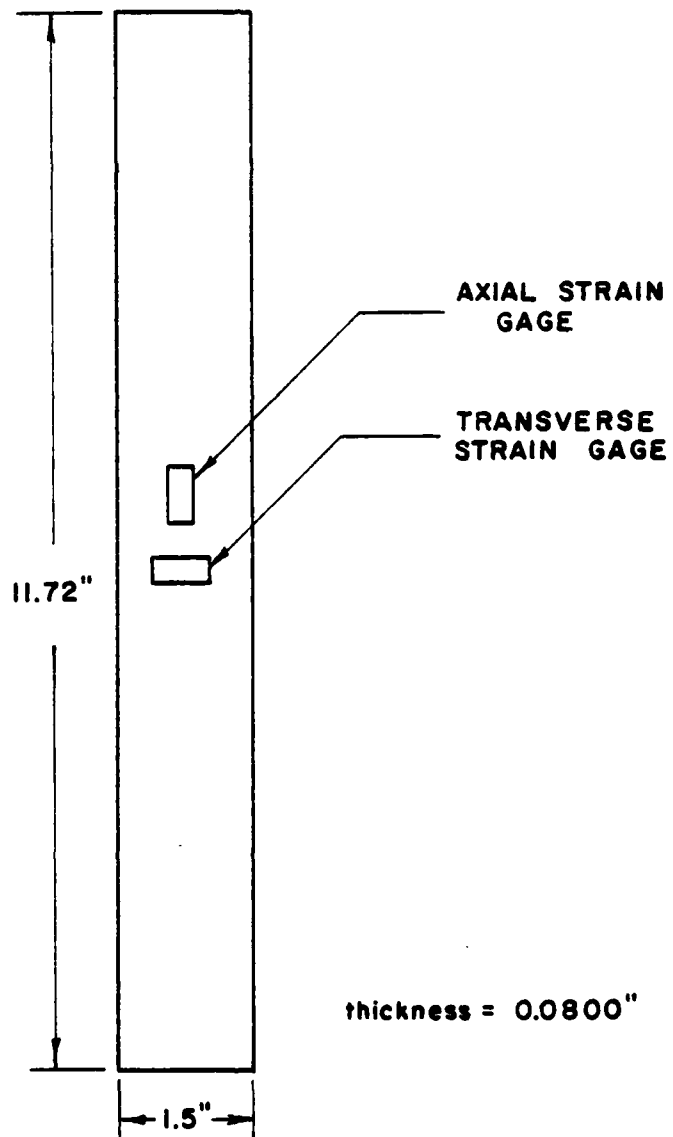
of the severe stress gradients near the deep notch analyzed the use of a finer element mesh near the notch would probably produce better results.

The effort involved in developing two meshes such as those used in this thesis is considerable. An automatic mesh generation capability would reduce the workload and allow experimentation with several types of element meshes.

ADINA proved to be a useful and powerful program, as expected, but something simpler and less awkward to use may be all that is required for two dimensional analysis. Such a system is already in use at NPS but does not offer non-linear capabilities. If use of ADINA is to be continued in this type of investigation, a post-processing program should be adapted. There are programs available to post-process ADINA data at NPS [Refs. 28 and 29] but they would require modifications to work with two dimensional analyses and the VERSATEC plotter.

Standardized material property testing would ease the inevitable task of obtaining basic material properties for use in analysis or experiments. Some form of automatic data collection with use of the MTS testing machine would allow testing of a larger sample population and provide statistically more accurate information.

FIGURE 1  
2 GAGE SPECIMEN



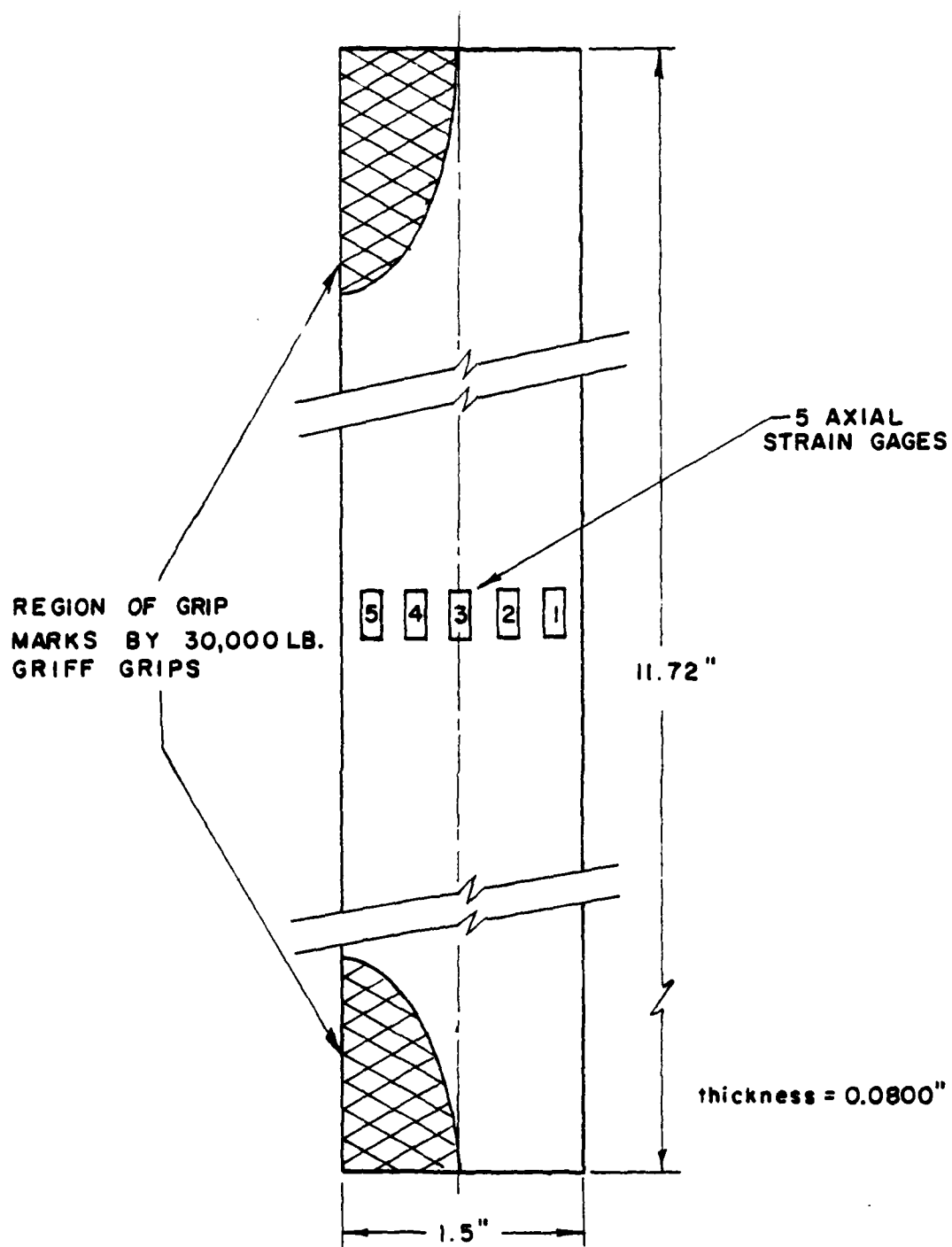


FIGURE 2

5 GAGE SPECIMEN



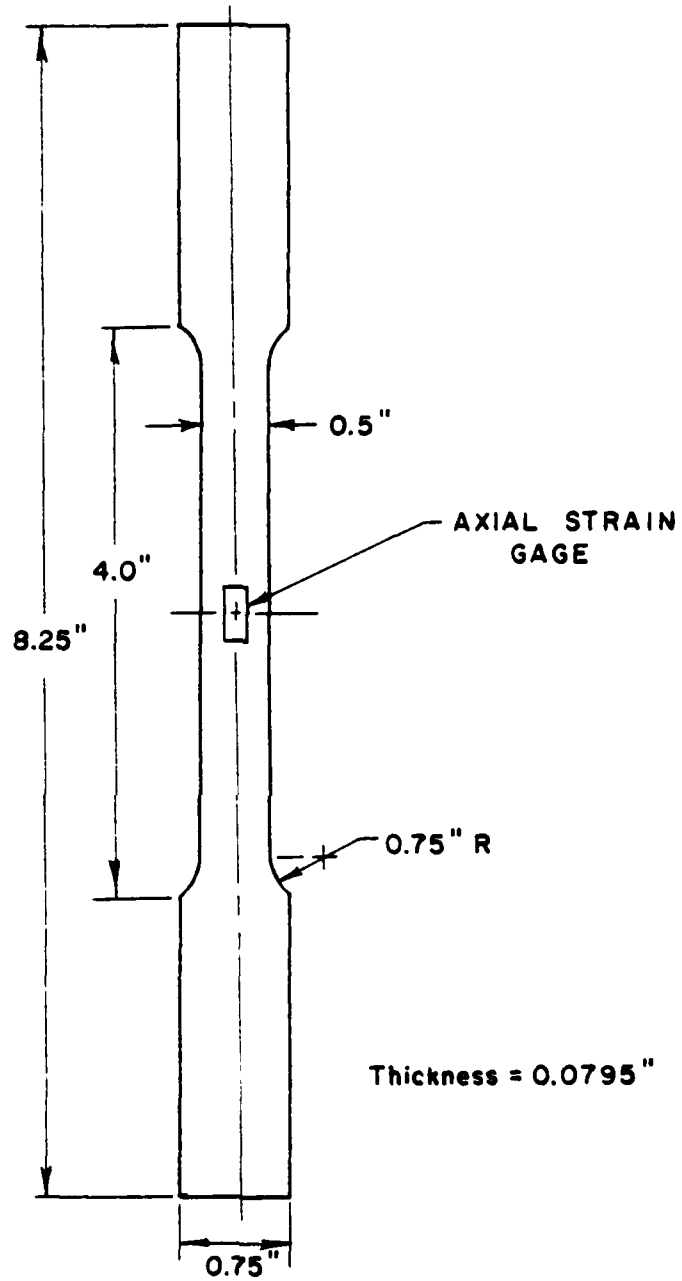
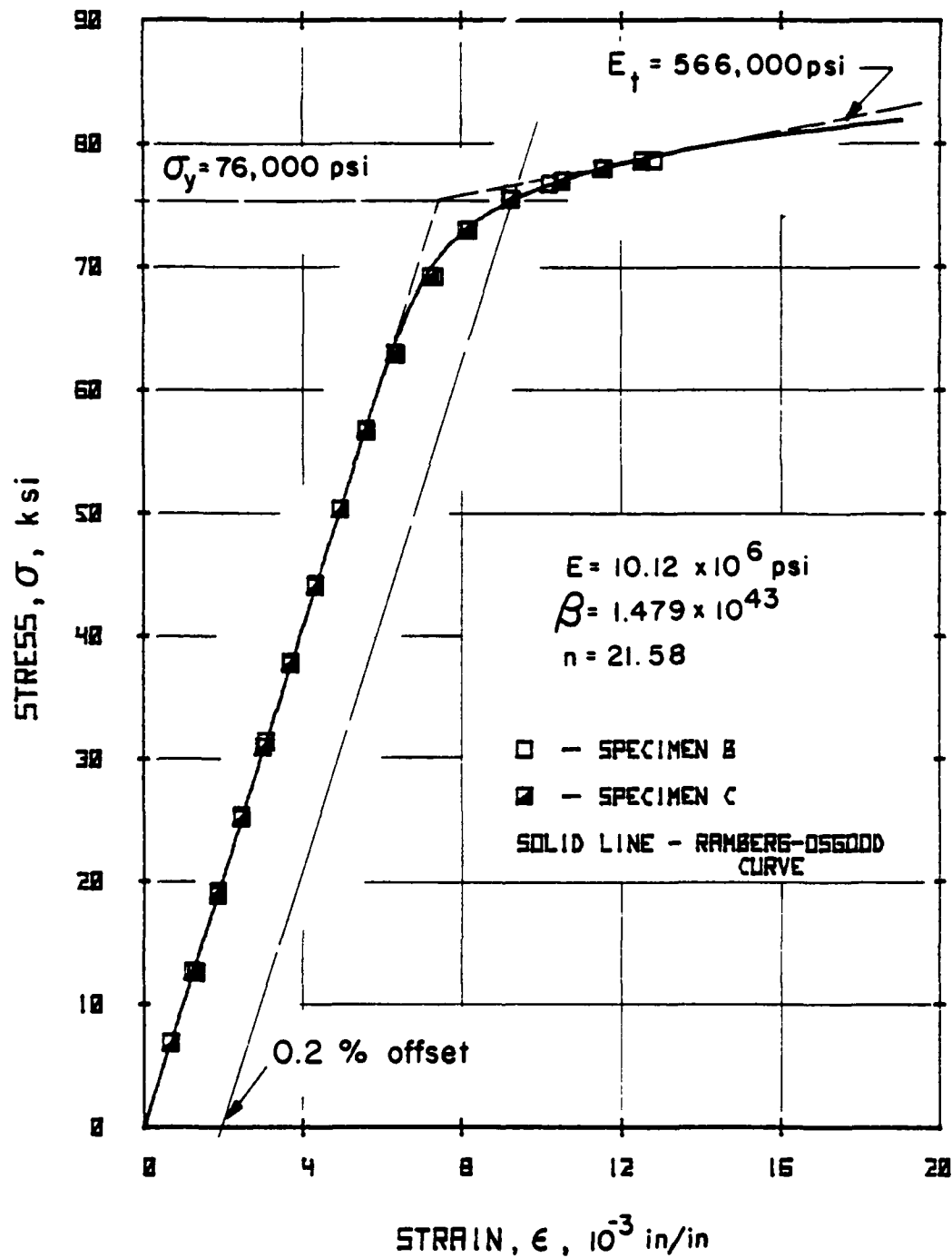


FIGURE 3

1 GAGE SPECIMEN

FIGURE 4

7075-T6 ALUMINUM STRESS-STRAIN CURVE



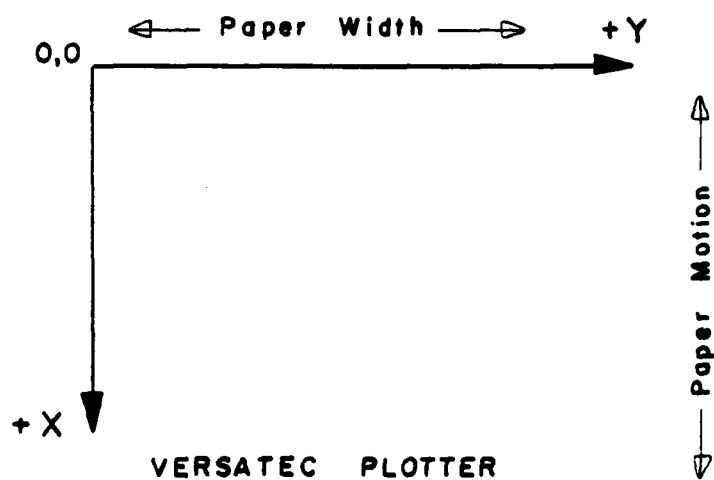
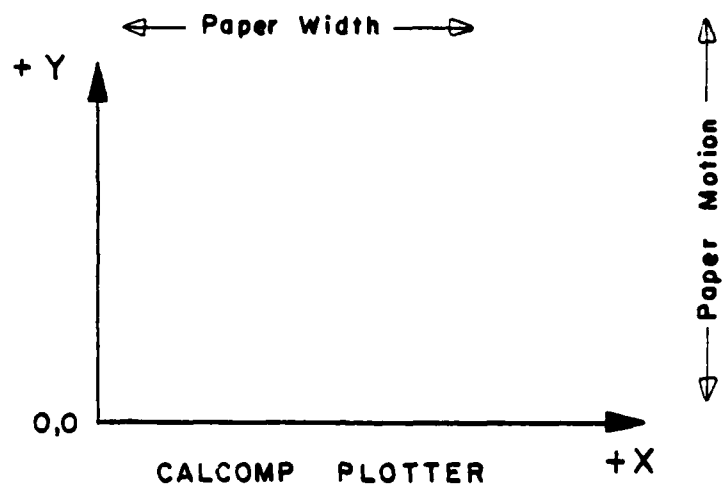


FIGURE 5

CALCOMP AND VERSATEC PLOTTER AXES

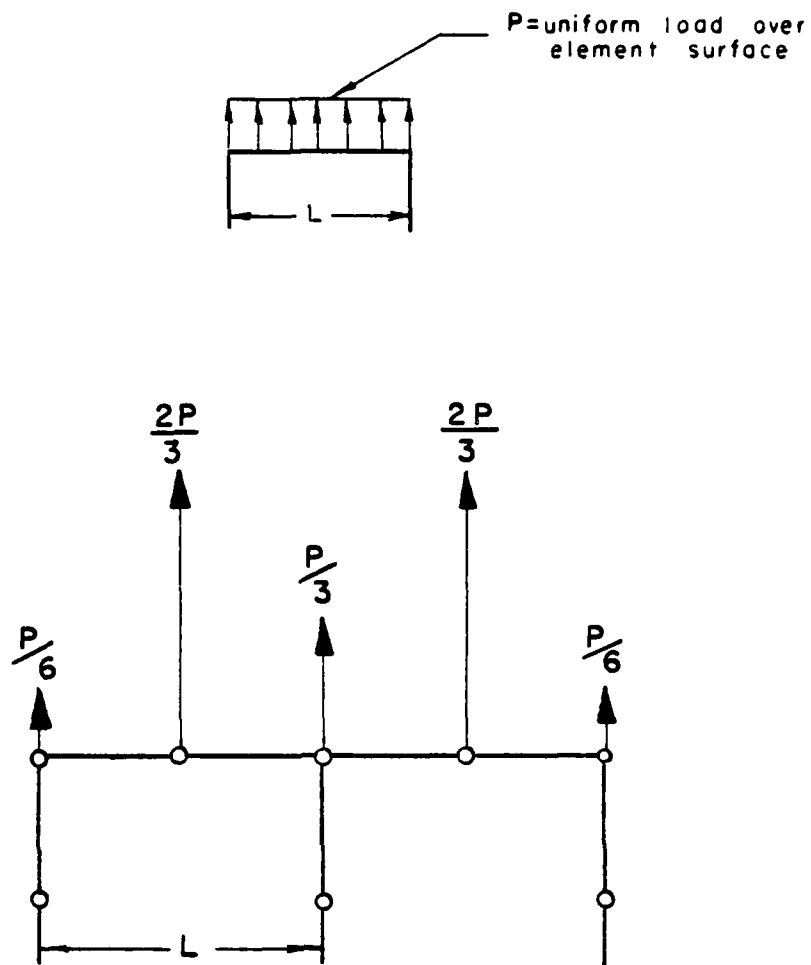


FIGURE 6

NODAL LOADING DIAGRAM

28 ELEMENTS (ISOPARAMETRIC)

111 NODES

192 DEGREES OF FREEDOM

DIMENSIONS

$\lambda$	W	L	RADIUS
0.2	5"	25"	1"
0.25	4.0625"	20"	1"

$$\lambda = \frac{\text{RADIUS}}{W}$$

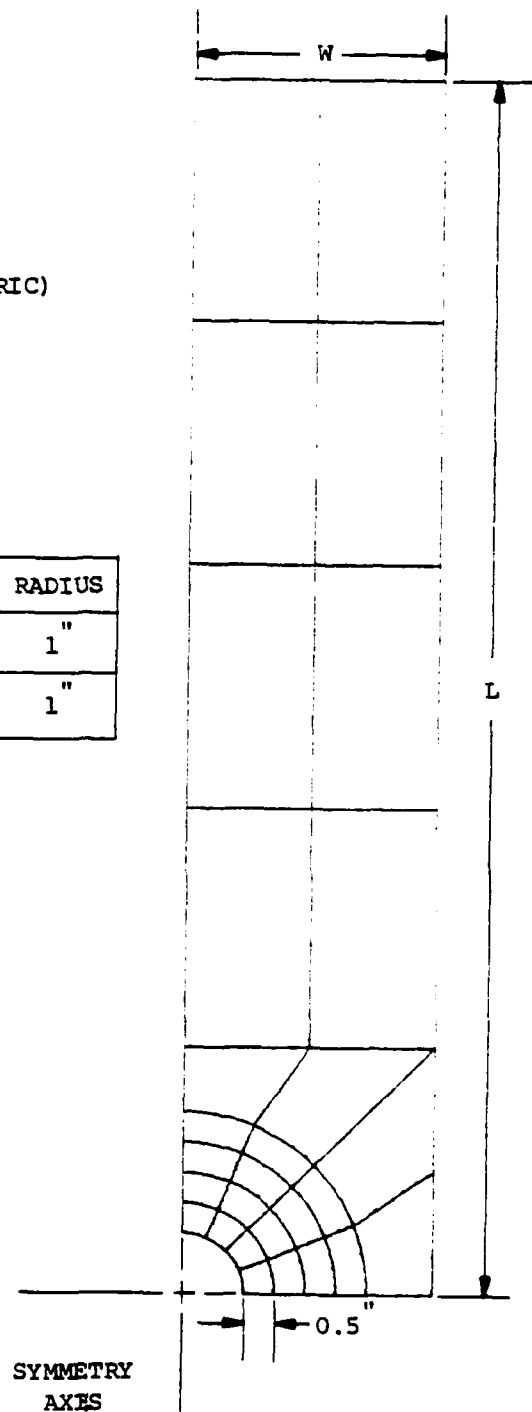


FIGURE 7

COURSE MESH FOR CIRCULAR HOLES

A SUBDIVISION OF COURSE MESH

112 ELEMENTS (ISOPARAMETRIC)

389 NODES

720 DEGREES OF FREEDOM

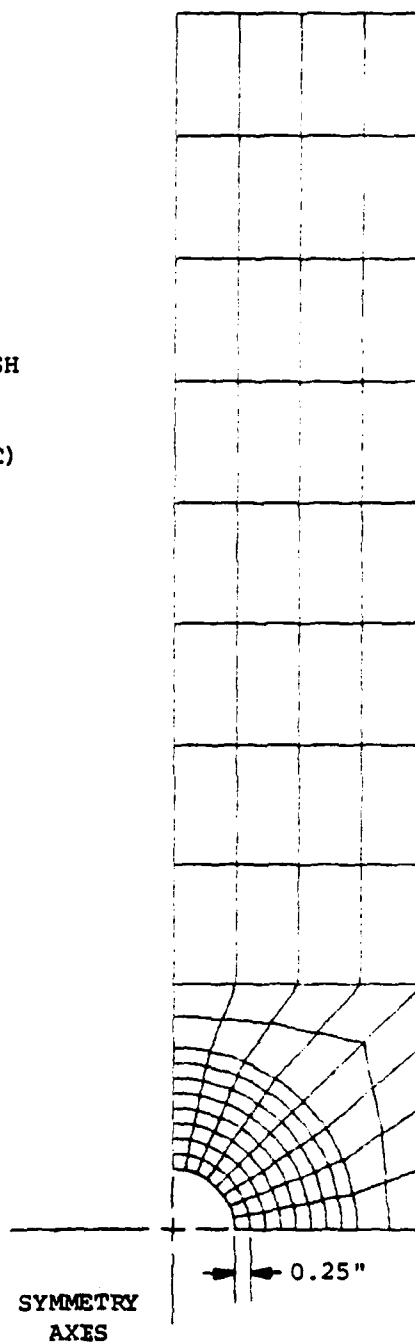


FIGURE 8

FINE MESH FOR CIRCULAR HOLES

FIGURE 9

COURSE MESH FOR SHALLOW NOTCH

60 ELEMENTS  
(ISOPARAMETRIC)  
219 NODES  
404 DEGREES OF  
FREEDOM

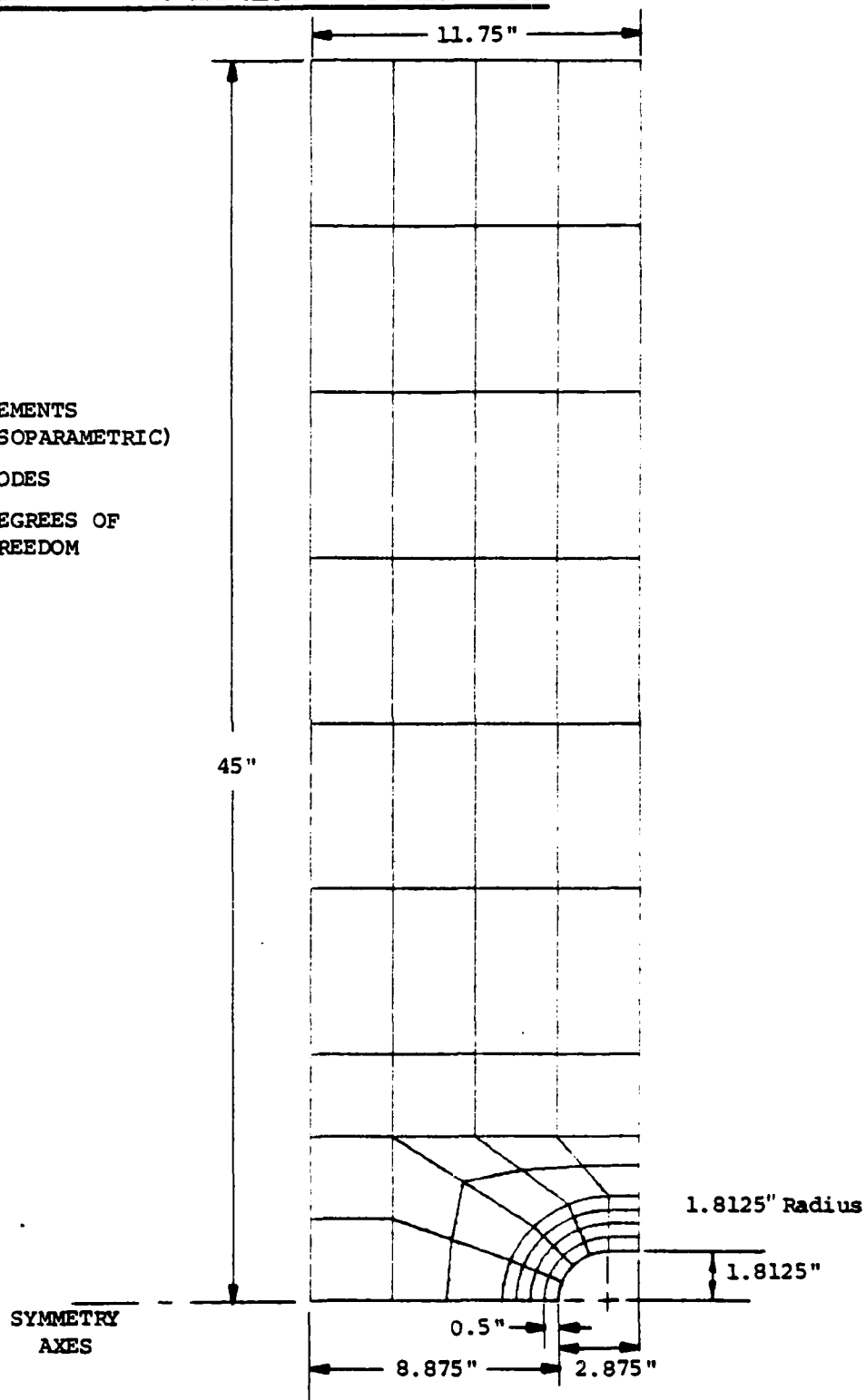


FIGURE 10 FINE MESH FOR SHALLOW NOTCH

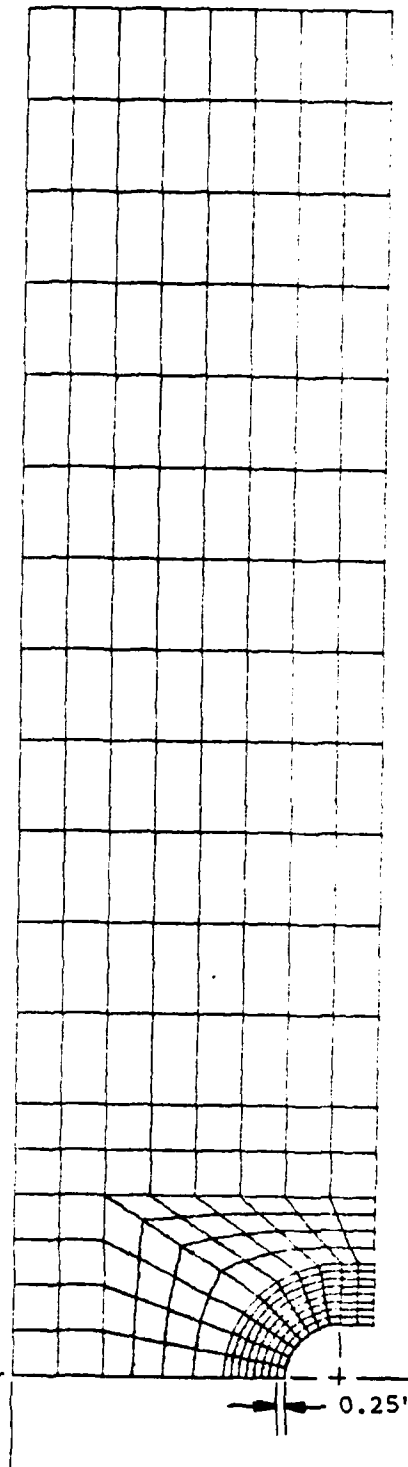
SUBDIVISION OF  
COURSE MESH

240 ELEMENTS  
(ISOPARAMETRIC)

797 NODES

1528 DEGREES OF  
FREEDOM

SYMMETRY  
AXES





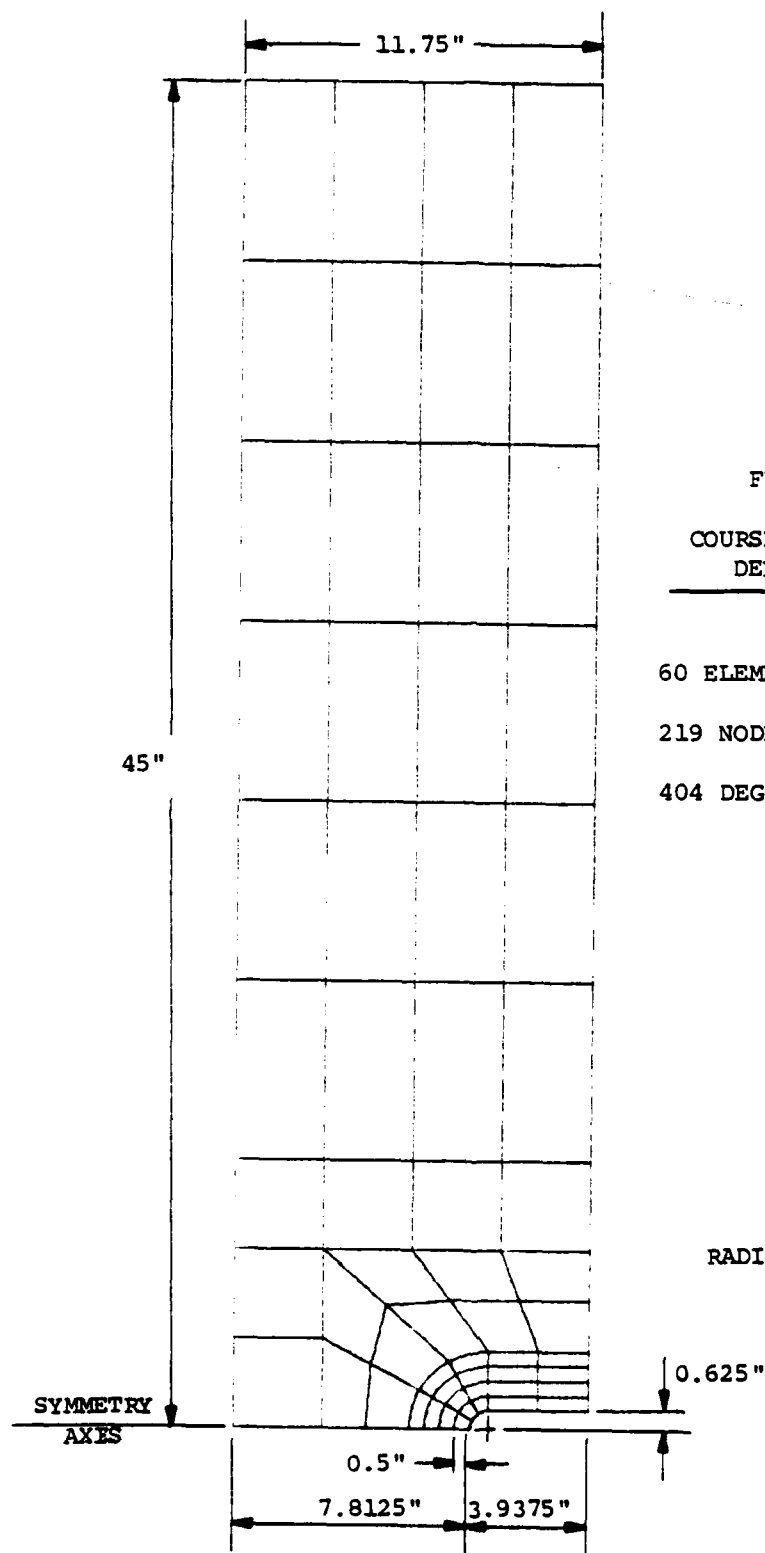


FIGURE 11

COURSE MESH FOR  
DEEP NOTCH

60 ELEMENTS (ISOPARAMETRIC)

219 NODES

404 DEGREES OF FREEDOM

RADIUS OF NOTCH= 0.625"

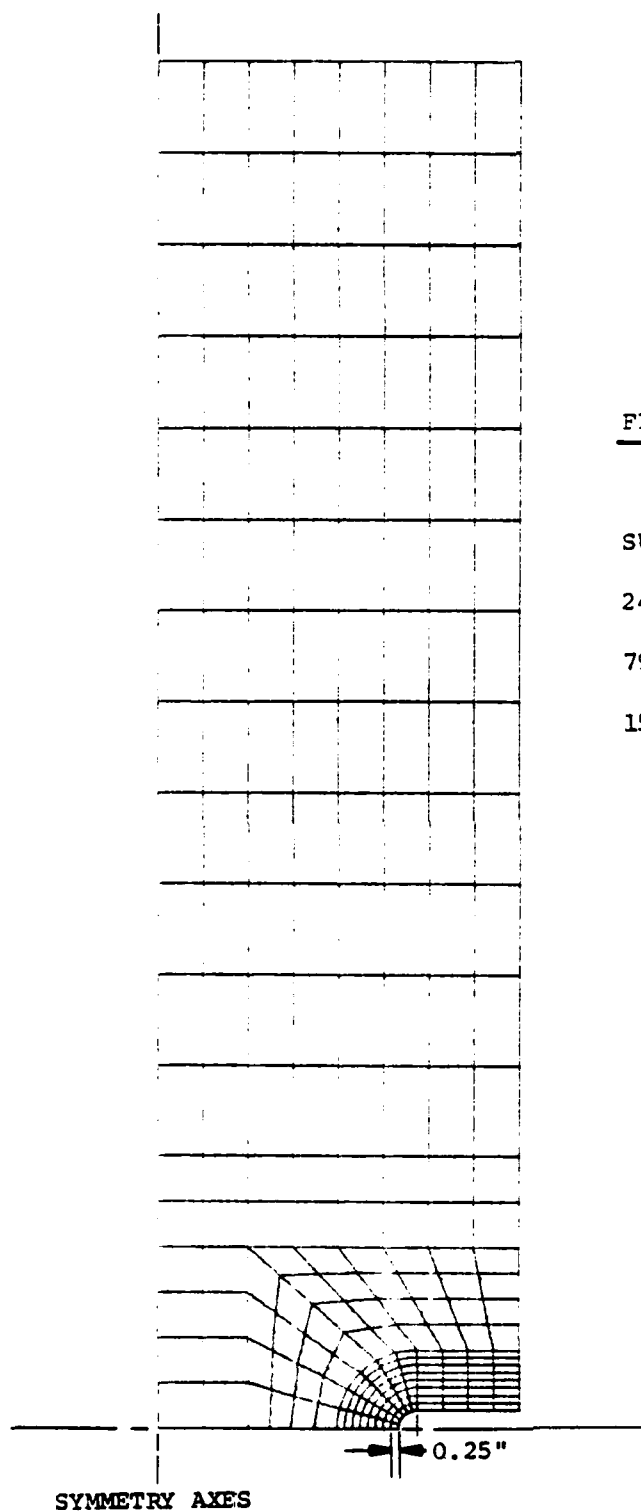


FIGURE 12

FINE MESH FOR DEEP NOTCH

SUBDIVISION OF COURSE MESH

240 ELEMENTS (ISOPARAMETRIC)

797 NODES

1528 DEGREES OF FREEDOM

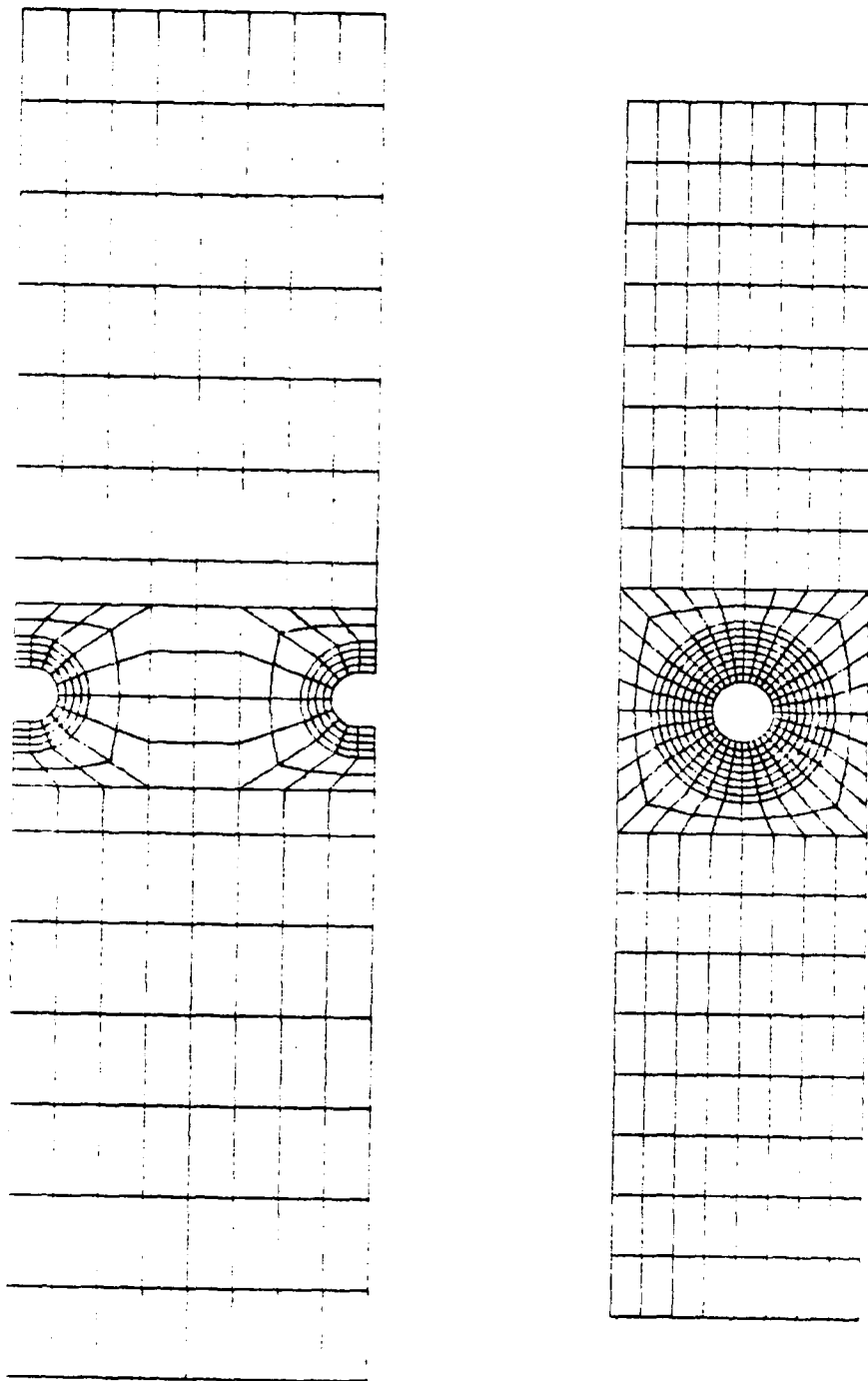


FIGURE 13

EXAMPLE OF COMPLETE PANEL MESHERS

FIGURE 14

COMPUTATIONAL FLOW CHART

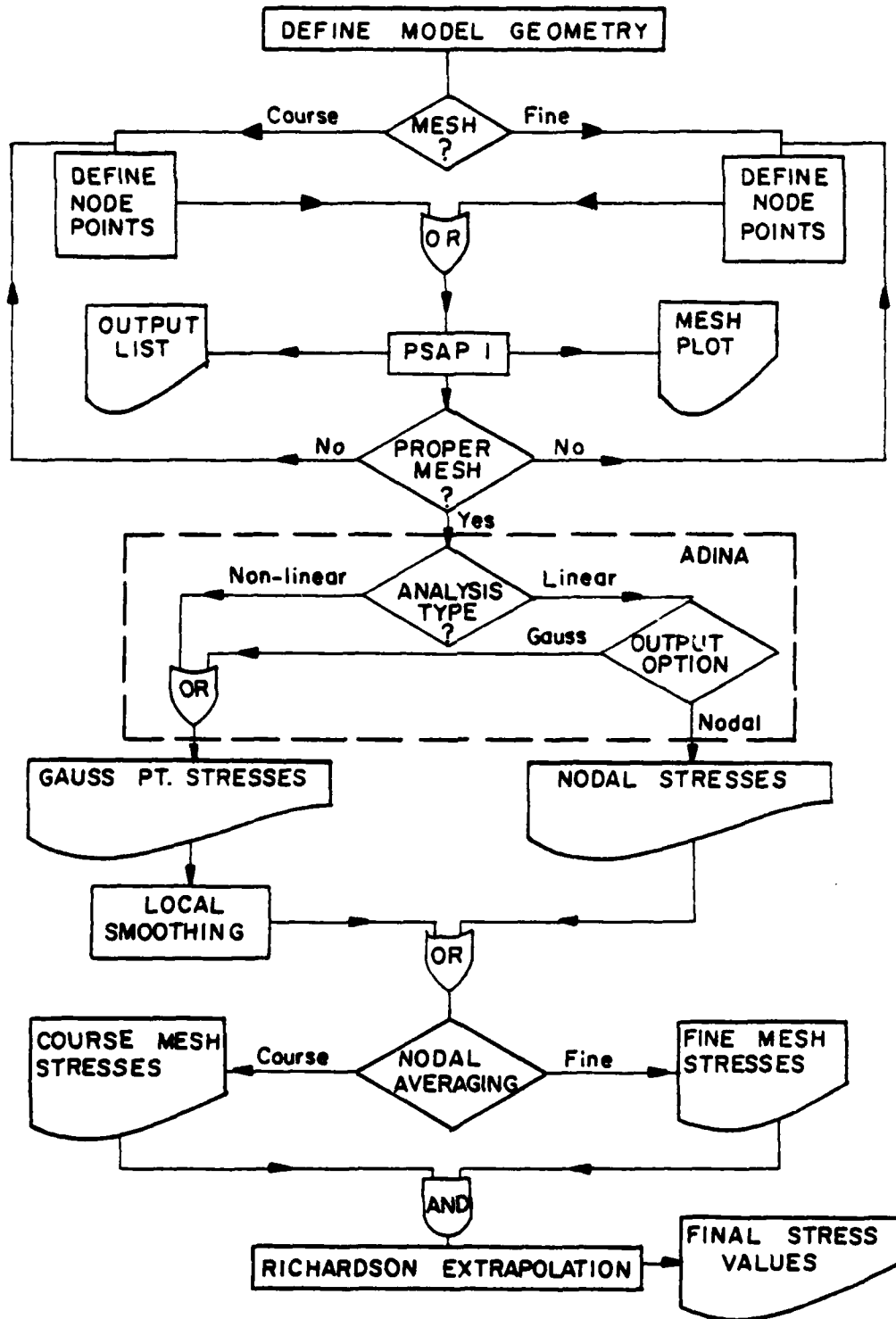


FIGURE 15  
CIRCULAR HOLE  $\lambda=0.2$  LINEAR RESULTS

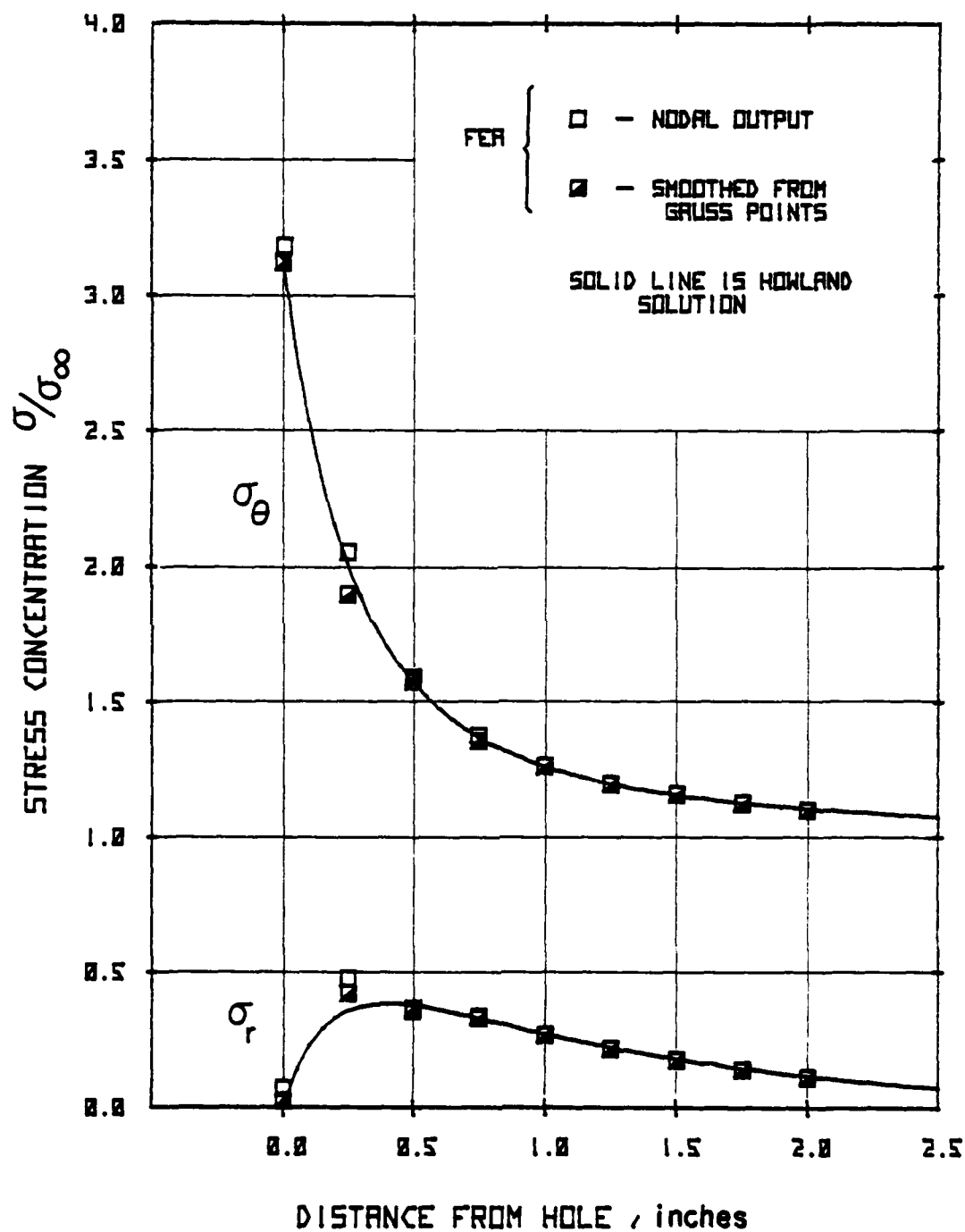


FIGURE 16  
CIRCULAR HOLE  $\lambda=0.25$  LINEAR RESULTS

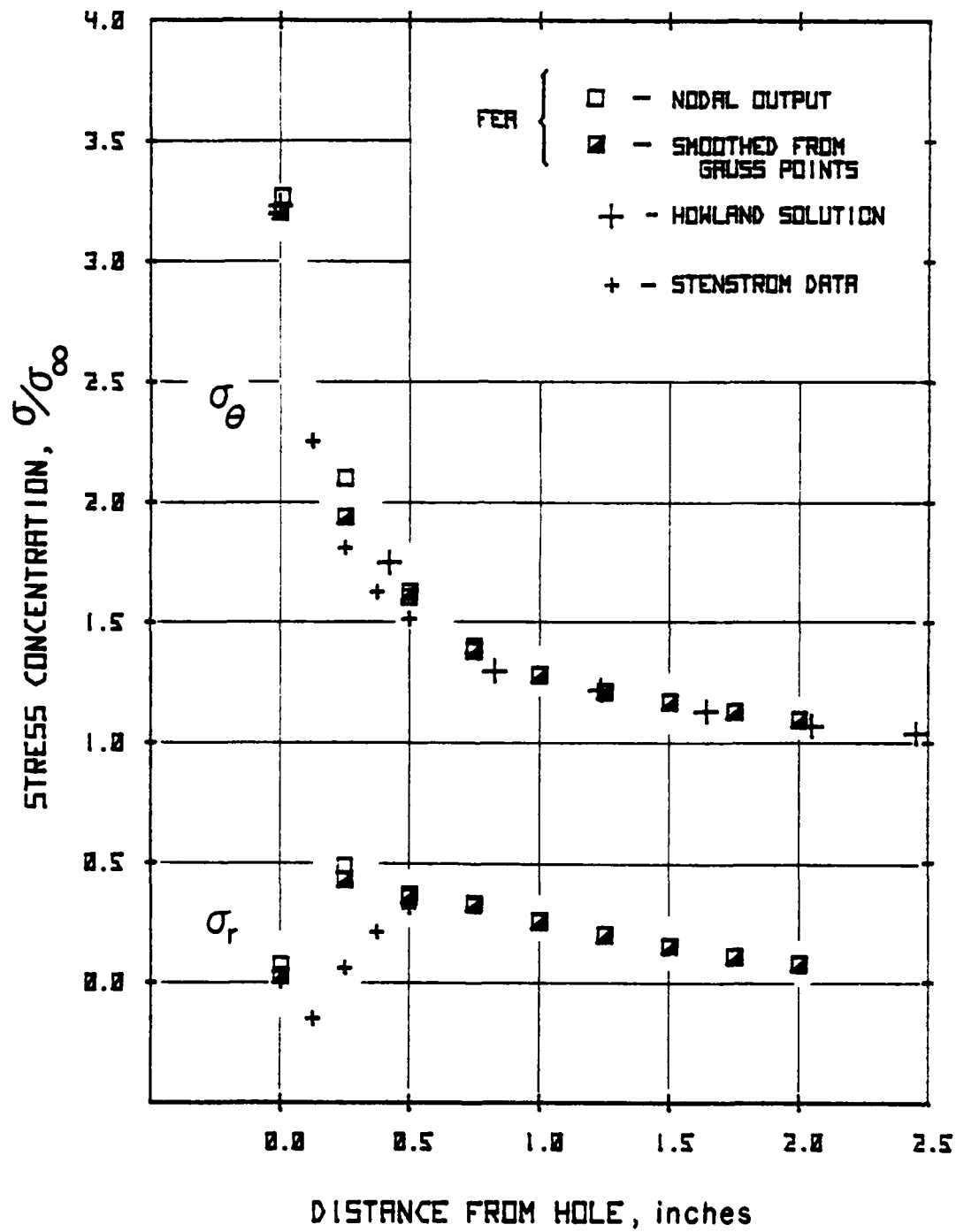


FIGURE 17

SHALLOW NOTCH LINEAR RESULTS

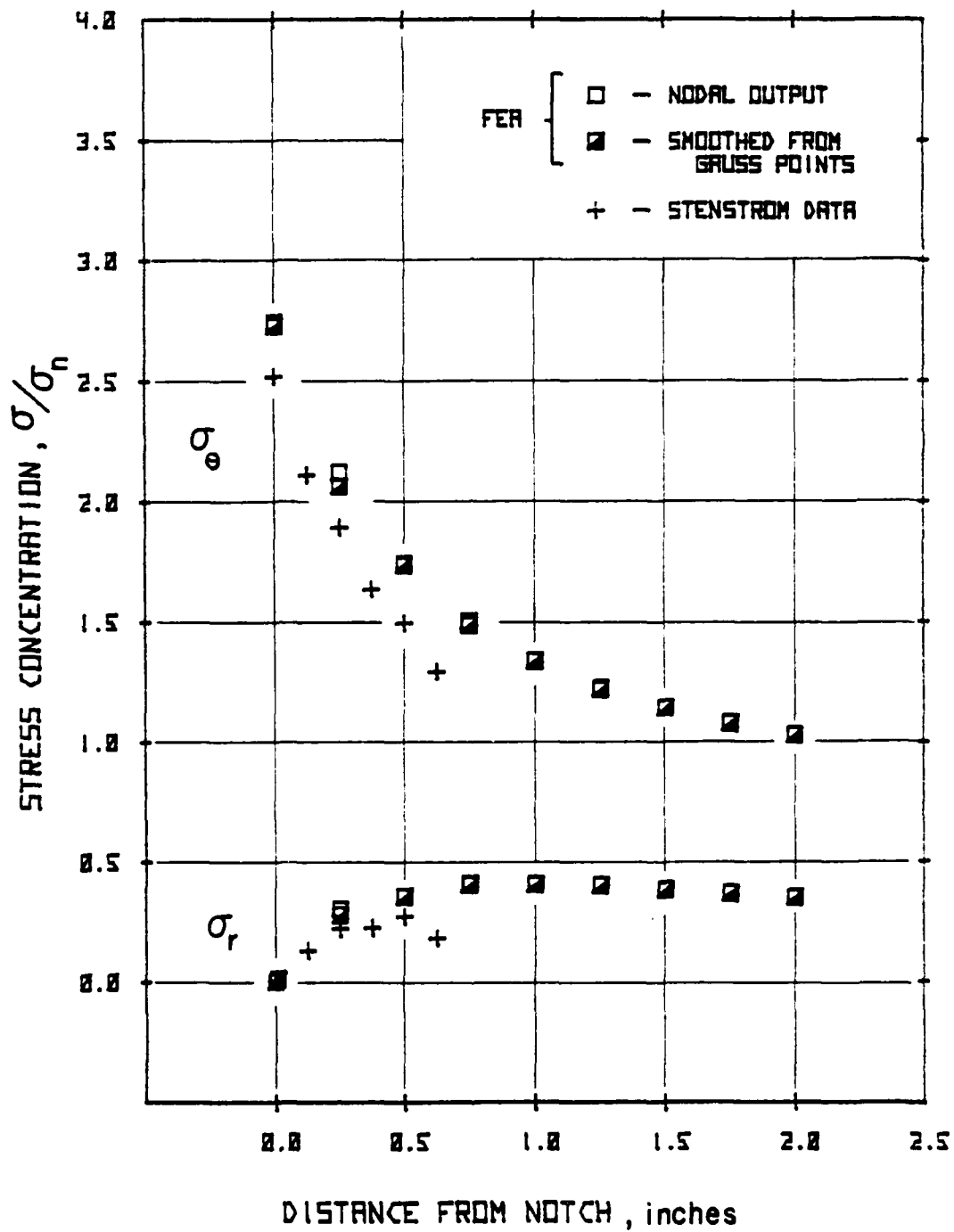


FIGURE 18  
DEEP NOTCH LINEAR RESULTS

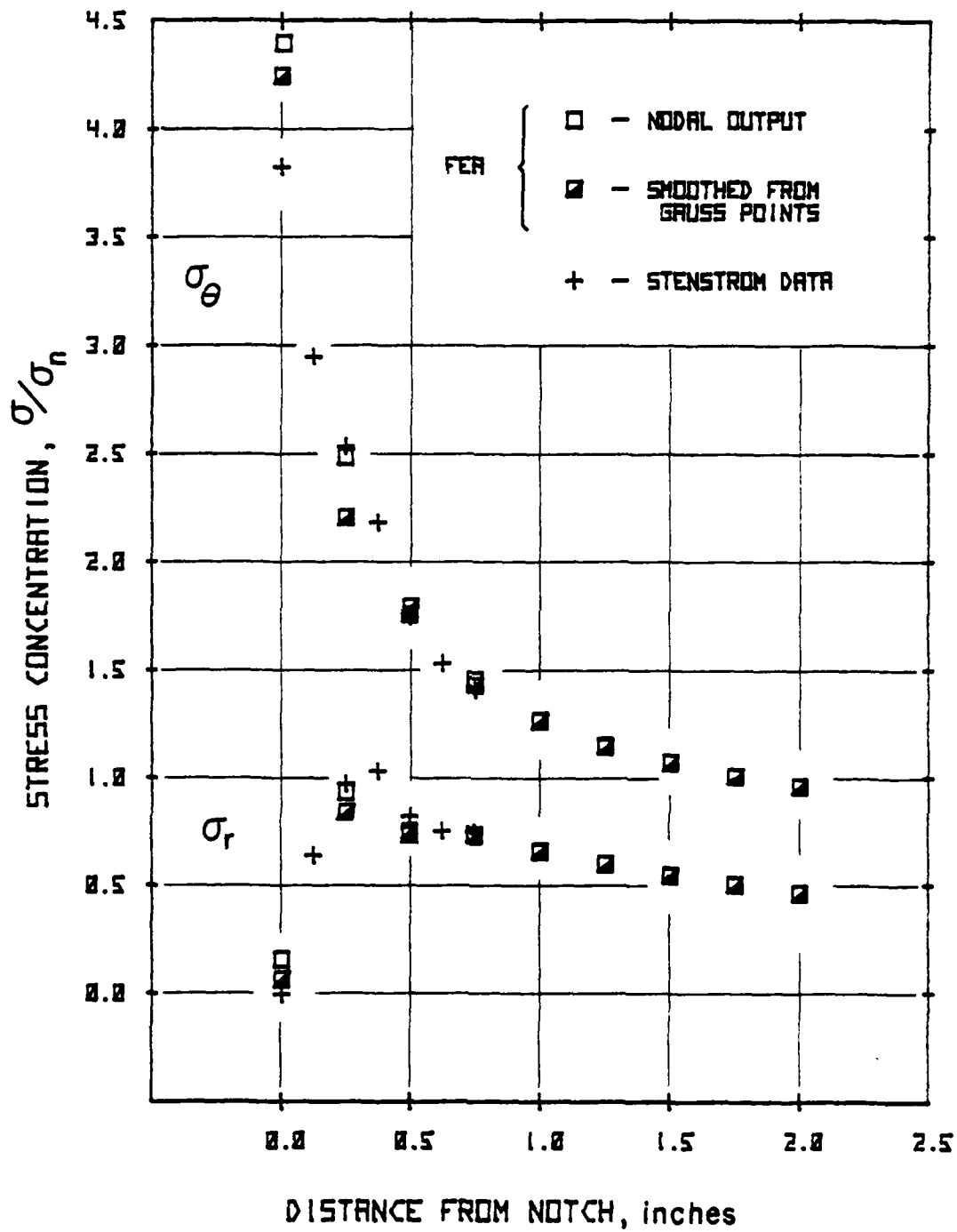




FIGURE 19

SHALLOW NOTCH 60000LB LOAD ELASTIC-PLASTIC RESULTS

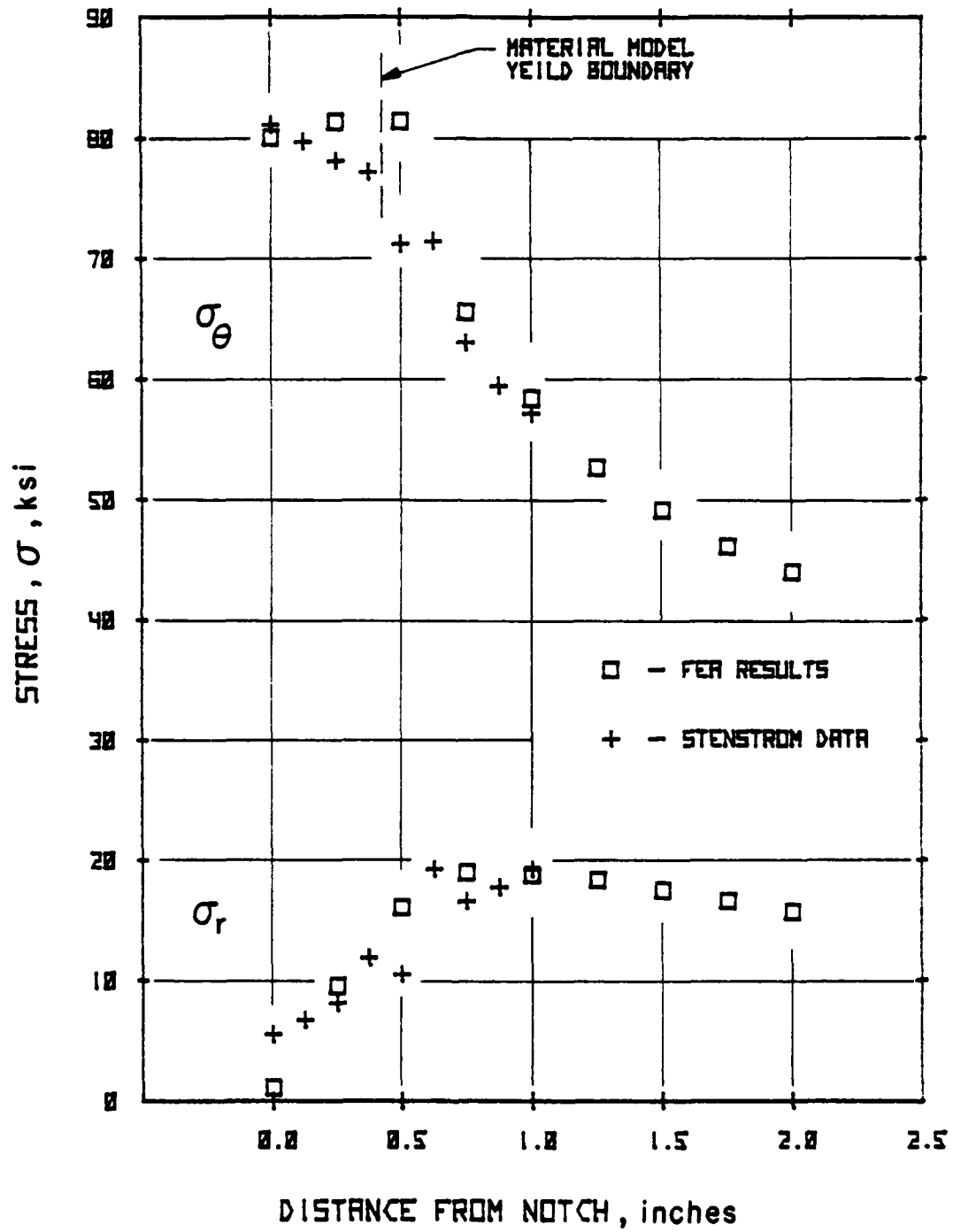


FIGURE 20

SHALLOW NOTCH 65000 LB LOAD ELASTIC-PLASTIC RESULTS

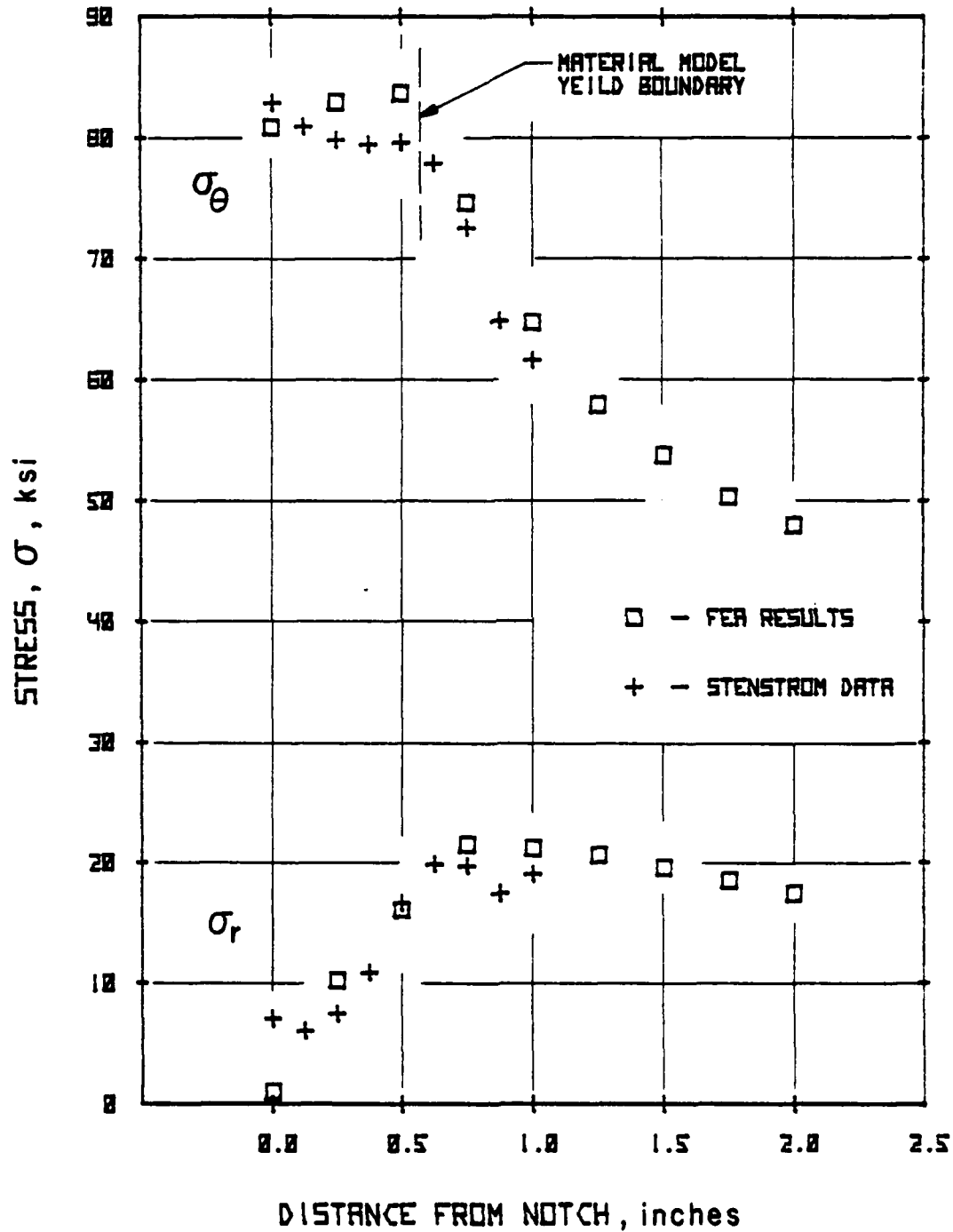
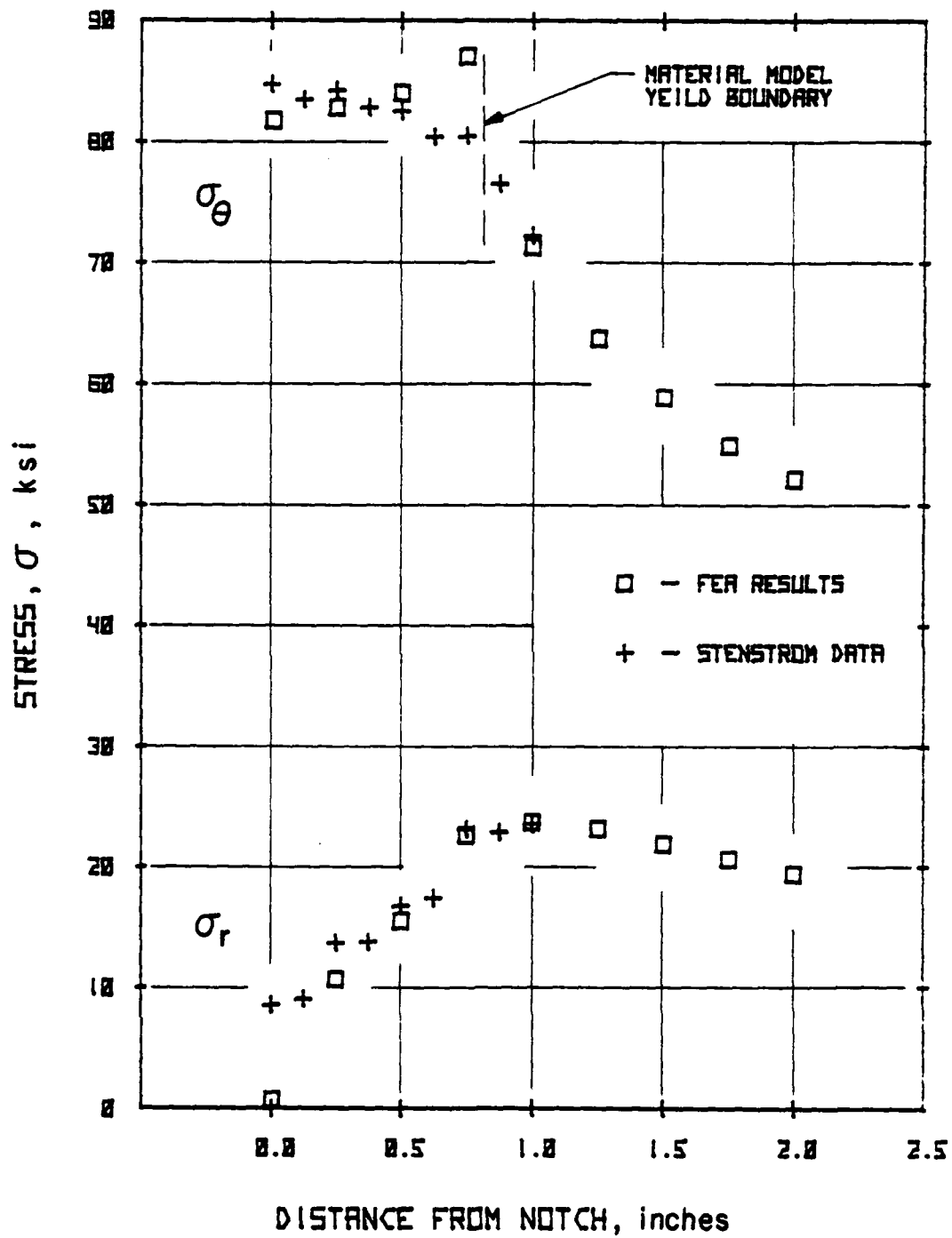


FIGURE 21

SHALLOW NOTCH 70000 LB LOAD ELASTIC-PLASTIC RESULTS



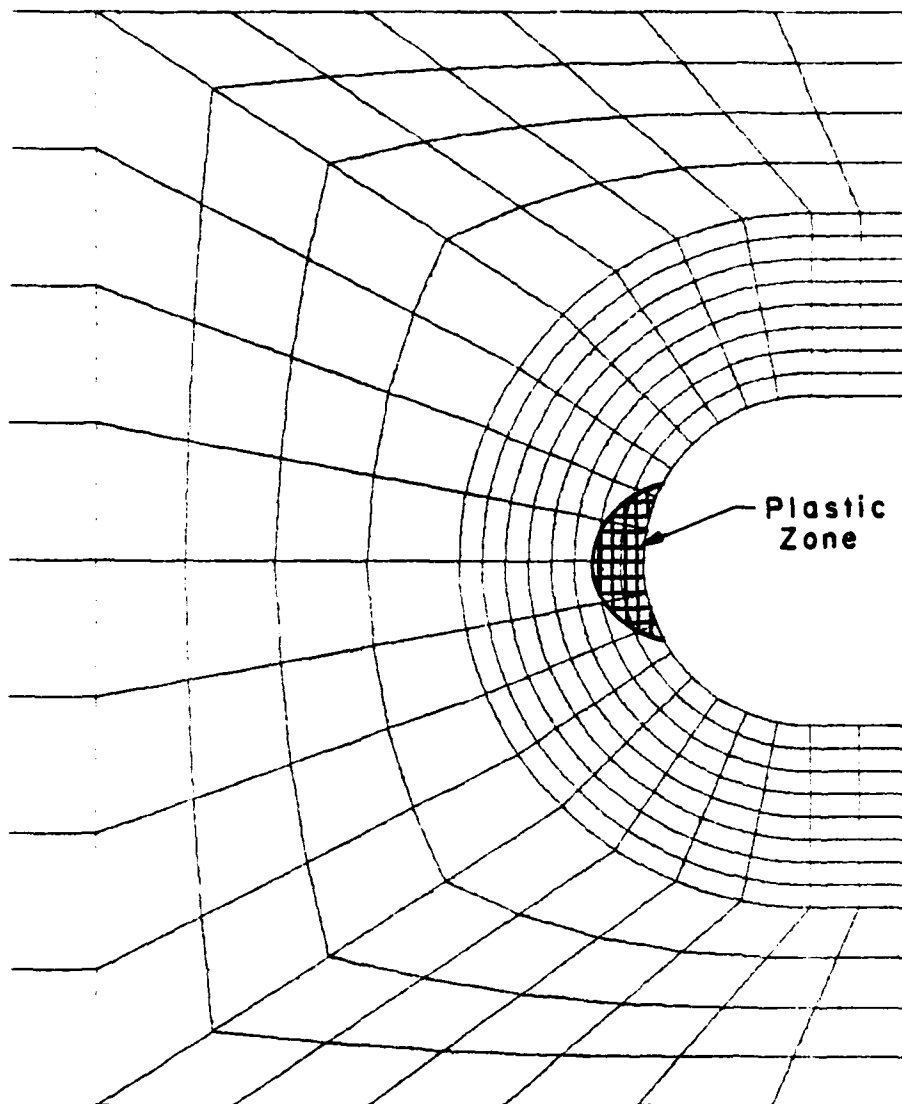


FIGURE 22

SHALLOW NOTCH 60,000 LB LOAD PLASTIC ZONE

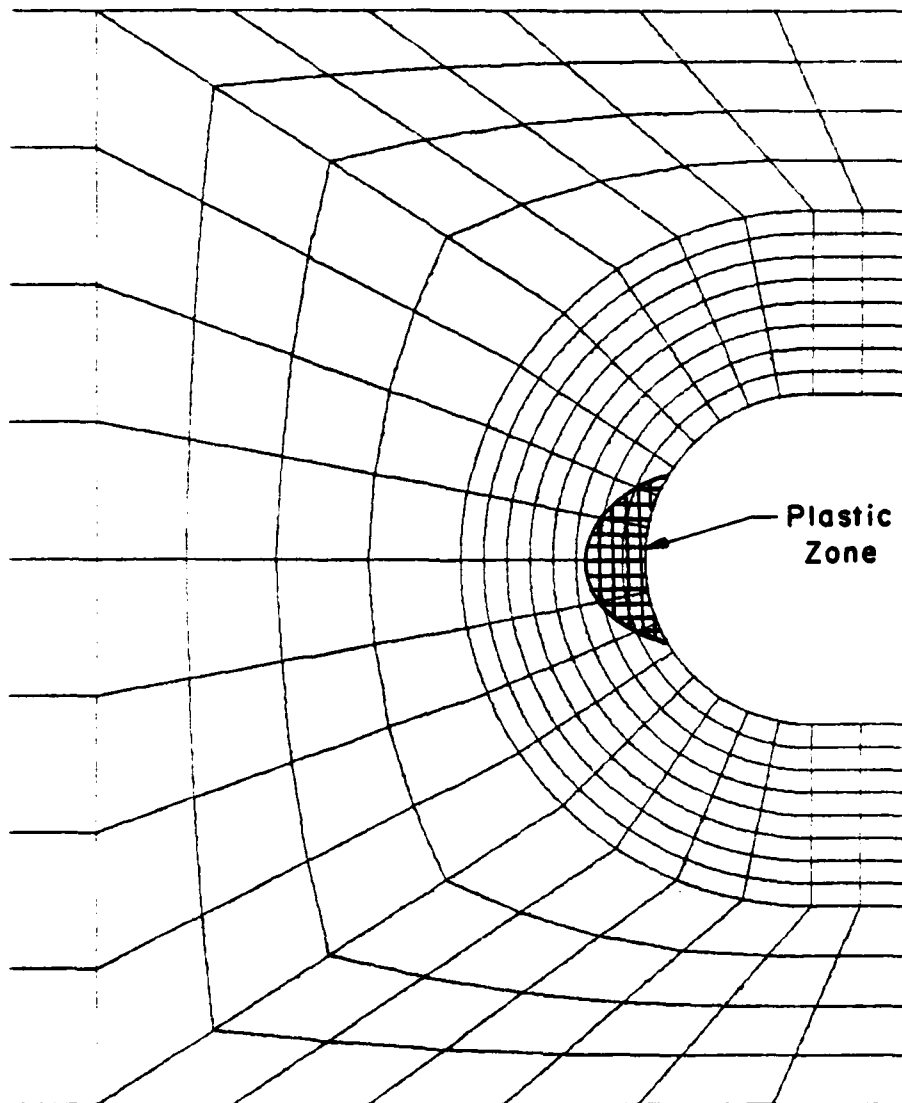


FIGURE 23

SHALLOW NOTCH 65,000 LB LOAD PLASTIC ZONE

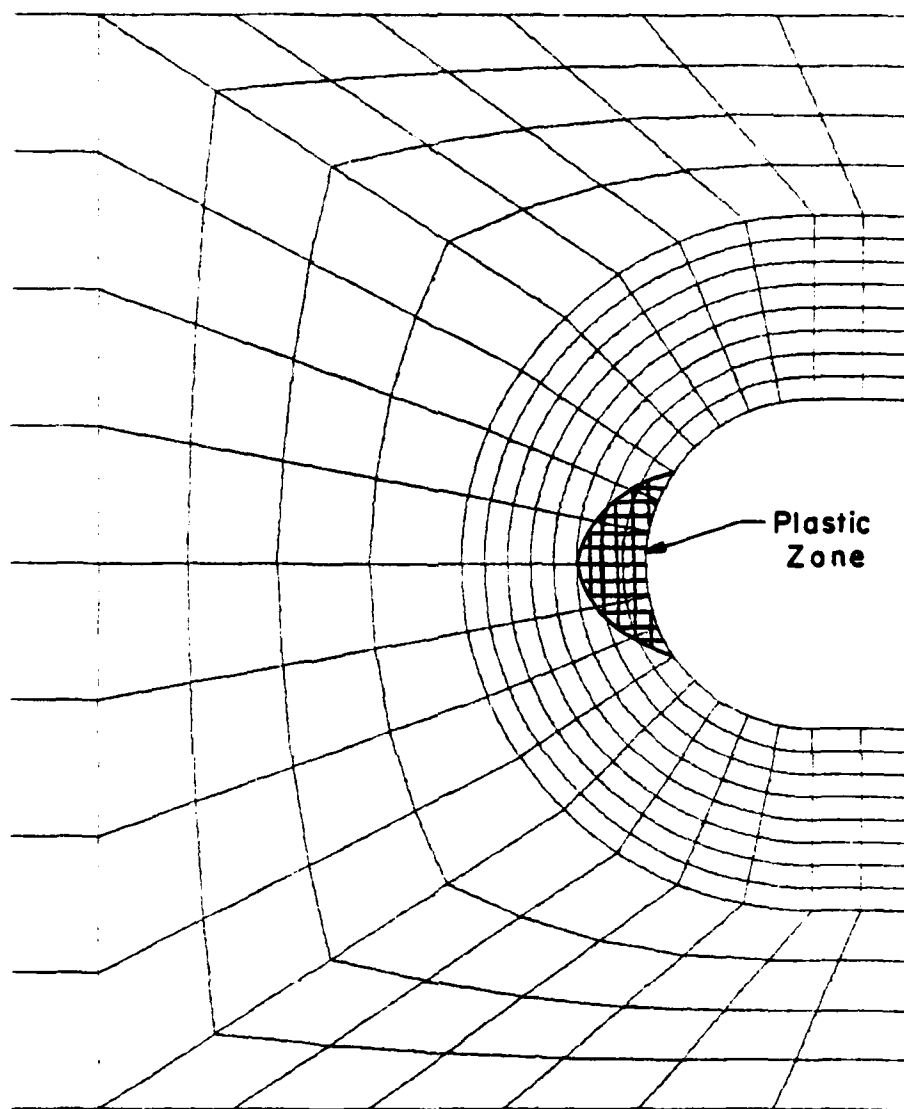


FIGURE 24

SHALLOW NOTCH 70,000 LB LOAD PLASTIC ZONE

FIGURE 25

SHALLOW NOTCH RESIDUAL  $\sigma_{\theta}$  FROM 50,000 LB LOAD

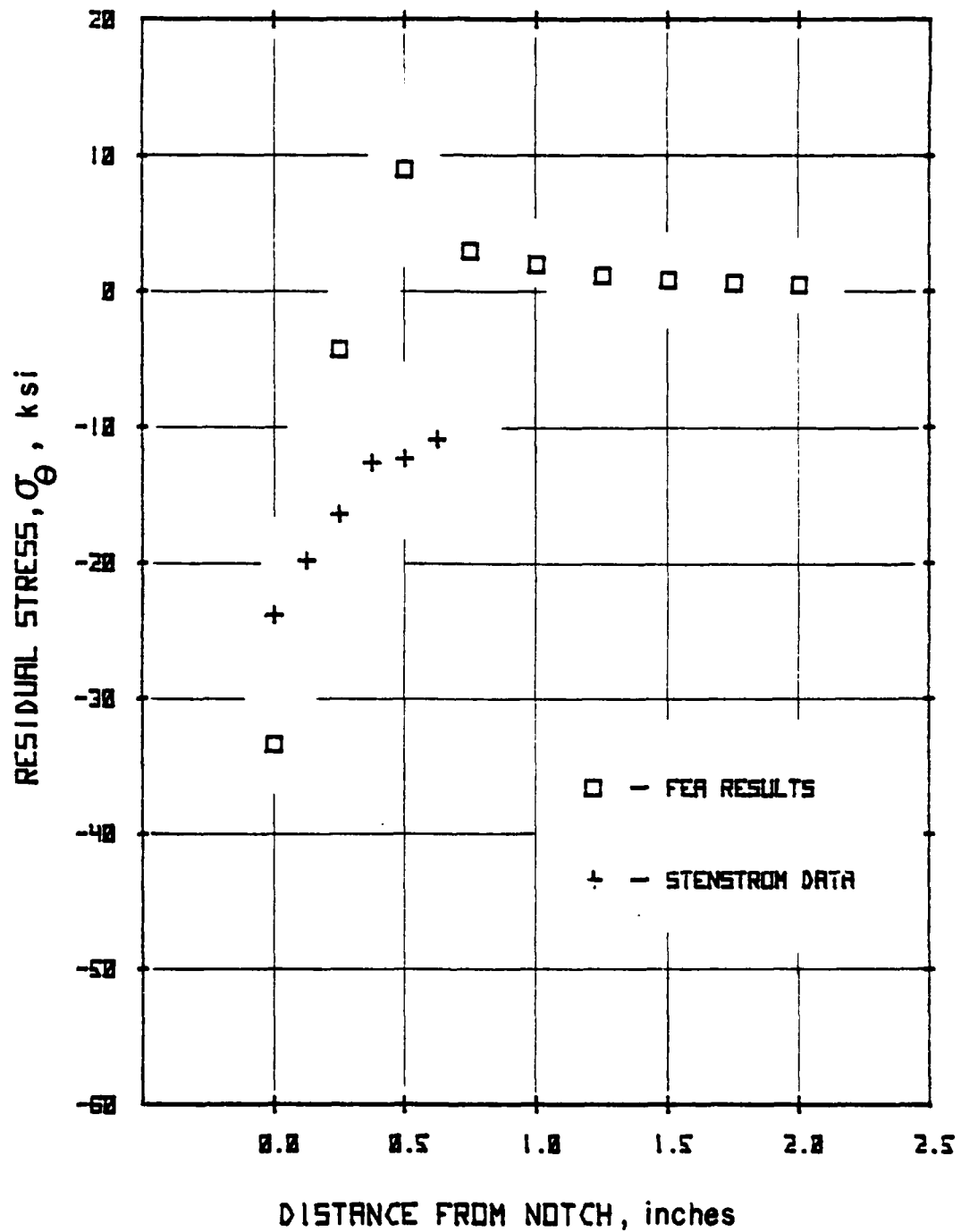


FIGURE 25

SHALLOW NOTCH RESIDUAL  $\sigma_r$  FROM 60,000 LB LOAD

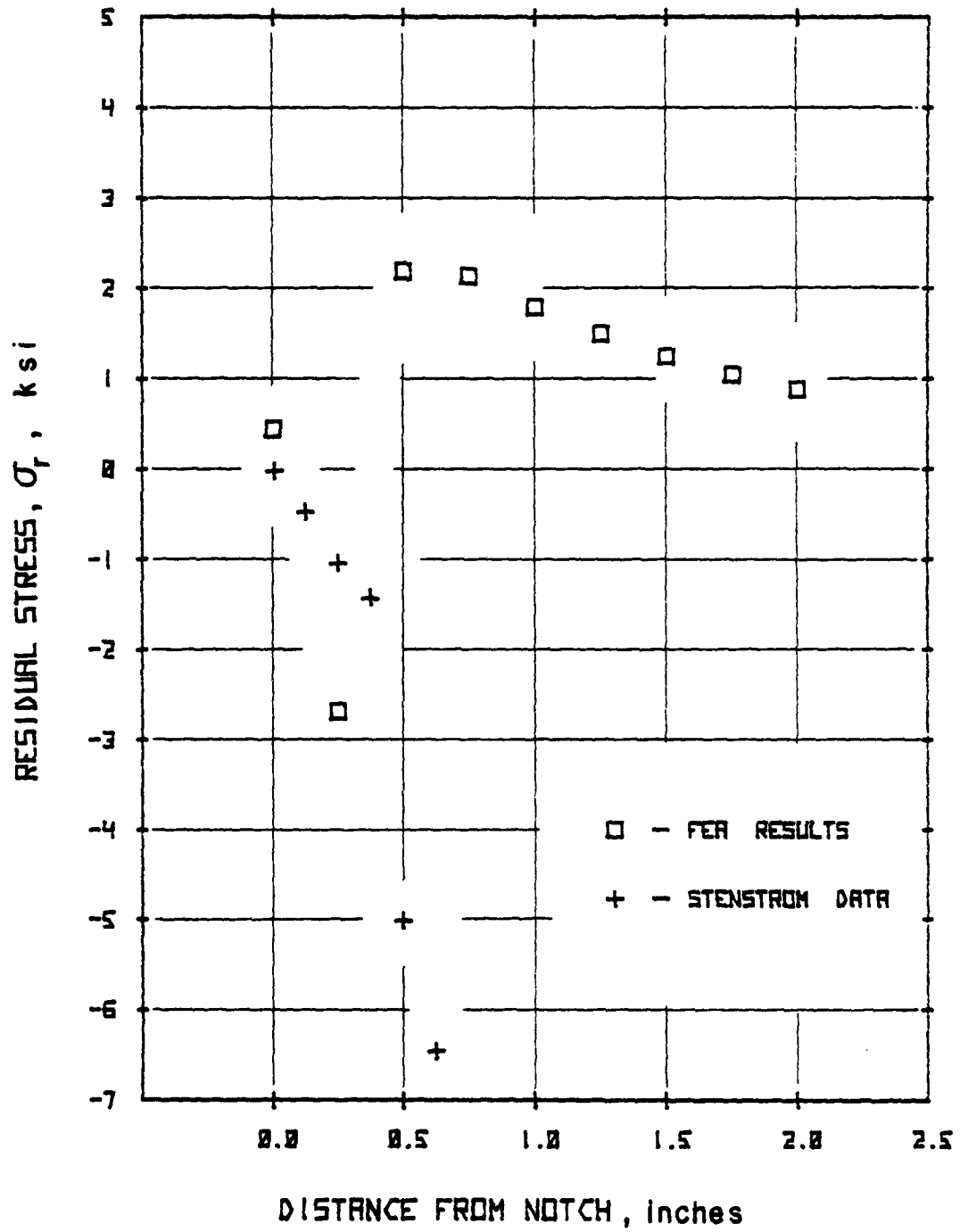




FIGURE 27

SHALLOW NOTCH RESIDUAL  $\sigma_{\theta}$  FROM 65,000 LB LOAD

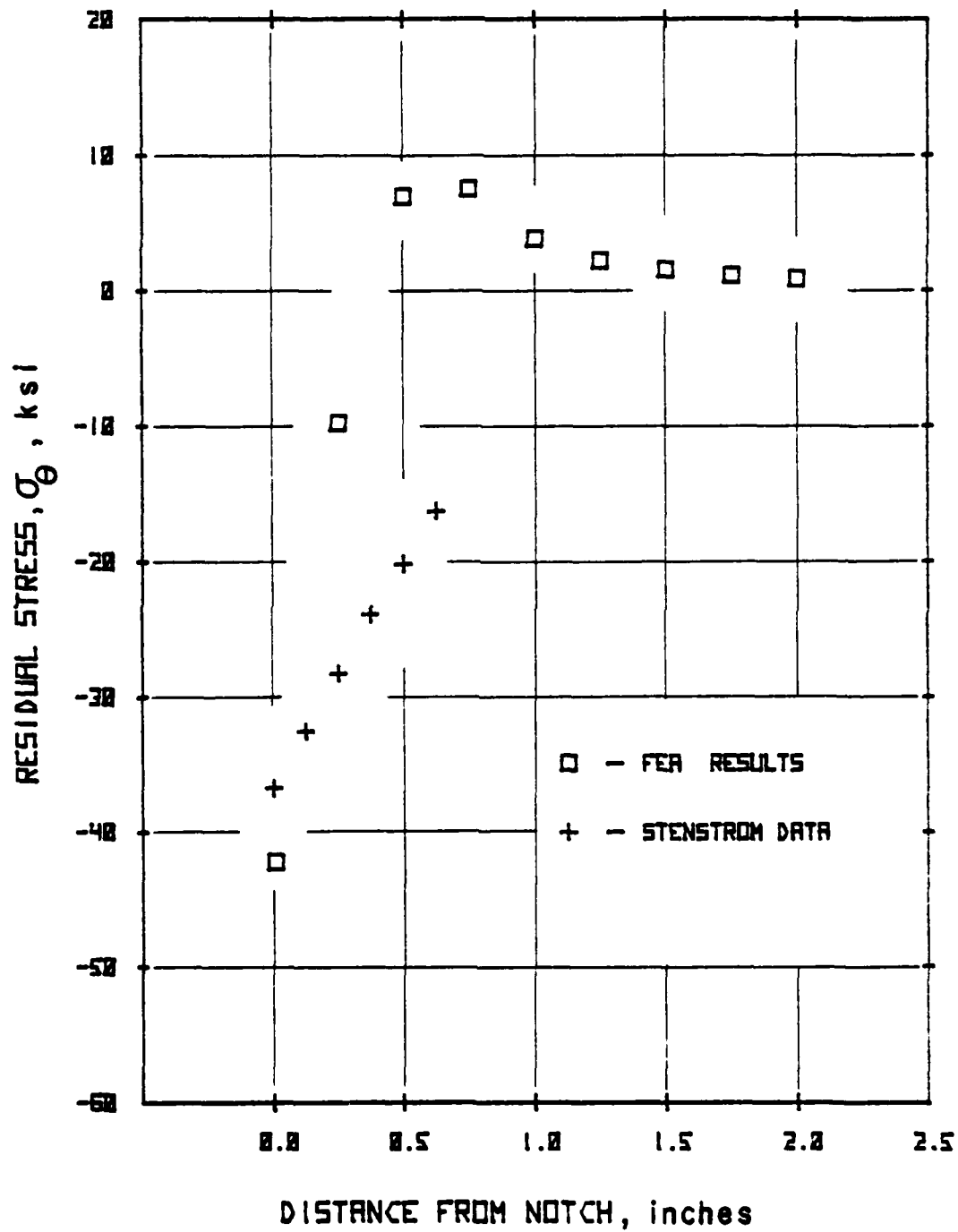


FIGURE 28

SHALLOW NOTCH RESIDUAL  $\sigma_r$  FROM 65,000 LB LOAD

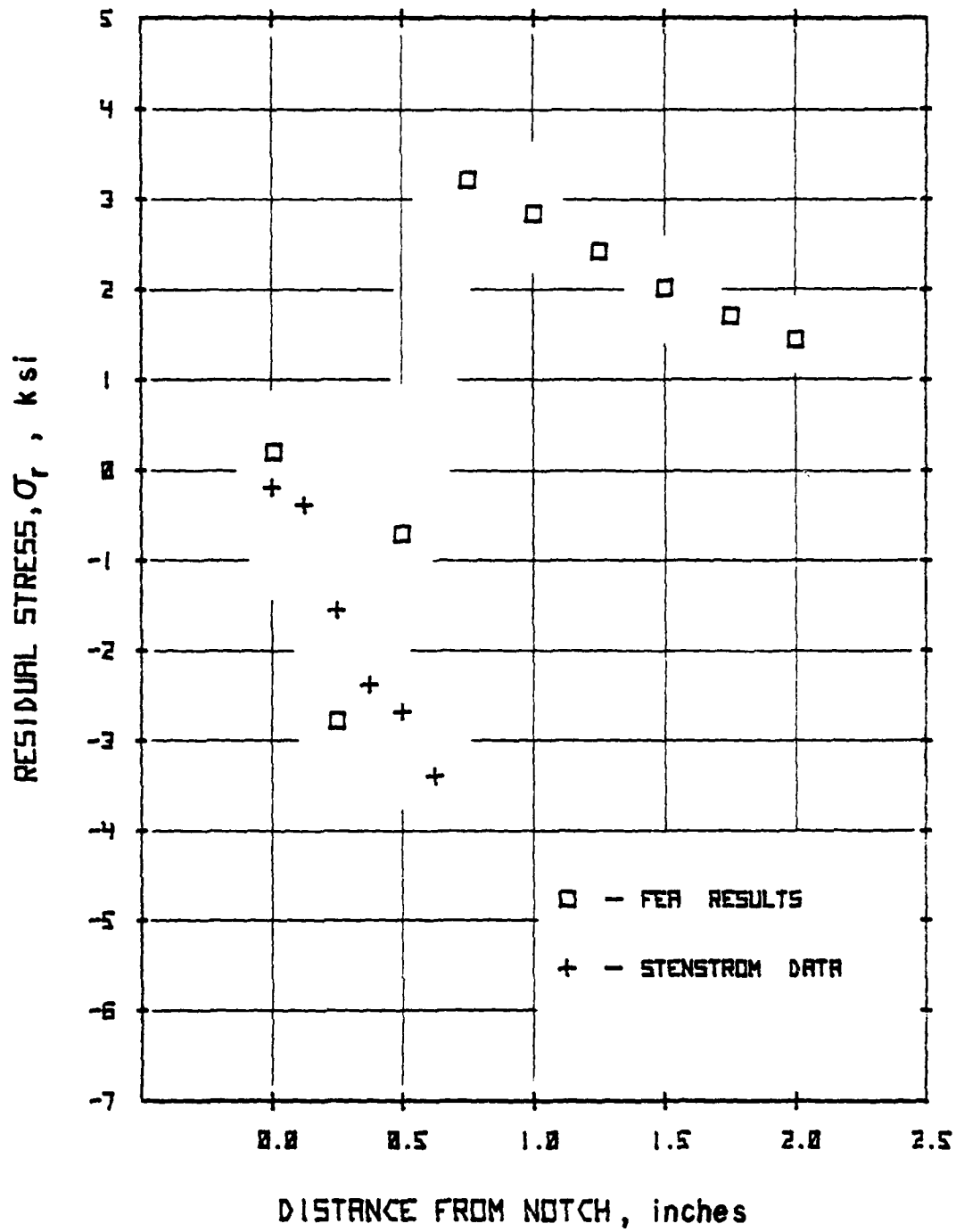


FIGURE 29

SHALLOW NOTCH RESIDUAL  $\sigma_{\theta}$  FROM 70,000 LB LOAD

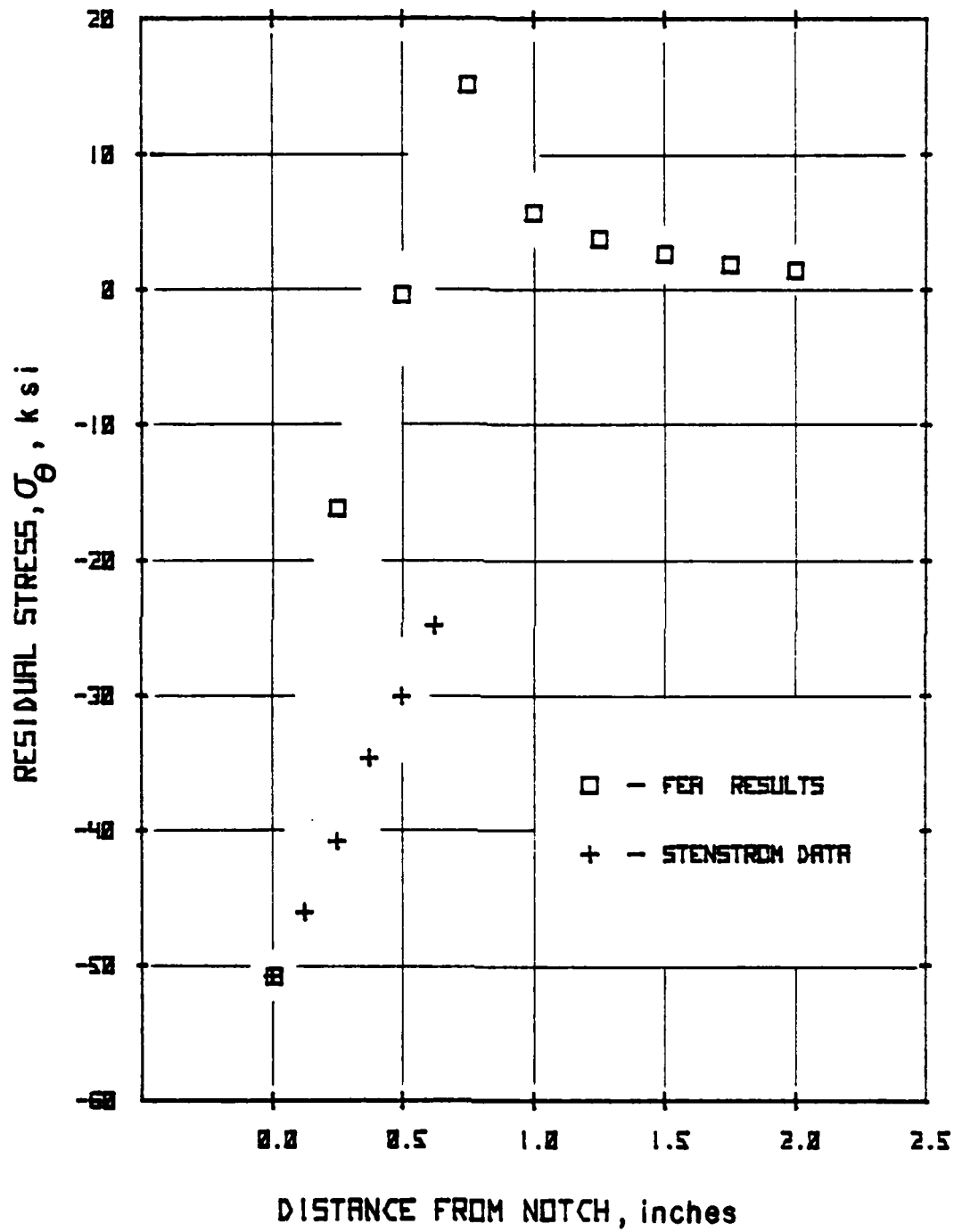


FIGURE 30

SHALLOW NOTCH RESIDUAL  $\sigma_r$  FROM 70,000 LB LOAD

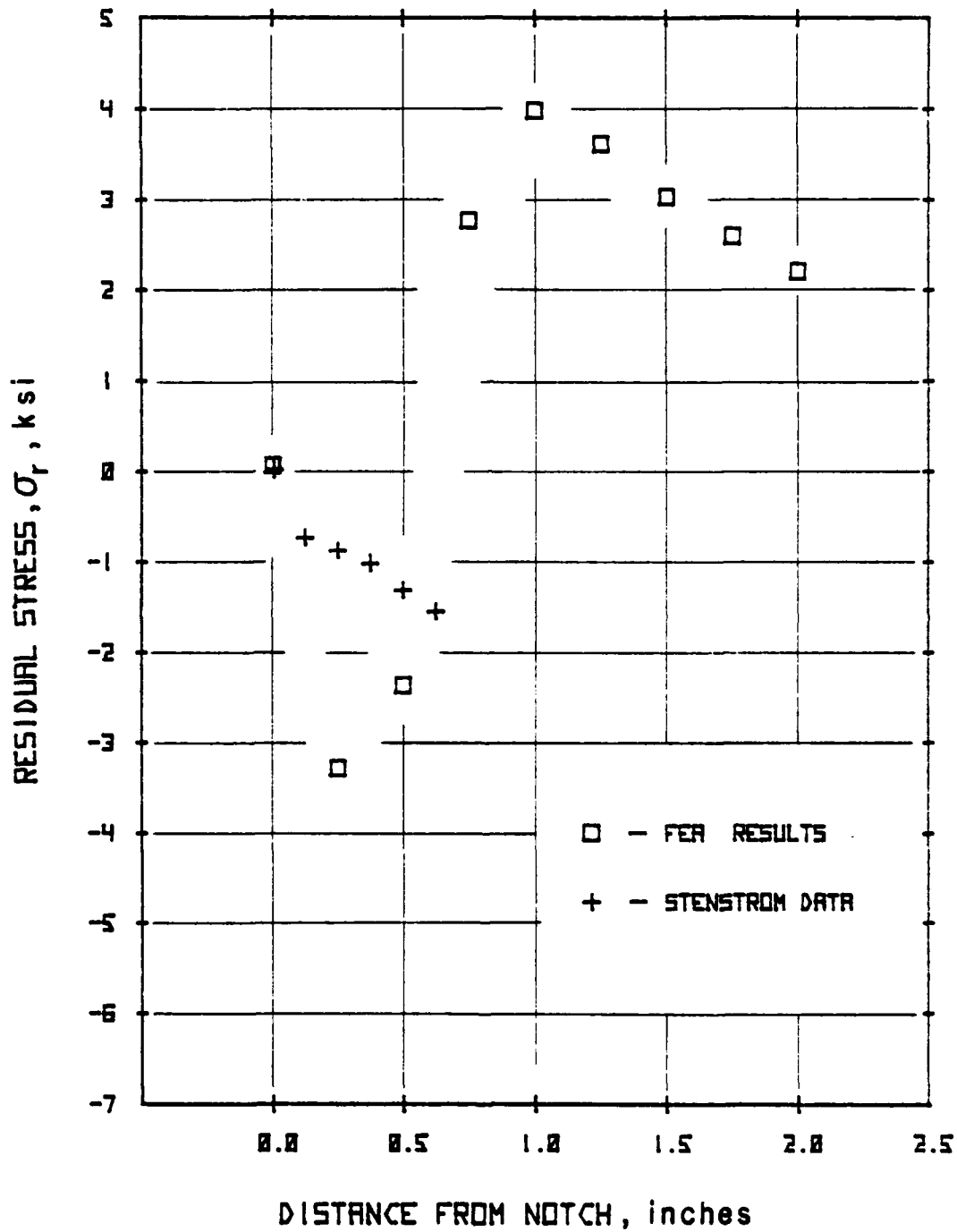


FIGURE 31

DEEP NOTCH PLASTIC LOADING RESULTS

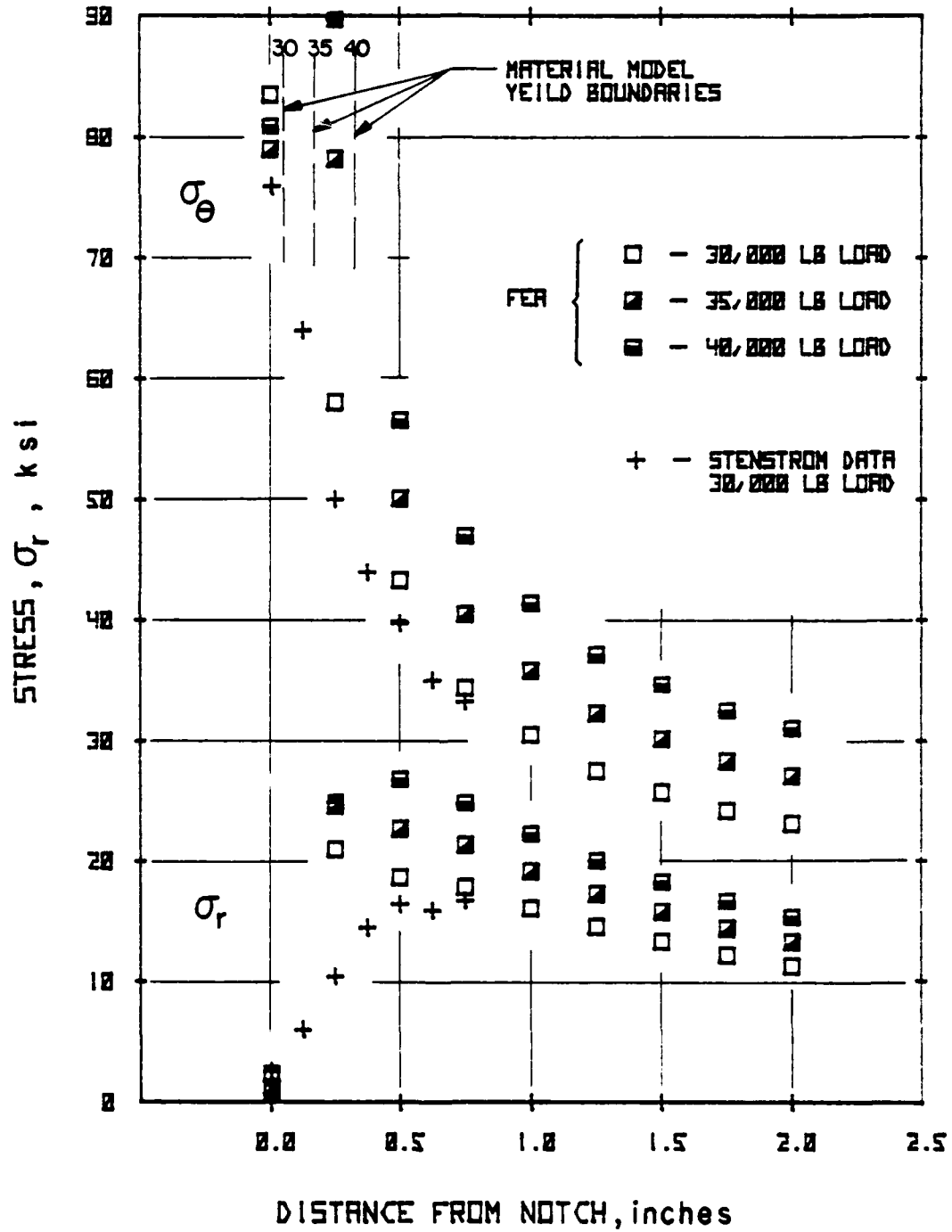


FIGURE 32

DEEP NOTCH  $\sigma_{\theta}$  RESIDUALS

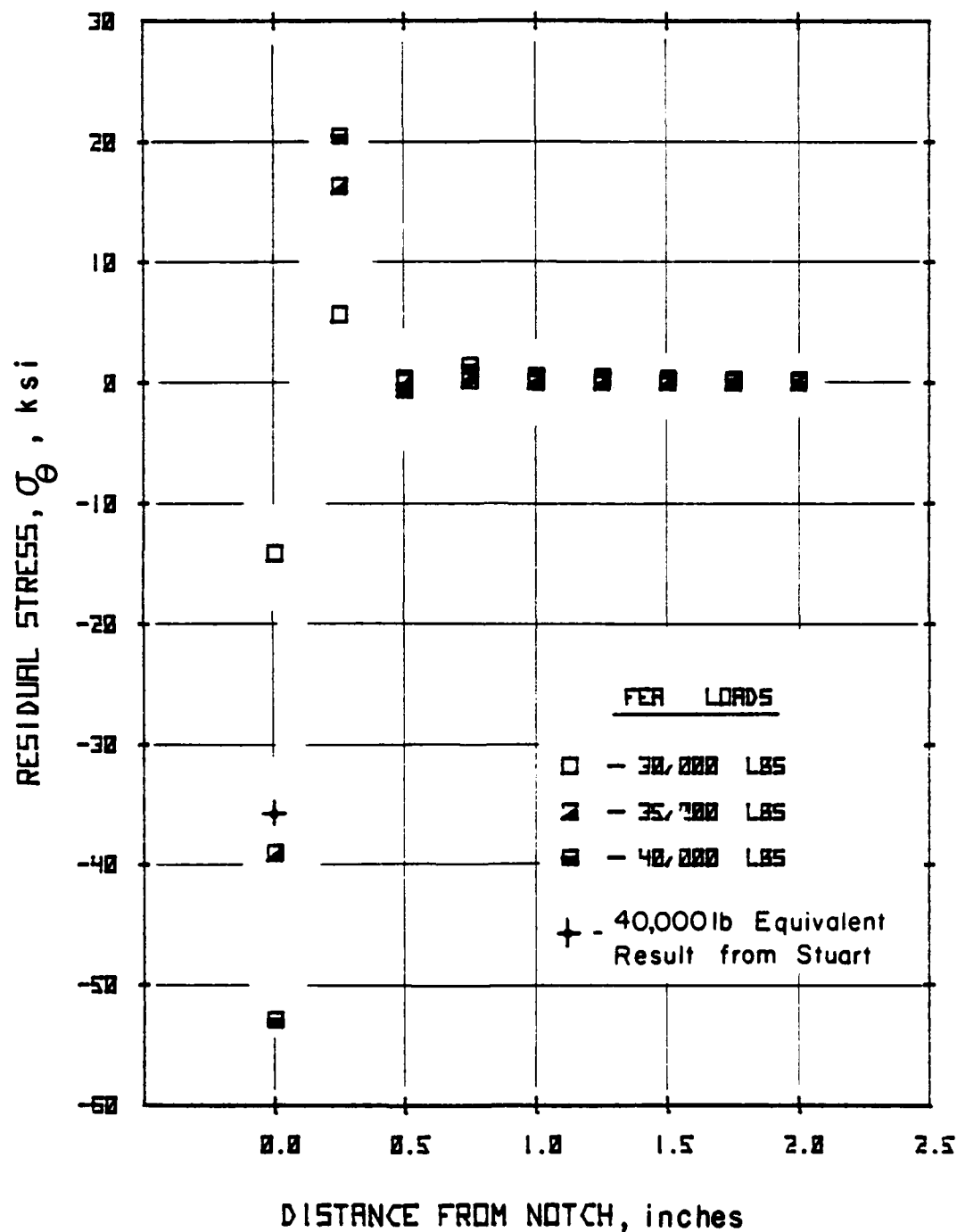
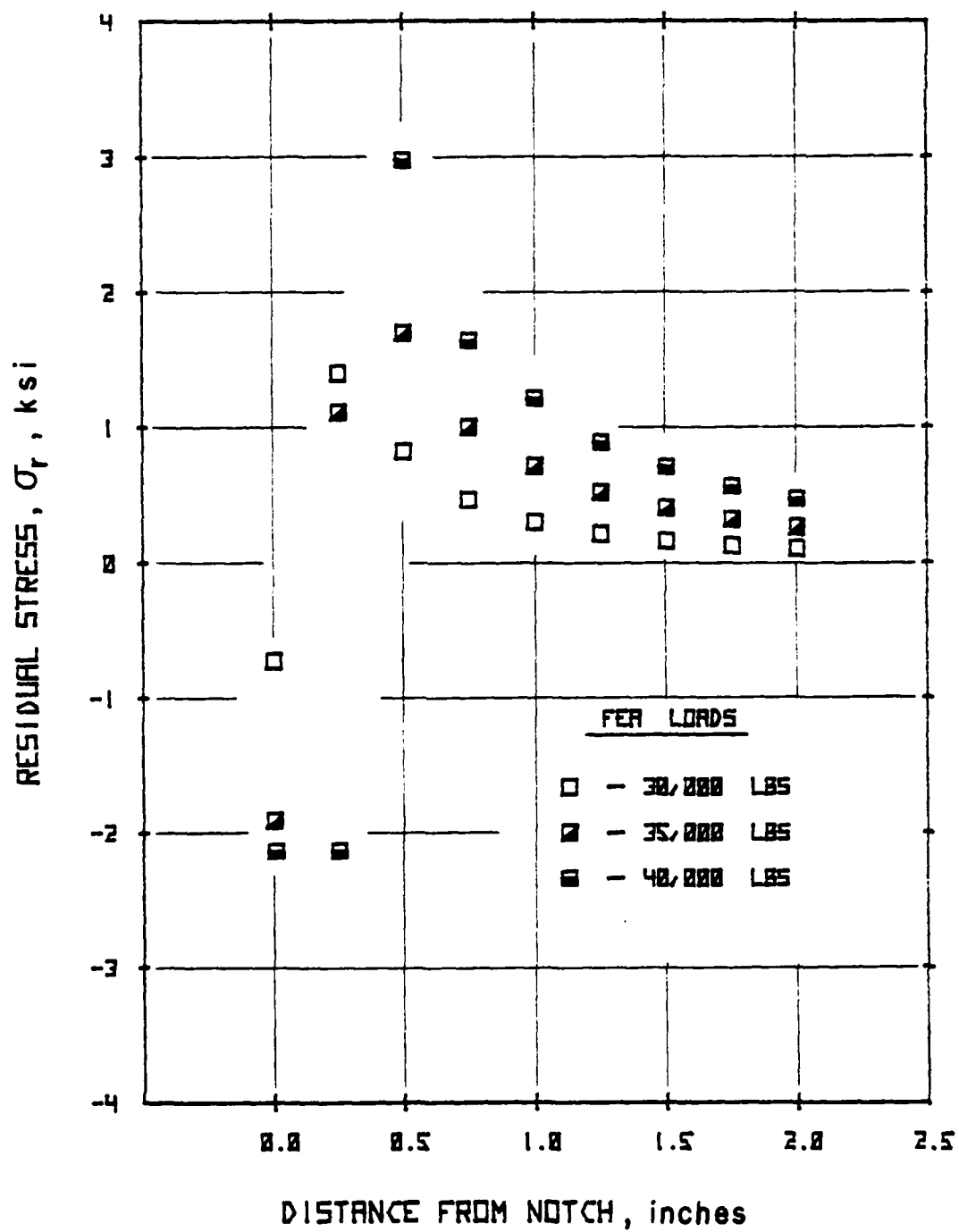


FIGURE 33

DEEP NOTCH  $\sigma_r$  RESIDUALS



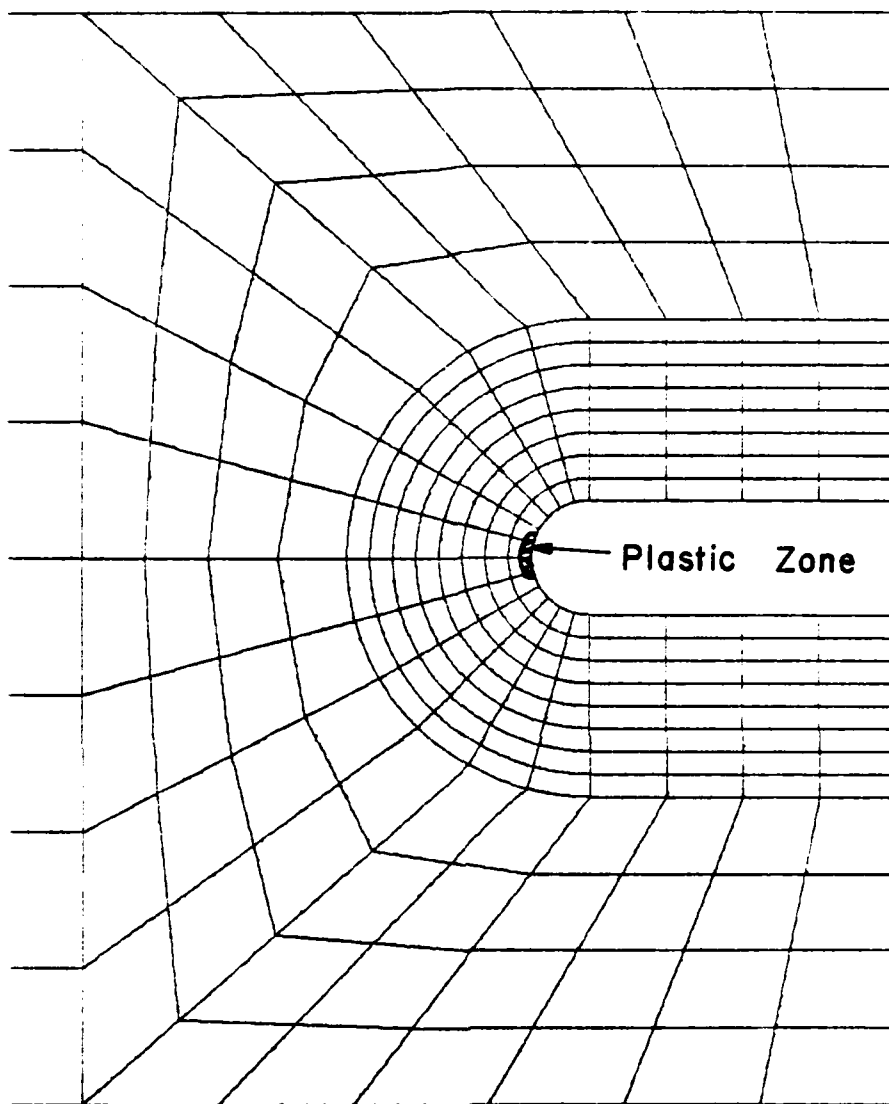


FIGURE 34

DEEP NOTCH 30,000 LB LOAD PLASTIC ZONE



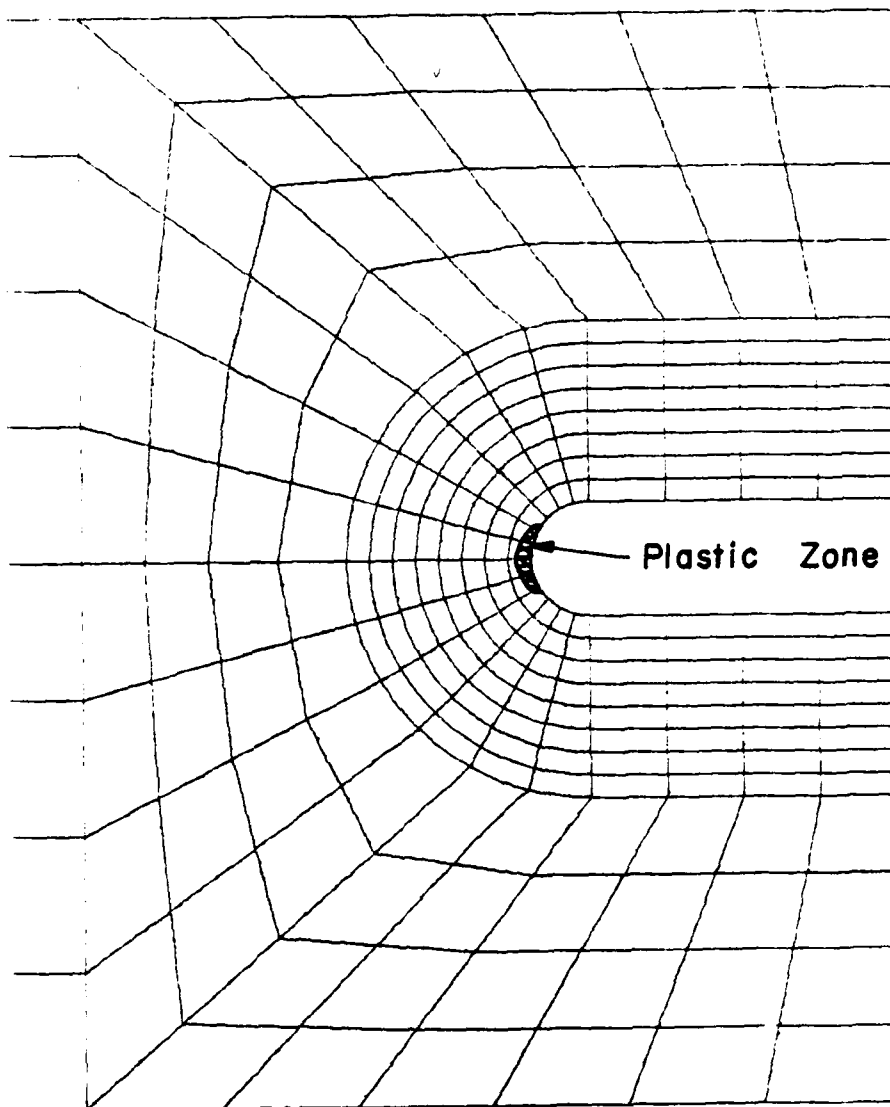


FIGURE 35

DEEP NOTCH 35,000 LB LOAD PLASTIC ZONE

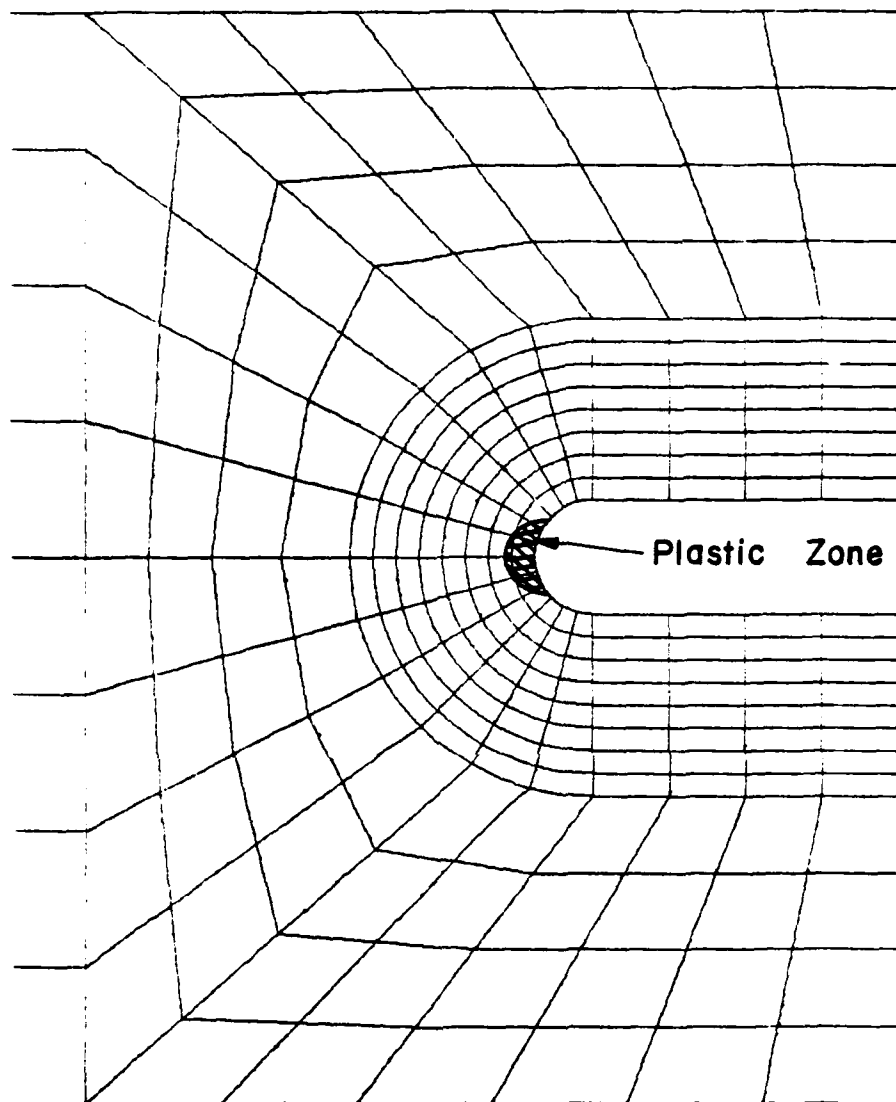
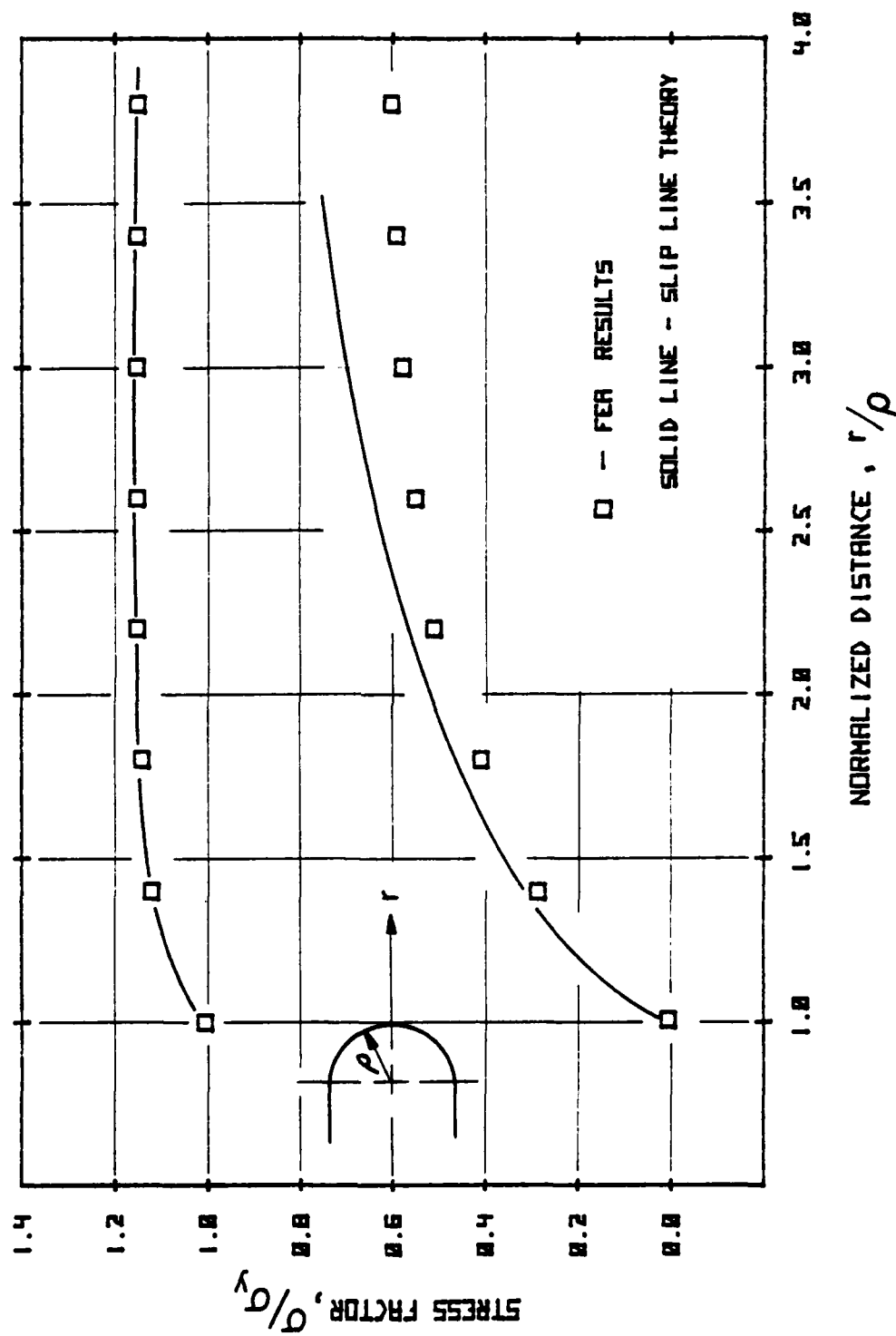


FIGURE 36

DEEP NOTCH 40,000 LB LOAD PLASTIC ZONE

FIGURE 37  
RIGID-PERFECTLY-PLASTIC RESULTS



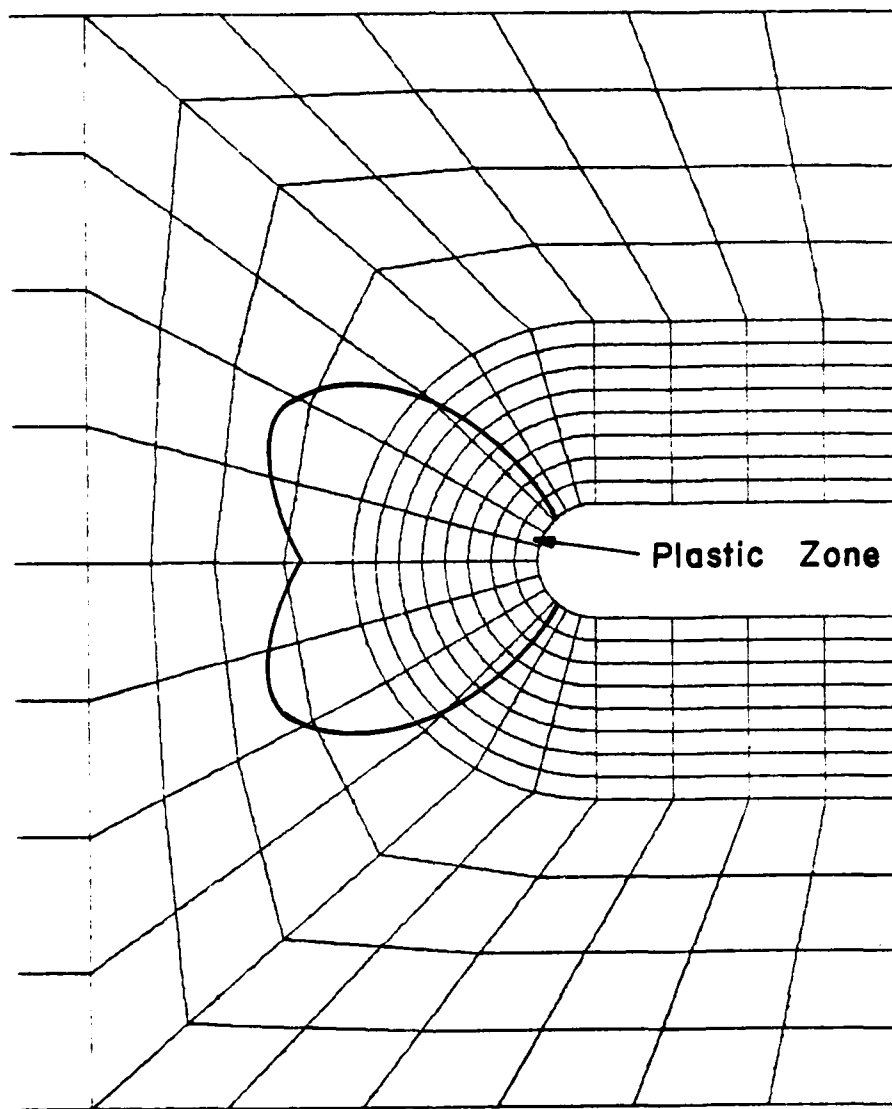


FIGURE 38

RIGID-PERFECTLY-PLASTIC INITIAL PLASTIC ZONE

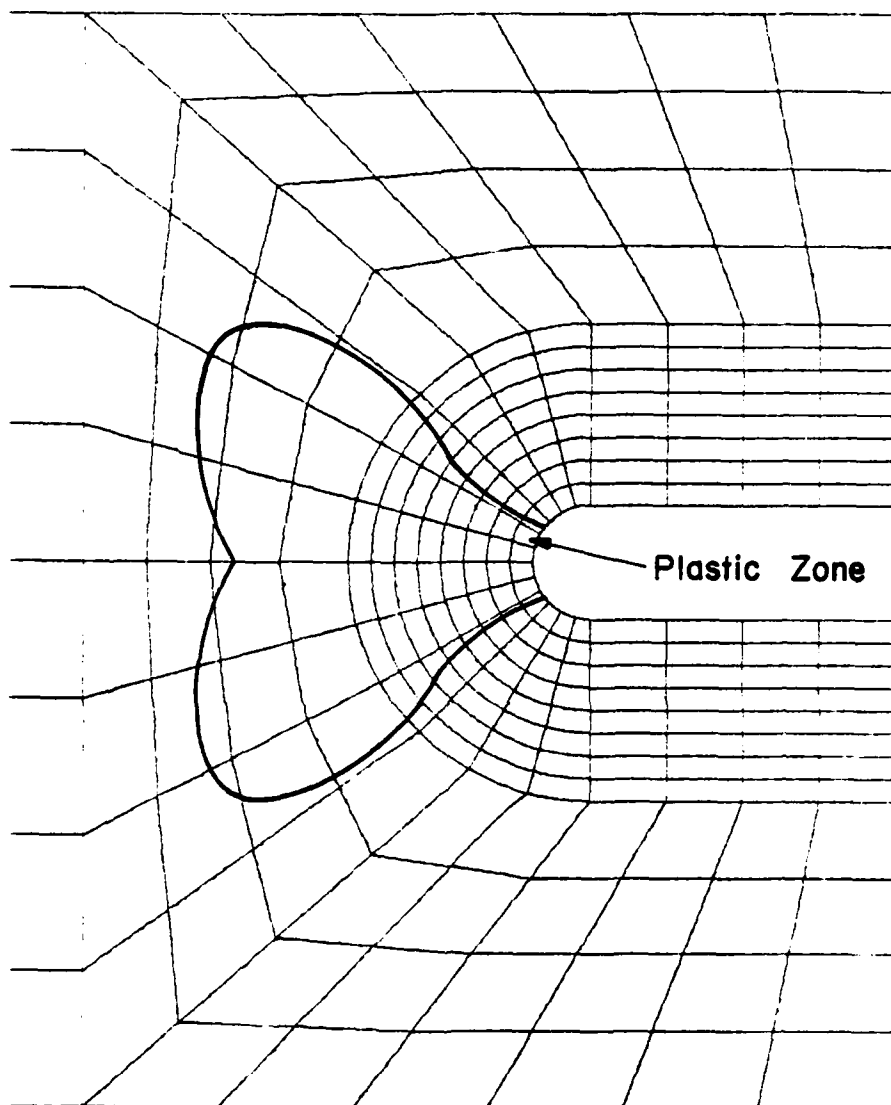


FIGURE 39

RIGID-PERFECTLY-PLASTIC INTERMEDIATE PLASTIC ZONE

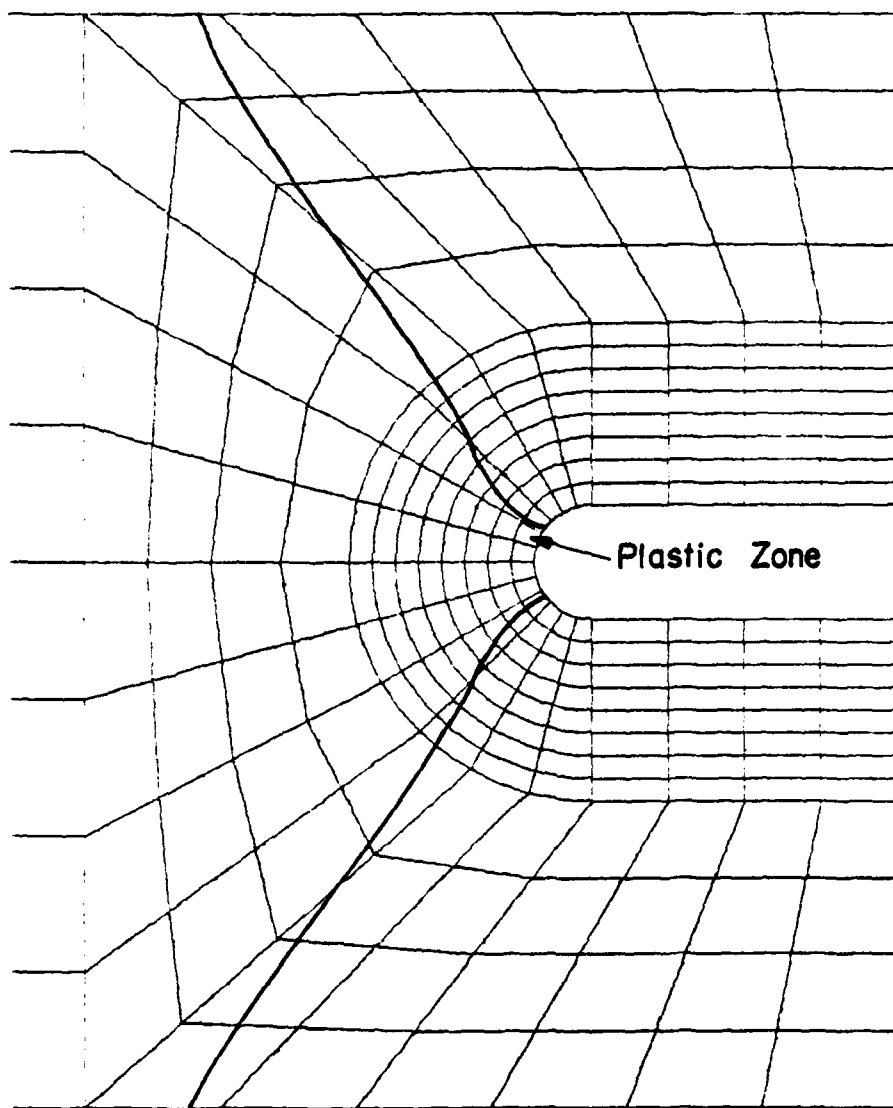


FIGURE 40

RIGID-PERFECTLY-PLASTIC FINAL PLASTIC ZONE

TABLE I. MTS AND REIHLE 5 GAGE TEST RESULTS

All Strains are  $10^{-6}$  in/in

MTS Test Machine

LOAD lbs	STRAINS				
	$\epsilon_1$	$\epsilon_2$	$\epsilon_3$	$\epsilon_4$	$\epsilon_5$
1000	-3	421	814	1235	1638
2000	338	1003	1619	2279	2903
3000	960	1714	2431	3186	3908

REIHLE Test Machine

LOAD lbs	STRAINS				
	$\epsilon_1$	$\epsilon_2$	$\epsilon_3$	$\epsilon_4$	$\epsilon_5$
1000	800	763	755	741	726
2000	1564	1540	1546	1543	1535
3000	2370	2334	2363	2377	2388

TABLE II. MTS SPECIMEN A TEST RESULTS

Cross-section =  $0.03975 \text{ in}^2$

Load, lbs.	Strain, $\epsilon, 10^{-6} \text{ in/in}$
256	615
503	1204
750	1801
1005	2423
1255	3042
1505	3665
1778	4352
2003	4925
2252	5587
2508	6355
2755	7365
2905	8230
2984	9045
3037	10150



TABLE III. MTS SPECIMEN B TEST RESULTS

Cross-section =  $0.03975 \text{ in}^2$

Load, lbs.	Strain, $\epsilon$ , $10^{-6} \text{ in/in}$
500	1250
750	1660
1000	2450
1250	3060
1500	3700
1750	4320
2000	4950
2250	5630
2500	6390
2750	7340
2900	8200
3000	9250
3050	10200
3100	11550
3125	12800

TABLE IV. MTS SPECIMEN C TEST RESULTS

Cross-section =  $0.03975 \text{ in}^2$

Load, lbs.	Strain, $\epsilon, 10^{-6} \text{ in/in}$
272	253
500	1220
762	1830
1008	2451
1231	3005
1506	3690
1755	4313
2000	4940
2255	5607
2503	6325
2750	7230
2900	8120
3000	9200
3060	10500
3100	11500
3125	12500

TABLE V.

## REIHLE SPECIMEN TEST RESULTS

Cross-section =  $0.12 \text{ in}^2$ 

LOAD lbs.	STRAIN, $\epsilon_1$ $10^{-6} \text{ in/in}$	STRAIN, $\epsilon_2$ $10^{-6} \text{ in/in}$
500	345	-115
1000	730	-240
1500	1114	-364
2000	1501	-492
2500	1895	-614
3000	2291	-745
3500	2705	-871
4000	3095	-999
4500	3515	-1126
5000	3912	-1255
5500	4332	-1384
6000	4750	-1516
6500	5199	-1647
7000	5595	-1784
7500	6075	-1923
8000	6649	-2103
8500	7285	-2275
9000	8663	-2638
9500	12245	-4435

TABLE VI.

 $\lambda = 0.2$  HOWLAND DATA

DISTANCE FROM HOLE in.	$\sigma_{\theta} / \sigma_{\infty}$
0.0	3.14
0.5	1.57
1.0	1.25
1.5	1.16
2.0	1.11
2.5	1.07
3.0	1.05
3.5	1.01
4.0	0.97

TABLE VII.

 $\lambda = 0.25$  HOWLAND DATA

DISTANCE FROM HOLE in.	$\sigma_{\theta} / \sigma_{\infty}$
0.000	3.23
0.422	1.75
0.828	1.30
1.234	1.22
1.641	1.13
2.047	1.07
2.453	1.04
2.859	0.97
3.063	0.95

TABLE VIII.

 $\lambda = 0.2$  FEA RESULTS - NODAL OUTPUT $\sigma_{\infty} = 5000$  psi

DISTANCE FROM HOLE, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	15938.2	351.7
0.25	10259.7	2375.8
0.50	7875.7	1782.5
0.75	6877.0	1673.1
1.00	6307.4	1334.7
1.25	5995.4	1087.2
1.50	5797.7	874.4
1.75	5633.9	700.5
2.00	5509.8	556.8

TABLE IX.

 $\lambda = 0.2$  FEA RESULTS - GAUSS OUTPUT $\sigma_{\infty} = 5000$  psi

DISTANCE FROM HOLE, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	15595.4	138.9
0.25	9482.6	2104.1
0.50	7953.8	1826.6
0.75	6783.7	1650.4
1.00	6316.1	1348.7
1.25	5975.6	1088.9
1.50	5795.8	890.1
1.75	5623.8	695.7
2.00	5499.9	554.9

TABLE X.

 $\lambda = 0.25$  FEA RESULTS - NODAL OUTPUT

DISTANCE FROM HOLE, in.	$\sigma_{\infty} = 4500$ psi	
	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	14745.5	229.8
0.25	9447.0	2181.1
0.50	7232.6	1599.3
0.75	6304.7	1473.5
1.00	5771.7	1139.6
1.25	5471.2	891.6
1.50	5266.9	671.0
1.75	5091.7	489.1
2.00	4940.2	354.5

TABLE XI.

 $\lambda = 0.25$  FEA RESULTS - GAUSS OUTPUT

DISTANCE FROM HOLE, in.	$\sigma_{\infty} = 4500$ psi	
	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	14422.8	121.2
0.25	8721.2	1927.1
0.50	7317.1	1639.7
0.75	6218.6	1452.4
1.00	5779.6	1151.5
1.25	5453.3	891.0
1.50	5267.1	677.3
1.75	5085.9	486.9
2.00	4937.9	343.8

TABLE XII. SHALLOW NOTCH FEA LINEAR RESULTS - NODAL

$$\sigma_n = 4225 \text{ psi}$$

DISTANCE FROM NOTCH, in.	$\sigma_\theta$ , psi	$\sigma_r$ , psi
0.00	11584.9	94.5
0.25	8966.1	1279.9
0.50	7335.6	1501.3
0.75	6347.3	1731.7
1.00	5655.0	1713.2
1.25	5184.1	1703.5
1.50	4832.4	1626.7
1.75	4564.7	1550.6
2.00	4361.1	1484.7

TABLE XIII. SHALLOW NOTCH FEA LINEAR RESULTS - GAUSS

$$\sigma_n = 4225 \text{ psi}$$

DISTANCE FROM NOTCH, in.	$\sigma_\theta$ , psi	$\sigma_r$ , psi
0.00	11530.2	11.5
0.25	8727.1	1177.7
0.50	7355.5	1504.8
0.75	6280.9	1713.6
1.00	5661.2	1720.4
1.25	5159.3	1699.7
1.50	4834.6	1634.2
1.75	4552.8	1561.7
2.00	4353.6	1486.0

TABLE XIV. DEEP NOTCH FEA LINEAR RESULTS - NODAL

$$\sigma_n = 4800 \text{ psi}$$

DISTANCE FROM NOTCH, in.	$\sigma_\theta$ , psi	$\sigma_r$ , psi
0.00	21142.8	755.2
0.25	11929.9	4424.3
0.50	8436.8	3530.4
0.75	6976.6	3324.4
1.00	6076.1	3152.2
1.25	5527.0	2993.4
1.50	5139.2	2823.4
1.75	4842.4	2412.0
2.00	4617.8	2234.2

TABLE XV. DEEP NOTCH FEA LINEAR RESULTS - GAUSS

$$\sigma_n = 4800 \text{ psi}$$

DISTANCE FROM NOTCH, in.	$\sigma_\theta$ , psi	$\sigma_r$ , psi
0.00	20368.1	314.0
0.25	10582.5	4035.2
0.50	8606.0	3611.8
0.75	6852.2	3591.9
1.00	6086.7	3172.2
1.25	5496.1	2884.7
1.50	5141.8	2643.0
1.75	4830.1	2421.2
2.00	4622.0	2238.8



TABLE XVI. SHALLOW NOTCH FEA NONLINEAR 60,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	80024.7	1094.5	-33411.4	439.1
0.25	81338.7	9523.3	-4250.9	-2690.1
0.50	81383.6	16052.6	9006.3	2188.8
0.75	65548.3	18993.2	2946.6	2133.4
1.00	58431.4	18799.6	1992.3	1796.7
1.25	52670.1	18347.9	1167.3	1501.6
1.50	49111.1	17477.1	825.7	1250.8
1.75	46080.7	16567.3	594.4	1046.7
2.00	43973.8	15667.8	471.6	885.0

TABLE XVII. SHALLOW NOTCH FEA NONLINEAR 65,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	80890.4	1004.6	-42092.3	224.7
0.25	82939.3	10229.5	-9796.3	-2782.0
0.50	83642.7	16054.6	6944.3	-711.2
0.75	74622.0	21475.5	7507.7	3214.2
1.00	64770.4	21171.3	3820.9	2838.7
1.25	57891.2	20593.6	2202.8	2419.0
1.50	53766.7	19537.0	1524.5	2018.5
1.75	50321.1	18474.8	1089.5	1710.1
2.00	47948.2	17421.4	853.7	1448.0

TABLE XVIII. SHALLOW NOTCH FEA NONLINEAR 70,000lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	81957.9	704.5	-50670.6	63.9
0.25	82822.6	10668.8	-16176.6	-3285.1
0.50	84007.3	15514.4	-376.9	-2366.8
0.75	87017.4	22662.9	15132.6	3762.3
1.00	71318.0	23765.2	5681.1	3983.0
1.25	63762.9	23215.2	3763.1	3616.6
1.50	58941.5	21920.1	2669.9	3030.0
1.75	54897.6	20670.6	1865.4	2596.7
2.00	52177.0	19426.6	1448.2	2207.0

TABLE XIX. DEEP NOTCH FEA NONLINEAR 30,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	83521.0	2332.4	-14138.0	-725.2
0.25	58038.6	20971.3	5680.1	1402.5
0.50	43309.1	18640.5	385.7	824.6
0.75	34402.6	17839.1	198.3	468.9
1.00	30522.9	16076.1	125.0	300.5
1.25	27543.7	14574.8	88.9	215.3
1.50	25757.0	13328.7	67.3	160.7
1.75	24188.8	12195.0	53.5	126.3
2.00	23141.6	11265.7	44.9	100.9

TABLE XX. DEEP NOTCH FEA NONLINEAR 35,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	79020.4	1258.1	-39063.2	-1909.2
0.25	78215.7	24598.9	16348.7	1113.1
0.50	50094.6	22675.1	105.7	1699.5
0.75	40516.0	21359.5	624.1	1002.8
1.00	35843.1	19156.9	380.6	716.7
1.25	32290.3	17299.1	257.8	521.7
1.50	30172.9	15782.1	203.5	405.9
1.75	28314.8	14410.8	156.9	319.5
2.00	27079.2	13294.4	133.7	261.5

TABLE XXI. DEEP NOTCH FEA NONLINEAR 40,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.00	80929.5	835.4	-52900.9	-2121.6
0.25	89730.7	24933.8	20503.1	-2132.3
0.50	56572.6	26815.4	-610.4	2977.0
0.75	46984.6	24878.1	1468.2	1643.1
1.00	41400.6	22286.3	610.0	1216.6
1.25	37164.5	20068.3	555.9	892.7
1.50	34690.9	18287.2	438.5	710.5
1.75	32512.6	16672.2	330.7	564.8
2.00	31078.3	15368.8	282.0	470.8

TABLE XXII.

RIGID - PERFECTLY - PLASTIC RESULTS

$$\sigma_y = 73,000 \text{ psi}$$

DISTANCE FROM NOTCH, in.	$\sigma_\theta$ , psi	$\sigma_r$ , psi
0.00	73334.4	519.9
0.25	81780.5	20863.6
0.50	83252.6	29813.6
0.75	84074.1	37321.1
1.00	84103.1	40244.0
1.25	84091.8	42244.0
1.50	84051.5	43249.8
1.75	84005.0	44032.9
2.00	83958.1	43875.6

TABLE XXIII. EXPERIMENTAL DATA  $\lambda = 0.25$  HOLE LINEAR LOADING

$$\sigma_\infty = 10,749 \text{ psi}$$

DISTANCE FROM HOLE, in.	$\sigma_\theta$ , psi	$\sigma_r$ , psi
0.000	35321.0	49.5
0.125	24639.0	-1650.5
0.250	19809.5	669.5
0.375	17799.0	2311.5
0.500	16588.0	3422.0

TABLE XXIV. EXPERIMENTAL DATA SHALLOW NOTCH LINEAR LOADING

15,000 lb Load

DISTANCE FROM HOLE, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.000	26511.7	53.2
0.125	22295.0	1392.6
0.250	19999.3	2352.7
0.375	17277.5	2387.6
0.500	15798.7	2880.5
0.625	13679.1	1929.3

TABLE XXV. EXPERIMENTAL DATA SHALLOW NOTCH 60,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.000	81111.5	5551.0	-23842.0	4.0
0.125	79710.7	6735.8	-19837.0	-475.0
0.250	78112.0	8141.5	-16409.0	-1044.0
0.375	77189.3	11899.5	-12620.0	-1426.0
0.500	71207.2	10509.4	-12319.0	-5008.0
0.625	71436.4	19260.0	-10908.0	-6453.0
0.750	63045.8	16580.9		
0.875	59477.4	17748.6		
1.000	57146.4	19275.4		

TABLE XXVI. EXPERIMENTAL DATA SHALLOW NOTCH 65,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.000	83073.9	7053.0	-36745.0	-191.0
0.125	80957.6	6078.9	-32543.0	-387.0
0.250	79854.9	7468.1	-29256.0	-1554.0
0.375	79482.8	10878.9	-23914.0	-3381.0
0.500	79638.4	16634.0	-20192.0	-2686.0
0.625	77887.0	19867.6	-16300.0	-3392.0
0.750	72556.4	19706.7		
0.875	64863.2	17433.1		
1.000	61648.0	19034.1		

TABLE XXVII. EXPERIMENTAL DATA SHALLOW NOTCH 70,000 lb LOAD

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi	NO LOAD RESIDUALS	
			$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.000	84748.9	8569.0	-50791.0	43.0
0.125	83492.9	9073.9	-46086.0	-736.0
0.250	84258.3	13718.2	-40774.0	-878.0
0.375	82846.8	13815.0	-34600.0	-1016.0
0.500	82524.6	16814.9	-30017.0	-1316.0
0.625	80446.4	17441.6	-24798.0	-1551.0
0.750	80455.5	23190.4		
0.875	76568.9	22939.7		
1.000	72264.1	23604.0		

TABLE XXVIII. EXPERIMENTAL DATA DEEP NOTCH LINEAR LOADING

15,000 lb Load

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.000	45907.0	-45.0
0.125	35372.0	7693.0
0.250	30410.0	11671.0
0.375	26177.0	12395.0
0.500	20939.0	9879.0
0.625	18373.0	9060.0
0.750	16891.0	8984.0

TABLE XXIX. EXPERIMENTAL DATA DEEP NOTCH 30,000 lb LOAD

Elastic - Plastic

DISTANCE FROM NOTCH, in.	$\sigma_{\theta}$ , psi	$\sigma_r$ , psi
0.000	76156.6	2587.4
0.125	63996.2	6057.7
0.250	50047.3	10456.3
0.375	43992.3	14485.8
0.500	39748.6	16433.6
0.625	35030.8	15899.3
0.750	33286.8	16720.0

# APPENDIX A

PSAP1 JCL

```

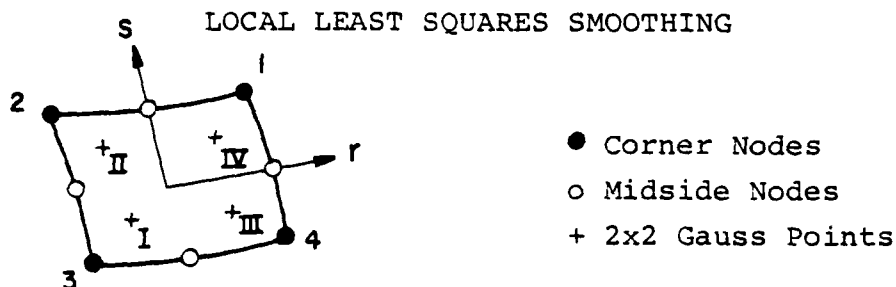
----- STANDARD JJB CARD -----
// EXEC FRTXCLGP
// FORT.SYSPRINT DD DUMMY
// FORT.SYSIN DD UNIT=3330,VOL=SER=DISK02,
// DSN=S2939.PSAP(PSAP),DISP=SHR,LABEL=(, , , IN)
// DD UNIT=3330,VOL=SER=DISK02,DSN=S2939.PSAP(PLOT),
// DISP=SHR,LABEL=(, , , IN)
// DD UNIT=3330,VOL=SER=DISK02,DSN=S2939.PSAP(INIT),
// DISP=SHR,LABEL=(, , , IN)
// DD UNIT=3330,VOL=SER=DISK02,DSN=S2939.PSAP(ELER),
// DISP=SHR,LABEL=(, , , IN)
// DD UNIT=3330,VOL=SER=DISK02,DSN=S2939.PSAP(SAPF),
// DISP=SHR,LABEL=(, , , IN)
// DD UNIT=3330,VOL=SER=DISK02,DSN=S2939.PSAP(ADNA),
// DISP=SHR,LABEL=(, , , IN)
// DD UNIT=3330,VOL=SER=DISK02,DSN=S2939.PSAP(AUXL),
// DISP=SHR,LABEL=(, , , IN)
// DD UNIT=3330,VOL=SER=DISK02,DSN=S2939.PSAP(ADPT),
// DISP=SHR,LABEL=(, , , IN)
// DD *
C *** MAIN PROGRAM ***
    DIMENSION ZZZ(3203),DISPD(5,3,800)
    CALL PSAP1(ZZZ,3203,DISPD,800)
    STOP
    END
***** DELIMITER CARD (/*) *****
//GO.FT10F001 DD UNIT=SYSDA,DISP=(,PASS),
// SPACE=(CYL,(2,2)),DSN=ETEMPL1
//GO.SYSIN DD *
***** INSERT PSAP1 DATA HERE *****
***** DELIMITER CARD (/*) *****

```



# APPENDIX B

## LOCAL LEAST SQUARES SMOOTHING



Two-Dimensional Isoparametric Element from ADINA [Ref. 4]

The local smoothing expression from Hinton and Campbell [Ref. 19] in ADINA coordinates becomes

$$\begin{Bmatrix} \tilde{\sigma}_1 \\ \tilde{\sigma}_2 \\ \tilde{\sigma}_3 \\ \tilde{\sigma}_4 \end{Bmatrix} = \begin{bmatrix} C & B & B & A \\ B & A & C & B \\ A & B & B & C \\ B & C & A & B \end{bmatrix} \times \begin{Bmatrix} \sigma_I \\ \sigma_{II} \\ \sigma_{III} \\ \sigma_{IV} \end{Bmatrix}$$

where  $A = 1 + \frac{\sqrt{3}}{2}$ ,  $B = -\frac{1}{2}$  and  $C = 1 - \frac{\sqrt{3}}{2}$ .

With  $\tilde{\sigma}_1$ ,  $\tilde{\sigma}_2$ ,  $\tilde{\sigma}_3$  and  $\tilde{\sigma}_4$  representing the smoothed corner node stresses and  $\sigma_I$ ,  $\sigma_{II}$ ,  $\sigma_{III}$ , and  $\sigma_{IV}$  as the unsmoothed stresses at the Gauss integration points, this expression can be written in an equivalent form.

$$\begin{Bmatrix} \tilde{\sigma}_3 \\ \tilde{\sigma}_4 \\ \tilde{\sigma}_1 \\ \tilde{\sigma}_2 \end{Bmatrix} = \begin{bmatrix} A & B & C & B \\ B & A & B & C \\ C & B & A & B \\ B & C & B & A \end{bmatrix} \times \begin{Bmatrix} \sigma_I \\ \sigma_{III} \\ \sigma_{IV} \\ \sigma_{II} \end{Bmatrix}$$

AD-A102 665

NAVAL POSTGRADUATE SCHOOL MONTEREY CA  
AN ELASTIC-PLASTIC FINITE ELEMENT ANALYSIS OF NOTCHED ALUMINUM --ETC(U)  
MAR 81 M J KAISER

F/G 11/6

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The midside node stress values may be obtained by averaging the values at the associated corner nodes, since the distribution of the smoothed stresses is linear along the sides of the element. Smoothed stress values obtained by this least squares method should subsequently be averaged to obtain unique values at nodal points shared by adjacent elements.

# APPENDIX C

## ADINA JCL

```

----- STANDARD J33 CARD -----
// EXEC FORTXCLG,REGION=2007K
//FORT.SYSPRINT DD DJMYY
//FORT.SYSIN DD *
      IMPLICIT REAL*8(A-H,O-Z)
      REAL A
      COMMON A(120010)
      MAX=120000
      CALL EXEC(MAX)
      STOP
      END
***** DELIMITER CARD (/*) *****
//LKED.JSDD DD DISP=SCR,DSN=MS.S2939.ADINA
//LKED.SYSIN DD *
      INCLUDE US03(LQADM)
      ENTRY MAIN
***** DELIMITER CARD (/*) *****
//GO.FT07F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=4000),SPACE=(CYL,(5,1))
//GO.FT01F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=4000),SPACE=(CYL,(5,1))
//GO.FT02F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=4000),SPACE=(CYL,(5,1))
//GO.FT03F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=1000),SPACE=(CYL,(5,1))
//GO.FT04F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=1000),SPACE=(CYL,(5,1))
//GO.FT08F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=4000),SPACE=(CYL,(5,1))
//GO.FT09F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=4000),SPACE=(CYL,(5,1))
//GO.FT10F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=4000),SPACE=(CYL,(5,1))
//GO.FT11F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=1000),SPACE=(CYL,(5,1))
//GO.FT12F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=1000),SPACE=(CYL,(5,1))
//GO.FT13F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=1000),SPACE=(CYL,(5,1))
//GO.FT56F001 DD UNIT=3330,VOL=SER=DISK01,
//   DSN=F0099.TEMP,DISP=SHR,LABEL=(, ,IN)
//GO.FT57F001 DD UNIT=SYSDA,DISP=(NEW,DELETE),
//   DCB=(RECFM=VBS,BLKSIZE=1000),SPACE=(CYL,(5,1))
//GO.SYSIN DD *
***** INSERT ADINA DATA HERE *****
----- BLANK CARD -----
----- BLANK CARD ----- (TWO BLANK CARDS STOP EXEC)
***** DELIMITER CARD (/*) *****

```

# APPENDIX D PSAP1 LISTING

```

PSAP1      MAR 1981  AS MODIFIED BY LCDR M.J. KAISER
SUBROUTINE PSAP1 DOCUMENTATION

      DESCRIPTION OF INPUT DATA CARDS

TITLE CARD - 80 ALPHANUMERIC CHARACTERS OF GRAPH TITLE INFORMATION
              TO BE PRINTED ABOVE AND BELOW THE GRAPH. THE FIRST 40
              CHARACTERS WILL FORM THE FIRST TITLE LINE. THE LAST 40
              THE SECOND LINE.

NAMELIST OPTION - CONTAINS VALUES TO VERIFY STORAGE IN BLANK
                  COMMON AND CONTROL VALUES NEEDED BY THE PROGRAM.

      THE FOLLOWING VALUES ARE INCLUDED---

NNDEST = ESTIMATE NUMBER OF GRID POINTS TO BE USED.  VALUE MUST
          BE GREATER THAN OR EQUAL TO THE ACTUAL NUMBER OF GRID
          POINTS.
          ** DEFAULT = 200 **
NUDISP = 0 FOR NO DISPLACEMENT DATA IN X-DIRECTION.
          ** DEFAULT = 1 **
NVDISP = 0 FOR NO DISPLACEMENT DATA IN Y-DIRECTION.
          ** DEFAULT = 1 **
NWDISP = 0 FOR NO DISPLACEMENT DATA IN Z-DIRECTION.
          ** DEFAULT = 1 **

KGEOM SPECIFIES SUBROUTINE AND CORRESPONDING METHOD OF INPUT
FOR MODEL GEOMETRY.
KGEOM = 1 FOR USER SUPPLIED SUBROUTINE - GEOM1
        = 2 FOR USER DEVELOPED TO READ ADINA GEOMETRY DATA - MAR 77
        = 9 FOR USER SUPPLIED SUBROUTINE - GEOM2
          FOR SAP IV DATA DECK INPUT SUBROUTINE - GEOM9.
          ** GEOM9 READS SAP IV GEOMETRY DATA - MODIFIED MAR 77
          ** DEFAULT = 9 **
KDATA SPECIFIES SUBROUTINE AND CORRESPONDING METHOD OF INPUT
FOR DISPLACEMENT DATA.
KDATA = 1 FOR SUBROUTINE DATA1 TO READ IN DISPLACEMENT DATA
          -- SUPPLIED BY THE USER.

```

APR1981  
DDCU0020  
DDCU0030  
DDCU0040  
DDCU0050  
DDCU0060  
DDCU0070  
DDCU0080  
DDCU0090  
DDCU0100  
DDCU0110  
DDCU0120  
DDCU0130  
DDCU0140  
DDCU0150  
DDCU0160  
DDCU0170  
DDCU0180  
DDCU0190  
DDCU0200  
DDCU0210  
DDCU0220  
DDCU0230  
DDCU0240  
DDCU0250  
DDCU0260  
DDCU0270  
DDCU0280  
DDCU0290  
DDCU0300  
DDCU0310  
DDCU0320  
DDCU0330  
DDCU0340  
DDCU0350  
DDCU0360  
DDCU0370  
DDCU0380  
DDCU0390  
DDCU0400  
DDCU0410  
DDCU0420  
DDCU0430  
DDCU0440  
DDCU0450  
DDCU0460  
DDCU0470  
DDCU0480

CC

```

= 5 FOR SUBROUTINE DAT5 TO READ IN DISPLACEMENT DATA
-- SUPPLIED BY THE USER.
= 9 FOR SUBROUTINE DATA9 TO READ SAP IV DATA.
** DEFAULT = 9 **
NVALUS - NOT USED AT NPS ----- ALLOW DEFAULT

** DE=AU LT = 0 **
IRESEQ - NOT USED AT NPS ----- ALLOW TO DEFAULT
** DEFAULT = 1 **
KPL0T SPECIFIES THE TYPE OF OUTPUT DEVICE TO BE USED.
KPL0T = 1 FOR CALCOMP.
= 2 FOR LANGLEY RESEARCH CENTER USE ONLY
= 3 FOR LRC USE ONLY.
= 4 FOR LRC USE ONLY
** DEFAULT = 1 **
YSPACE = SPACE BETWEEN PLOTS IN Y DIRECTION (INCHES) WHEN
MULTIPLE PLOTS ARE PRODUCED. YSPACE/2.0 IS SPACE
BETWEEN TITLE BLOCK AND PLOT.
** DEFAULT = 2.0 **
PSIZE = PAPER SIZE IN X DIRECTION. USED IN SCALING OF
PLOTS TO INSURE THIS DIMENSION IS NOT EXCEEDED.
** DEFAULT = 9.0 **
IDCASE = 0 FOR NO TITLE CARD PRECEDING
DECKS OF DISPLACEMENT VALUES.
= 1 FOR TITLE CARD PRECEDING
DECKS OF DISPLACEMENT VALUES.
** DEFAULT = 0 **

```

MODEL GEOMETRY IS NOW INPUT IN ONE OF THE FOLLOWING FORMS,  
DEPENDING ON THE VALUE OF KGEOM SPECIFIED IN NAMELIST OPTION.

```

USE IF KGEOM = 1
CALL SUBROUTINE GEOM1 WHICH READS ADINA GEOMETRY DATA

USE IF KGEOM = 2
CALL SUBROUTINE GEOM2 WHICH IS PREPARED BY THE USER TO
READ GEOMETRY DATA.

```

```
USE IF KGEOM = 9
CALL SUBROJTINE GEOM9 WHICH READS SAP IV GEOMETRY DATA.
```

0000CU0970  
0000CU0980  
0000CU0990  
0000CU1000  
0000CU1010  
0000CU1020  
0000CU1030  
0000CU1040  
0000CU1050  
0000CU1060  
0000CU1070  
0000CU1080  
0000CU1090  
0000CU1100  
0000CU1110  
0000CU1120  
0000CU1130  
0000CU1140  
0000CU1150  
0000CU1160  
0000CU1170  
0000CU1180  
0000CU1190  
0000CU1200  
0000CU1210  
0000CU1220  
0000CU1230  
0000CU1240  
0000CU1250  
0000CU1260  
0000CU1270  
0000CU1280  
0000CU1290  
0000CU1300  
0000CU1310  
0000CU1320  
0000CU1330  
0000CU1340  
0000CU1350  
0000CU1360  
0000CU1370  
0000CU1380  
0000CU1390  
0000CU1400  
0000CU1410  
0000CU1420  
0000CU1430  
0000CU1440

**CASE IDENTIFICATION CARD.**

THIS CARD IS OMITTED IF IOCASE=0 IS SPECIFIED IN &OPTION  
IF PRESENT, THIS CARD CONTAINS ANY DESIRED ALPHANUMERIC  
INFORMATION, IN COLUMNS 1-80 WILL NOT APPEAR ON PLOT BUT WILL  
APPEAR ON PRINTOUT ABOVE DISPLACEMENT DATA

DATA TO BE PLOTTED IS NOW INPUT IN ONE OF THE FOLLOWING FORMS,  
DEPENDING ON THE VALUE OF KDATA SPECIFIED IN NAMELIST OPTION.

USE IF KDATA = 1  
CALL SUBROUTINE DATA WHICH IS PREPARED BY THE USER

USE IF KDATA = 5  
CALL SUBROUTINE DATAS WHICH IS PREPARED BY THE USER

```
USE IF KDATA = 9
```

CALL SUBROUTINE DATA9 WHICH READS SAP IV DISPLACEMENT DATA.  
A DISPLACEMENT DATA DECK CAN BE PREPARED FOR ADINA IN A  
FORMAT COMPATIBLE WITH DATA9.

**NAMELIST PICT - CONTAINS VALUES NEEDED TO GENERATE PLOTS.**

THE FOLLOWING VALUES ARE INCLUDED---

```

KXHORZ = INTEGER DESIGNATING HORIZONTAL AXIS OF VIEWING PLANE,
      WHERE 1=X, 2=Y, 3=Z.
** DEFAULT = 1 **
KXVERT = INTEGER DESIGNATING VERTICAL AXIS OF VIEWING PLANE,
      WHERE 1=X, 2=Y, 3=Z.
** DEFAULT = 2 **
PHI = ANGULAR ROTATION OF MODEL ABOUT ITS X-AXIS, IN DEGREES
      (MUST BE TAKEN THIRD).
** DEFAULT = 0.0 **
THETA = ANGULAR ROTATION OF MODEL ABOUT ITS Y-AXIS, IN DEGREES
      (MUST BE TAKEN SECOND).
** DEFAULT = 0.0 **
PSI = ANGULAR ROTATION OF MODEL ABOUT ITS Z-AXIS, IN DEGREES

```

[illegible]





```

CCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCCC
KSYMYZ = 1 FOR SYMMETRY ABOUT Y-Z PLANE.
** DEFAULT = 0 **
XXMAX, YYMAX, ZZMAX, XXMIN, YYMIN, ZZMIN LOCATE CUTTING PLANES
PARALLEL TO PRINCIPAL (X-Y, X-Z, Y-Z) PLANES
TO LIMIT PLOT.
** DEFAULT XXMAX=YYMAX=ZZMAX=1.0E+20 **
** DEFAULT XXMIN=YYMIN=ZZMIN=-1.0E+20 **
NDMAX = MAXIMUM GRID PT. TO BE INCLUDED IN PLOT.
** DEFAULT = 999999999 **
NDMIN = MINIMUM GRID PT. TO BE INCLUDED IN PLOT.
** DEFAULT = 0 **
NELMAX = MAXIMUM ELEMENT NUMBER TO BE INCLUDED IN PLOT.
** DEFAULT = 999999999 **
NELMIN = MINIMUM ELEMENT NUMBER TO BE INCLUDED IN PLOT.
** DEFAULT = 0 **
CODE SPECIFIES CONTROL OPTION AFTER PLOT IS COMPLETE.
CODE = 0, LAST PLOT, EXIT FROM PROGRAM.
= 1, READ ANOTHER NAMELIST PICT.
= 2, READ A NEW SET OF DISPLACEMENT DATA, INCLUDING A
= 3, CASE IDENTIFICATION CARD IF PRESENT.
= 3, READ A COMPLETE NEW SET OF INPUT DATA,
** INCLUDING A TITLE CARD.
** DEFAULT = 0 **

THE ABOVE COMPRISES A COMPLETE BASIC SET OF INPUT DATA IF
CODE = 0 IN &PICT. FOR CODE = 1, 2, OR 3, ADDITIONAL SECTIONS OF
THE BASIC DECK MUST BE REPEATED. THE DECK MUST END WITH
NAMELIST &PICT HAVING CODE = 0.
*****

SUBROUTINE PSAP1 IS A MODIFICATION TO NAVAL POSTGRADUATE
SCHOOL THESIS BY LT. D. M. LOSH, DECEMBER 1976. MODIFICATION
INCLUDED SAP IV 8-21 NODE BRICK ELEMENTS, BOUNDARY ELEMENTS AND
ADINA TRUSS, PLANE, BRICK, BEAM ELEMENTS, AND OTHER MINOR
IMPROVEMENTS.

MODIFIED BY ADRIAN E. KIBLER JR.
LT USN
NAVAL POSTGRADUATE SCHOOL
MONTEREY, CA.
JAN - JUN 1977
*****
DDCU1930
DDCU1940
DDCU1950
DDCU1960
DDCU1970
DDCU1980
DDCU1990
DDCU2000
DDCU2010
DDCU2020
DDCU2030
DDCU2040
DDCU2050
DDCU2060
DDCU2070
DDCU2080
DDCU2090
DDCU2100
DDCU2110
DDCU2120
DDCU2130
DDCU2140
DDCU2150
DDCU2160
DDCU2170
DDCU2180
DDCU2190
DDCU2200
DDCU2210
DDCU2220
DDCU2230
DDCU2240
DDCU2250
DDCU2260
DDCU2270
DDCU2280
DDCU2290
DDCU2300
DDCU2310
DDCU2320
DDCU2330
DDCU2340
DDCU2350
DDCU2360
DDCU2370
DDCU2380
DDCU2390
DDCU2400
DDCU2410
DDCU2420
DDCU2430
DDCU2440
DDCU2450
DDCU2460
DDCU2470
DDCU2480
DDCU2490
DDCU2500
DDCU2510
DDCU2520
DDCU2530
DDCU2540
DDCU2550

```





```

C *** TO ZERO NODE AND ELEMENT SJMMATION COUNTERS
C
C      ILOOP = 0
C      NNODE = 0
C      YPMAX=0.0

C *** TO DEFINE THE ORIGIN AND OPEN PLOTTING DATA SETS
C
C      CALL CALCMP
C      CONTINUE
C      REWIND 10
C      WRITE(6,8)
C      FORMAT(1H1)
C      8
C *** TO READ TITLE CARD FOR RJM
C
C      READ(5,9004,END=999) (ABCD1(I),I=1,10),(ABCD2(I),I=1,10)
C      FORMAT(20A4)
C      9004
C      WRITE(6,9006) (ABCD1(I),I=1,10),(ABCD2(I),I=1,10)
C      9006
C      FORMAT(///,20X,20A4,///)
C      CALL INITIAL

C *** TO PLOT THE TITLE CARD AT THE BEGINING OF THE PLOT
C
C      CALL CALPLT(0.0,0.62,3)
C      CALL CALPLT(0.0,0.0,2)
C      CALL CALPLT(9.0,0.0,2)
C      CALL NOTATE(0.8,0.4,0.21,ABCD1,0.0,40)
C      CALL NOTATE(0.8,0.10,0.21,ABCD2,0.0,40)
C      CALL CALPLT(0.0,1.62+YSPACE/2.0,-3)

C *** TO SET POINTERS FOR BLANK COMMON STORAGE ZZZ
C *** (WITH INTEGER NAMES OF ARRAYS USED IN CALLED SUBROUTINES)
C
C      NUMPT = 1
C      XPT = NUMPT+NNDEST
C      YPT = XPT+NNDEST
C      ZPT = YPT+NNDEST
C      UPT = ZPT+NNDEST
C      IF(NUDISP.EQ.0) VPT = UPT+1
C      IF(NUDISP.NE.0) VPT = UPT+NNDEST
C      IF(NVDISP.EQ.0) WPT = VPT+1
C      IF(NVDISP.NE.0) WPT = VPT+NNDEST
C      IF(NWDISP.EQ.0) NEND = WPT+1-1
C      IF(NWDISP.NE.0) NEND = WPT+NNDEST-1
C      WRITE(6,15) NEND
C      15
C      FORMAT(///,20X,'BLANK COMMON STORAGE ZZZ REQUIRES AT LEAST ',I6,

```

```

PSAP0310
PSAP0320
PSAP0330
PSAP0340
PSAP0350
PSAP0360
PSAP0370
PSAP0380
PSAP0390
APR1981
PSAP0410
PSAP0420
PSAP0430
PSAP0440
PSAP0450
PSAP0460
PSAP0470
PSAP0480
PSAP0490
PSAP0500
PSAP0510
PSAP0520
PSAP0530
PSAP0540
PSAP0550
APR1981
APR1981
APR1981
APR1981
APR1981
PSAP0630
PSAP0640
PSAP0650
PSAP0660
PSAP0670
PSAP0680
PSAP0690
PSAP0700
PSAP0710
PSAP0720
PSAP0730
PSAP0740
PSAP0750
PSAP0760
PSAP0770
PSAP0780
PSAP0790

```



```

CALL CALPLT(0.0,1.62,2)
CALL CALPLT(9.0,1.62,2)
CALL NOTATE(0.8,1.31,.21,ABCD1,0.0,40)
CALL NOTATE(0.8,1.0,21,ABCD2,0.0,40)
CALL CALPLT(0.0,1.62,YSPACE,-3)
IF(LOOP=0)
  IF(KODE.EQ.3) GO TO 500
  WRITE(6,9008)
  FORMAT(//,5X,'TERMINATION NORMAL DUE TO KODE = 0')
  CALL PSTOP
  RETURN
999 CALL ERROR(2)
  RETURN
END
SUBROUTINE PLOTX(NUMPT,XPT,YPT,ZPT,UPT,VPT,WPT)
  * * * * *
  *** FOR GENERATING PLOTS.
  *** CALLED BY PSAP1
  * * * * *
  COMMON/CONTRL/ KGEOM,KDATA,KPLOT,KSYMXY,KSYMZX,KSYMZY,NOTAT,XLHT,
  1KHORZ,KVERT,PHI,THETA,PSI,NEWFR,ISCALE,PLOTSZ,XORGN,YORGN,
  2PSCALE,KDISP,DYAG,KODE
  COMMON/LIMITS/ XXMAX,YYMAX,ZZMAX,XXMIN,YYMIN,ZZMIN,NDMAX,NDMIN,
  1NELMAX,NELMIN
  COMMON/XYZLIM/ XYZMAX(3),XYZMIN(3)
  COMMON/CORGN/ YPMAX,YSPACE,PSIZE
  COMMON/GLOOP/ ILOOP
  COMMON/ABLK/ A(3,3)
  COMMON/KOUNT/ NNODE,NNEST,NUDISP,NWDISP
  COMMON/PDELS/ DELX,DELY
  DIMENSION NUMPT(1),XPT(1),YPT(1),ZPT(1),UPT(1),VPT(1),WPT(1)
  DIMENSION NODE(20),X(20),Y(20),Z(20),XDISP(20),YDISP(20),
  1ZDISP(20),XROT(20),YROT(20),XP(23),YP(23)
  * * * * *
  *** TO MAKE ALL GRID POINT NUMBERS NEGATIVE
  DO 50 I=1,NNODE
    IF(NUMPT(I).GT.0) NUMPT(I)=-NUMPT(I)
  50 CONTINUE
  * * * * *
  *** TO MAKE FRAME CHANGE IF NEWFR = 1
  *** NO FRAME CHANGE ON FIRST PLOT AFTER NAMELIST OPTION
  YMOVE=0.0
  * * * * *
  PSAP1340
  PSAP1350
  PSAP1360
  PSAP1370
  PSAP1380
  PSAP1390
  PSAP1400
  PSAP1410
  PSAP1420
  PLOT0010
  PLOT0020
  PLOT0030
  PLOT0040
  PLOT0050
  PLOT0060
  PLOT0070
  PLOT0080
  PLOT0090
  PLOT0100
  PLOT0110
  PLOT0120
  PLOT0130
  PLOT0140
  PLOT0150
  PLOT0160
  PLOT0170
  PLOT0180
  PLOT0190
  PLOT0200
  PLOT0210
  PLOT0220
  PLOT0230
  PLOT0240
  PLOT0250
  PLOT0260
  PLOT0270
  PLOT0280
  PLOT0290
  PLOT0300
  PLOT0310
  PLOT0320
  PLOT0330
  PLOT0340

```

PLOT0350  
PLOT0360  
APR1981  
PLOT0380  
PLOT0390  
APR1981  
PLOT0410  
PLOT0420  
PLOT0430  
PLOT0440  
APR1981  
PLOT0460  
PLOT0470  
PLOT0480  
PLOT0490  
PLOT0500  
PLOT0510  
PLOT0520  
PLOT0530  
PLOT0540  
PLOT0550  
PLOT0560  
PLOT0570  
PLOT0580  
PLOT0590  
PLOT0600  
PLOT0610  
PLOT0620  
PLOT0630  
PLOT0640  
PLOT0650  
PLOT0660  
PLOT0670  
PLOT0680  
PLOT0690  
PLOT0700  
PLOT0710  
PLOT0720  
PLOT0730  
PLOT0740  
PLOT0750  
PLOT0760  
PLOT0770  
PLOT0780  
PLOT0790  
PLOT0800  
PLOT0810  
PLOT0820

```

IF(ILOOP.EQ.0) GO TO 70
IF(NEWFR.EQ.1) YMOVE=YPMAX+YSPACE
70 CALL CALPLT(0.0,YMOVE,-3)
GO TO (710,710,703,710),KPL0T
703 CONTINUE
C
710 CONTINUE
IF(ISCALC.NE.0) DELX=0.0
IF(ISCALC.NE.0) DELY=0.0
IF(ISCALC.EQ.1) CALL XYSICAL
CALL CALPLT(XORGN,YORGN,-3)
XSHIFT = 0.0
YSHIFT = 0.0
ZSHIFT = 0.0
YPMAX=-1.0E20
C
C *** LOOPS TO ACCOUNT FOR SYMMETRY
C
ZSIGN = +1.0
DO 500 II=1,2
IF(II.EQ.2.AND.KSYMXY.NE.1) GO TO 500
IF(II.EQ.2.AND.KSYMXY.EQ.1) ZSIGN = -1.0
YSIGN = +1.0
DO 510 JJ=1,2
IF(JJ.EQ.2.AND.KSYMxz.NE.1) GO TO 510
IF(JJ.EQ.2.AND.KSYMxz.EQ.1) YSIGN = -1.0
XSIGN = +1.0
DO 520 KK=1,2
IF(KK.EQ.2.AND.KSYMyz.NE.1) GO TO 520
IF(KK.EQ.2.AND.KSYMyz.EQ.1) XSIGN = -1.0
C
C *** TO DETERMINE PROJECTED COORDINATES OF ELEMENTS
C
REWIND 10
100 CONTINUE
READ(10,END=1000) NEND,NUMEL,{NODE(J),J=1,NEND}
IF(NUMEL.LT.NELMIN.OR.NUMEL.GT.NELMAX) GO TO 100
DO 10 I=1,NEND
ND = NODE(I)
IF(NODE(I).EQ.0) GO TO 10
C
C *** TO MAKE GRID JOINT NUMBERS CONNECTED BY ELEMENTS POSITIVE
NUMPT(ND) = IABS(NUMPT(ND))
IF(NUMPT(ND).LT.NDMIN.OR.NUMPT(ND).GT.NDMAX) GO TO 100
10 CONTINUE
I = KHORZ
J = KVERT
DO 20 N=1,NEND

```

PLOT0830  
 PLOT0840  
 PLOT0850  
 PLOT0860  
 PLOT0870  
 PLOT0880  
 PLOT0890  
 PLOT0900  
 PLOT0910  
 PLOT0920  
 PLOT0930  
 PLOT0940  
 PLOT0950  
 PLOT0960  
 PLOT0970  
 PLOT0980  
 PLOT0990  
 PLOT1000  
 PLOT1010  
 PLOT1020  
 PLOT1030  
 PLOT1040  
 PLOT1050  
 PLOT1060  
 PLOT1070  
 PLOT1080  
 PLOT1090  
 PLOT1100  
 PLOT1110  
 PLOT1120  
 PLOT1130  
 PLOT1140  
 PLOT1150  
 PLOT1160  
 PLOT1170  
 PLOT1180  
 PLOT1190  
 PLOT1200  
 PLOT1210  
 PLOT1220  
 PLOT1230  
 PLOT1240  
 PLOT1250  
 PLOT1260  
 PLOT1270  
 PLOT1280  
 PLOT1290  
 PLOT1300

```

IF(NODE(N).EQ.0) GO TO 20
ND = NODE(N)
IF(XPT(ND).GT.XXMAX) GO TO 100
IF(XPT(ND).LT.XXMIN) GO TO 100
IF(YPT(ND).GT.YYMAX) GO TO 100
IF(YPT(ND).LT.YYMIN) GO TO 100
IF(ZPT(ND).GT.ZZMAX) GO TO 100
IF(ZPT(ND).LT.ZZMIN) GO TO 100
XDISP(N) = 0.0
YDISP(N) = 0.0
ZDISP(N) = 0.0
IF(KDISP.EQ.1.AND.NUDISP.NE.0) XDISP(N) = UPT(ND)
IF(KDISP.EQ.1.AND.NVDISP.NE.0) YDISP(N) = VPT(ND)
IF(KDISP.EQ.1.AND.NWDISP.NE.0) ZDISP(N) = WPT(ND)
X(N) = XSIGN*(XPT(ND)+XDISP(N)*DMAG+XSHIFT)/PSCALE
Y(N) = YSIGN*(YPT(ND)+YDISP(N)*DMAG+YSHIFT)/PSCALE
Z(N) = ZSIGN*(ZPT(ND)+ZDISP(N)*DMAG+ZSHIFT)/PSCALE
20 CONTINUE
IF(KDISP.EQ.2) CALL XPLOD(NEND,X,Y,Z,NODE)
XCENT = 0.0
YCENT = 0.0
FND=0.0
DO 25 N=1,NEND
  IF(NODE(N).EQ.0) GO TO 25
  XROT(N) = A(1,1)*X(N)+A(1,2)*Y(N)+A(1,3)*Z(N)
  YROT(N) = A(J,1)*X(N)+A(J,2)*Y(N)+A(J,3)*Z(N)
  IF(N.GT.8) GO TO 24
  FND=FND+1.0
  XCENT = XCENT+XROT(N)
  YCENT = YCENT+YROT(N)
24 CONTINUE
  XROT(N) = XROT(N)+DELX
  YROT(N) = YROT(N)+DELY
  IF(YROT(N).GT.YPMAX) YPMAX=YROT(N)
25 CONTINUE
  IF(NOTAT.NE.2) GO TO 29
  XCENT = XCENT/FND-(6.0/7.0)*XLHT
  YCENT = YCENT/FND-XLHT/2.0
  XCENT = XCENT+DELX
  YCENT = YCENT+DELY
  AL = NUMEL
29 CONTINUE
  IF(NOTAT.EQ.2) CALL CALNUM(XCENT,YCENT,XLHT,AL,0.0,-1)
C *** TO PLOT ELEMENTS
C
  IF(NEND.EQ.2) GO TO 280
  IF(NEND.EQ.4) GO TO 300
  
```



PLOT1310  
PLOT1320  
PLOT1330  
PLOT1340  
PLOT1350  
PLOT1360  
PLOT1370  
PLOT1380  
APR1981  
APR1981  
PLOT1410  
PLOT1420  
PLOT1430  
PLOT1440  
PLOT1450  
APR1981  
PLOT1470  
APR1981  
PLOT1490  
APR1981  
PLOT1510  
PLOT1520  
PLOT1530  
PLOT1540  
PLOT1550  
PLOT1560  
PLOT1570  
PLOT1580  
APR1981  
PLOT1600  
APR1981  
PLOT1620  
APR1981  
PLOT1640  
PLOT1650  
PLOT1660  
PLOT1670  
APR1981  
PLOT1700  
PLOT1710  
PLOT1720  
PLOT1730  
PLOT1740  
PLOT1750  
PLOT1760  
PLOT1770  
PLOT1780

```

IF(NEND.EQ.8) GO TO 320
IF(NEND.EQ.12) GO TO 340
IF(NEND.EQ.20) GO TO 340
CALL ERROR(4)

C
C
C *** TO PLOT 2 NODE ELEMENT
C
280 CONTINUE
CALL CALPLT(XROT(1),YROT(1),3)
CALL CALPLT(XROT(2),YROT(2),2)
GO TO 430

C
C *** TO PLOT 3 AND 4 NODE PLANE ELEMENT
C
300 CONTINUE
CALL CALPLT(XROT(1),YROT(1),3)
DO 305 NP=2,NEND
CALL CALPLT(XROT(NP),YROT(NP),2)
305 CONTINUE
CALL CALPLT(XROT(1),YROT(1),2)
GO TO 430

C
C *** TO PLOT 8 NODE 3-D BRICK
C
320 CONTINUE
LP=1
DO 330 NP=2,6,4
NP2=NP+2
CALL CALPLT(XROT(LP),YROT(LP),3)
DO 325 MP=NP,NP2
CALL CALPLT(XROT(MP),YROT(MP),2)
CONTINUE
CALL CALPLT(XROT(LP),YROT(LP),2)
LP=LP+4
CONTINUE
DO 335 NP=1,4
NP4=NP+4
CALL CALPLT(XROT(NP),YROT(NP),3)
CALL CALPLT(XROT(NP4),YROT(NP4),2)
CONTINUE
GO TO 430

C
C *** TO PLOT VARIABLE 4-8 NODE PLANE AND 8-20 NODE BRICK ELEMENTS
C
340 CONTINUE
LP=1
KP=8
DO 365 NP=2,6,4

```

```

NP2=NP+2
CALL CALPLT(XROT(LP),YROT(LP),3)
DO 345 MP=NP,NP2
  KP=KP+1
  N=2
  CALL CALWH(XP(1),YP(1))
  XP(2)=XROT(MP)
  YP(2)=YROT(MP)
  XP(3)=XROT(KP)
  YP(3)=YROT(KP)
  IF(NODE(KP).NE.0) CALL CURVE(XP,YP,N)
  CALL CALINE(XP,YP,N)
  CONTINUE
  KP=KP+1
  N=2
  CALL CALWH(XP(1),YP(1))
  XP(2)=XROT(LP)
  YP(2)=YROT(LP)
  XP(3)=XROT(KP)
  YP(3)=YROT(KP)
  IF(NODE(KP).NE.0) CALL CURVE(XP,YP,N)
  CALL CALINE(XP,YP,N)
  LP=LP+4
  IF(NEND.E2.12) GO TO 430
  CONTINUE
  DO 390 NP=1,4
    NP4=NP+4
    KP=NP+16
    N=2
    XP(1)=XROT(NP)
    YP(1)=YROT(NP)
    XP(2)=XROT(NP4)
    YP(2)=YROT(NP4)
    XP(3)=XROT(KP)
    YP(3)=YROT(KP)
    IF(NODE(KP).NE.0) CALL CURVE(XP,YP,N)
    CALL CALINE(XP,YP,N)
    CONTINUE
  390 CONTINUE
  430 GO TO 100
  1000 CONTINUE
  IF(KDISP.NE.3) GO TO 650
  600 CONTINUE
C *** TO PLOT VECTORS AT GRID POINTS
C
DO 601 ND=1,NNODE
IF(NUMPT(ND).LE.0) GO TO 601

```

```

PLOT1790
APR1981
PLOT1810
PLOT1820
PLOT1830
APR1981
PLOT1850
PLOT1860
PLOT1870
PLOT1880
PLOT1890
APR1981
PLOT1910
PLOT1920
PLOT1930
APR1981
PLOT1950
PLOT1960
PLOT1970
PLOT1980
PLOT1990
APR1981
PLOT2010
PLOT2020
PLOT2030
PLOT2040
PLOT2050
PLOT2060
PLOT2070
PLOT2080
PLOT2090
PLOT2100
PLOT2110
PLOT2120
PLOT2130
PLOT2140
APR1981
PLOT2160
PLOT2170
PLOT2180
PLOT2190
PLOT2200
PLOT2210
PLOT2220
PLOT2230
PLOT2240
PLOT2250
PLOT2260

```

```

IF(NUMPT(ND),LT,NDMIN,OR,NUMPT(ND),GT,NDMAX) GO TO 601
IF(XPT(ND),GT,XYZMAX(1)) GO TO 601
IF(YPT(ND),GT,XYZMAX(2)) GO TO 601
IF(ZPT(ND),GT,XYZMAX(3)) GO TO 601
X(1) = XSIGN*(XPT(ND)+XSHIFT)/PSCALE
Y(1) = YSIGN*(YPT(ND)+YSHIFT)/PSCALE
Z(1) = ZSIGN*(ZPT(ND)+ZSHIFT)/PSCALE
XDISP(1) = 0.0
YDISP(1) = 0.0
ZDISP(1) = 0.0
IF(NUDISP.NE.0) XDISP(1) = UPT(ND)
IF(NVDISP.NE.0) YDISP(1) = VPT(ND)
IF(NWDISP.NE.0) ZDISP(1) = WPT(ND)
X(2) = XSIGN*(XPT(ND)+XDISP(1)*DMAG+XSHIFT)/PSCALE
Y(2) = YSIGN*(YPT(ND)+YDISP(1)*DMAG+XSHIFT)/PSCALE
Z(2) = ZSIGN*(ZPT(ND)+ZDISP(1)*DMAG+XSHIFT)/PSCALE
I = KHRZ
J = KVERT
DO 605 N=1,2
  XROT(N) = A(I,1)*X(N)+A(I,2)*Y(N)+A(I,3)*Z(N)
  YROT(N) = A(J,1)*X(N)+A(J,2)*Y(N)+A(J,3)*Z(N)
  XROT(N) = XROT(N)+DELX
  YROT(N) = YROT(N)+DELY
CONTINUE
605 XARW = 0.06
  YARW = XARW/3.0
  CALL GARROW(XROT(1),YROT(1),XROT(2),YROT(2),1,XARW,YARW)
601 CONTINUE
650 CONTINUE
520 CONTINUE
510 CONTINUE
500 CONTINUE
C *** TO PLOT NODE POINT NUMBERS
C
C IF(NOTAT.EQ.1) CALL NDLET(NUMPT,XPT,YPT,ZPT,UPT,VPT,WPT)
C CALL CALPLT(-XROGN,-YROGN,-3)
C *** TO MAKE ALL GRID POINT NUMBERS POSITIVE AGAIN
C
C DO 1100 I=1,NNODE
C   NUMPT(I)=1ABS(NUMPT(I))
C   CONTINUE
C   RETURN
C   END
1100

```

```

PLOT2270
PLOT2280
PLOT2290
PLOT2300
PLOT2310
PLOT2320
PLOT2330
PLOT2340
PLOT2350
PLOT2360
PLOT2370
PLOT2380
PLOT2390
PLOT2400
PLOT2410
PLOT2420
PLOT2430
PLOT2440
PLOT2450
PLOT2460
PLOT2470
PLOT2480
PLOT2490
PLOT2500
PLOT2510
PLOT2520
PLOT2530
PLOT2540
PLOT2550
PLOT2560
PLOT2570
PLOT2580
PLOT2590
PLOT2600
PLOT2610
PLOT2620
PLOT2630
PLOT2640
PLOT2650
APR1981
PLOT2670
PLOT2680
PLOT2690
PLOT2700
PLOT2710
PLOT2720
PLOT2730
PLOT2740

```

```

C C C C C C C C C C
SUBROUTINE CURVE(XP,YP,N)
* * * * *
*** THIS SUBROUTINE INTERPOLATES ALONG THE EDGES OF ISOPARAMETRIC
*** ELEMENTS USING SHAPE FUNCTIONS
*** CALLED BY PLOTX
* * * * *
DIMENSION XP(1),YP(1)
DIMENSION X(3),Y(3)
DO 100 I=1,3
  X(I)=XP(I)
  Y(I)=YP(I)
CONTINUE
100 R=-1.0
DO 200 I=1,21
  YP(I)=Y(1)*R*(R-1.0)/2.0-Y(3)*(R+1.0)*(R-1.0)+Y(2)*R*(R+1.0)/2.0
  XP(I)=X(1)*R*(R-1.0)/2.0-X(3)*(R+1.0)*(R-1.0)+X(2)*R*(R+1.0)/2.0
  R=R+0.1
CONTINUE
200 N=21
RETURN
END
SUBROUTINE DSCALE(NUMPT,XPT,YPT,ZPT,UPT,VPT,WPT)
* * * * *
*** THIS SUBROUTINE DETERMINES THE SCALE FACTOR FOR DISPLACEMENTS
*** CALLED BY PSAPI
* * * * *
COMMON/CONTRL/ KGEOM,KDATA,KPLOT,KSYMXY,KSYMxz,KSYMZY,NOTAT,XLHT,
1KHORZ,KVERT,PHI,THETA,PSI,NEWFR,ISCALE,PLOTSZ,XORGN,YORGN,
2PSCALE,KDISP,DMAG,KODE
COMMON/SAVEV/ DMAGS,DMAG
COMMON/KOUNT/ NNODE,NNDIST,NUDISP,NVDISP,NWDISP
DIMENSION NUMPT(1),XPT(1),YPT(1),ZPT(1),UPT(1),VPT(1),WPT(1)
IF(KDISP.EQ.0-OR-KDISP.EQ.2) GO TO 10
GO TO (10,20), DMAG
10 CONTINUE
DMAG = DMAGS
GO TO 30
20 CONTINUE
DMAX = 0.0
DO 100 I=1,NNODE

```

PLOT2750  
 PLOT2760  
 PLOT2770  
 PLOT2780  
 PLOT2790  
 PLOT2800  
 PLOT2810  
 PLOT2820  
 PLOT2830  
 PLOT2840  
 PLOT2850  
 PLOT2860  
 PLOT2870  
 PLOT2880  
 PLOT2890  
 PLOT2900  
 PLOT2910  
 PLOT2920  
 PLOT2930  
 PLOT2940  
 PLOT2950  
 PLOT2960  
 PLOT2970  
 PLOT2980  
 PLOT2990  
 PLOT3000  
 PLOT3010  
 PLOT3020  
 PLOT3030  
 PLOT3040  
 PLOT3050  
 PLOT3060  
 PLOT3070  
 PLOT3080  
 PLOT3090  
 PLOT3100  
 PLOT3110  
 PLOT3120  
 PLOT3130  
 PLOT3140  
 PLOT3150  
 PLOT3160  
 PLOT3170  
 PLOT3180  
 PLOT3190  
 PLOT3200  
 PLOT3210  
 PLOT3220

```

IF(NUDISP.EQ.0) GO TO 500
IF(ABS(UPT(I)).GT.DMAX) DMAX = ABS(UPT(I))
500 CONTINUE
IF(NVDISP.EQ.0) GO TO 501
IF(ABS(VPT(I)).GT.DMAX) DMAX = ABS(VPT(I))
501 CONTINUE
IF(NWDISP.EQ.0) GO TO 502
IF(ABS(WPT(I)).GT.DMAX) DMAX = ABS(WPT(I))
502 CONTINUE
100 CONTINUE
30 CONTINUE
DMAG = DMAGS/DMAX
RETURN
END
SUBROUTINE ROTAT
* * * * *
*** SETS UP COEFFICIENTS OF ROTATION MATRIX
*** CALLED BY PSAP1
* * * * *
COMMON/CONTROL/ KGEOM, KDATA, KPLOT, KSYMXY, KSYMZ, KSYMZ, NOTAT, XLHT,
1KHORZ, KVERT, PHI, THETA, PSI, NEWFR, ISCALE, PLOTSZ, XORGN, YORGN,
2PSCALE, KDISP, DMAG, KODE
COMMON/ABLK/ A(3,3)
PI = 3.1415926536
SINPHI = SIN(PHI*PI/180.0)
COSPHI = COS(PHI*PI/180.0)
SINTHE = SIN(THETA*PI/180.0)
COSTHE = COS(THETA*PI/180.0)
SINPSI = SIN(PSI*PI/180.0)
COSPSI = COS(PSI*PI/180.0)
A(1,1) = COSTHE*COSPSI
A(1,2) = COSTHE*SINPSI-SINPHI-SINPSI*COSPSI
A(1,3) = SINPHI+COSPHI*SINPSI
A(2,1) = SINTHE*COSTHE
A(2,2) = SINTHE*SINPSI+SINPSI*COSPSI
A(2,3) = -SINTHE+COSPHI*SINPSI
A(3,1) = -SINTHE
A(3,2) = COSTHE*SINPHI
A(3,3) = COSTHE*COSPHI
RETURN
END
SUBROUTINE XYSCAL
* * * * *
C * * * * *

```

```

C *** TO DETERMINE SCALE FACTOR FOR MODEL GEOMETRY.
C *** CALLED BY PLOTX
C *
C * * * * *
C * COMMON/CONTRL/ KGEOM, KDATA, KPLOT, KSYMXY, KSYMXYZ, KSYMZY, NOTAT, XLHT,
C * 1KHORZ, KVERT, PHI, THETA, PSI, NEWFR, ISCALE, PLOTSZ, XDRGN, YDRGN,
C * 2PSCALE, KDISP, DMAG, KODE
C * COMMON/XYZLIM/ XYZMAX(3), XYZMIN(3)
C * COMMON/CORGN/ VPMAX, YSPACE, PSIZE
C * COMMON/ABLK/ A(3,3)
C * COMMON/PDELS/ DELX, DELY
C * I = KHORZ
C * J = KVERT
C * DMAX = 0.0
C * DO 5 N=1,3
C * VDU = ABS(XYZMAX(N)-XYZMIN(N))
C * IF(VDU .GT. DMAX) DMAX = VDU
C * 5 CONTINUE
C * PSCALE = DMAX/PLOTSZ
C * DO 10 L=1,2
C * DO 10 M=1,2
C * DO 10 N=1,2
C * X = XYZMIN(1)
C * IF(L .EQ. 2) X = XYZMAX(1)
C * Y = XYZMIN(2)
C * IF(M .EQ. 2) Y = XYZMAX(2)
C * Z = XYZMIN(3)
C * IF(N .EQ. 2) Z = XYZMAX(3)
C * XROT = A(I,1)*X+A(I,2)*Y+A(I,3)*Z
C * YROT = A(J,1)*X+A(J,2)*Y+A(J,3)*Z
C * IF(L*M*N.NE.1) GO TO 30
C * 20 CONTINUE
C * XRMIN = XROT
C * XRMAX = XROT
C * YRMIN = YROT
C * YRMAX = YROT
C * 30 CONTINUE
C * IF(XROT .GT. XRMAX) XRMAX = XROT
C * IF(XROT .LT. XRMIN) XRMIN = XROT
C * IF(YROT .GT. YRMAX) YRMAX = YROT
C * IF(YROT .LT. YRMIN) YRMIN = YROT
C * 10 CONTINUE
C * XR=ABS(XRMAX-XRMIN)
C * IF(XR/PSCALE .GT. PSIZE) PSCALE=XR/PSIZE
C * XRMAX = XRMAX/PSCALE
C * XRMIN = XRMIN/PSCALE
C * YRMIN = YRMIN/PSCALE
C * YRMAX = YRMAX/PSCALE

```

```

PLOT3720
PLOT3730
PLOT3740
PLOT3750
PLOT3770
PLOT3780
PLOT3790
PLOT3800
PLOT3810
PLOT3820
PLOT3830
PLOT3840
PLOT3850
PLOT3860
PLOT3870
PLOT3880
PLOT3890
PLOT3900
PLOT3910
PLOT3920
PLOT3930
PLOT3940
PLOT3950
PLOT3960
PLOT3970
PLOT3980
PLOT3990
PLOT4000
PLOT4010
PLOT4020
PLOT4030
PLOT4040
PLOT4050
PLOT4060
PLOT4070
PLOT4080
PLOT4090
PLOT4100
PLOT4110
PLOT4120
PLOT4130
PLOT4140
PLOT4150
PLOT4160
PLOT4170
PLOT4180
PLOT4190
PLOT4200

```









PLOT5650  
 PLOT5660  
 PLOT5670  
 PLOT5680  
 PLOT5690  
 PLOT5700  
 PLOT5710  
 PLOT5720  
 PLOT5730  
 PLOT5740  
 PLOT5750  
 PLOT5760  
 PLOT5770  
 PLOT5780  
 PLOT5790  
 PLOT5800  
 PLOT5810  
 PLOT5820  
 PLOT5830  
 APR1981  
 PLOT5850  
 PLOT5860  
 PLOT5870

```

DO 500 I=1, NNJDE
  IF (NUMPT(I).LE.0) GO TO 500
  IF (NUMPT(I).LT.NDMIN.OR.NUMPT(I).GT.NDMAX) GO TO 500
  IF (XPT(I).GT.XYZMAX(1)) GO TO 500
  IF (XPT(I).LT.XYZMIN(1)) GO TO 500
  IF (YPT(I).GT.XYZMAX(2)) GO TO 500
  IF (YPT(I).LT.XYZMIN(2)) GO TO 500
  IF (ZPT(I).GT.XYZMAX(3)) GO TO 500
  IF (ZPT(I).LT.XYZMIN(3)) GO TO 500
  X = (XPT(I)+XSHIFT)/PSCALE
  Y = (YPT(I)+YSHIFT)/PSCALE
  Z = (ZPT(I)+ZSHIFT)/PSCALE
  XROT = A(I,1)*X+A(I,2)*Y+A(I,3)*Z
  YROT = A(JJ,1)*X+A(JJ,2)*Y+A(JJ,3)*Z
  XL = XROT+XLHT/2.0
  YL = YROT+XLHT/2.0
  XL = XL+DELY
  YL = YL+DELY
  AL = NUMPT(I)
  CALL CALNUM(XL,YL,XLHT,AL,3.0,-1)
  CONTINUE
500 RETURN
END
  
```

```

SUBROUTINE INITIAL
* * * * *
*** TO SET UP VALUES FOR CONTROL PARAMETERS
*** CALLED BY PSAPI
* * * * *
COMMON/CDATA/VTIME,NTLC
COMMON/CONTRL/ KGEOM, KDATA, KPLDT, KSYMXY, KSYMxz, KSYMZY, NOTAT, XLHT,
1KHORZ, <VERT, PHI, THETA, PSI, NEWFR, ISCALF, PLOTSZ, XORG, YORG,
2PSCALE, KDISP, DMAG, KODE
COMMON/LIMITS/ XXMAX, YYMAX, ZZMAX, XXMIN, YYMIN, ZZMIN, NDMAX, NDMIN,
1NELMAX, NELMIN
COMMON/CORGN/ YPMAX, YSPACE, PSIZE
COMMON/SAVEV/ DMAGS, IDMA3
COMMON/KOUNT/ VNODE, NNDEST, NUDISP, NVDISP, NWDISP
COMMON/SEQUENCE/ IRESEQ
COMMON/VALUES/ NVALUS
COMMON/CASEID/ IDCASE
NAMELIST/OPTION/ NNDEST, NUDISP, NVDISP, NWDISP,
1KGEOM, KDATA, NVALUS, IRESEQ, KPLOT, YSPACE, PSIZE, IDCASE
*** DESCRIPTION OF VALUES IN &OPTION GIVEN IN SUBROUTINE DDCANT
C C C C C
*** TO SET DEFAULT VALUES FOR &OPTION
***
NNDEST = 200
NUDISP = 0
NVDISP = 0
NWDISP = 0
KGEOM = 9
KDATA = 9
NTIME = 0
NVALUS = 0
IRESEQ = 1
KPLOT = 1
YSPACE = 2.0
PSIZE = 9.0
IDCASE = 0
C C C C C
*** TO SET DEFAULT VALUES FOR &PICT
KHORZ = 1
C C C C C
INIT0010
INIT0020
INIT0030
INIT0040
INIT0050
INIT0060
INIT0070
INIT0080
INIT0090
INIT0100
INIT0110
INIT0120
INIT0130
INIT0140
INIT0150
INIT0160
INIT0170
INIT0180
INIT0190
INIT0200
INIT0210
INIT0220
INIT0230
INIT0240
INIT0250
INIT0260
INIT0270
INIT0280
INIT0290
INIT0300
INIT0310
INIT0320
INIT0330
INIT0340
INIT0350
INIT0360
INIT0370
INIT0380
INIT0390
INIT0400
INIT0410
INIT0420
INIT0430
INIT0440
INIT0450
INIT0460
INIT0470
INIT0480

```



ENIT0970  
 ENIT0980  
 ENIT0990  
 ENIT1000  
 ENIT1010  
 ENIT1020  
 ENIT1030  
 ENIT1040  
 ENIT1050  
 ENIT1060  
 ENIT1070  
 ENIT1080  
 ENIT1090  
 ENIT1100  
 ENIT1110  
 ENIT1120  
 ENIT1130  
 ENIT1140  
 ENIT1150  
 ENIT1160  
 ENIT1170  
 ENIT1180  
 ENIT1190  
 ENIT1200  
 ENIT1210  
 ENIT1220  
 ENIT1230  
 ENIT1240  
 ENIT1250  
 ENIT1260  
 ENIT1270  
 ENIT1280  
 ENIT1290  
 ENIT1300  
 ENIT1310  
 ENIT1320  
 ENIT1330  
 ENIT1340  
 ENIT1350  
 ENIT1360  
 ENIT1370  
 ENIT1380  
 ENIT1390  
 ENIT1400  
 ENIT1410  
 ENIT1420  
 ENIT1430  
 ENIT1440



```

16 WRITE(6,16) 5X,'GRID POINT INFORMATION',///
17 FORMAT(///, 5X,'RESEQUENCED',4X,'USER INPUT',
15X,'GRID POINT',5X,'GRID POINT',
25X,'NUMBER',9X,'X',14X,'Y',14X,'Z',///)
DO 30 I=1,NNODE
WRITE(6,18) I,NUMPT(I),XPT(I),YPT(I),ZPT(I)
30 CONTINUE
19 WRITE(6,19) 5X,'ELEMENT INFORMATION - WITH RESEQUENCED GRID POINTS'
1.///
9008 WRITE(6,9008) RESEQUENCED,4X,'USER INPUT',25X,'GRID POINTS',/
11X,'ELEMENT',8X,'ELEMENT',/
21X,'NUMBER',9X,'X',14X,'Y',14X,'Z',///
3 8 9 10 11 12 13 14 15 16 17 18 19 20 21
REWIND 10
I = 0
35 CONTINUE
I = I+1
READ(10,END=999) NEND,NUMEL,(NODE(J),J=1,NEND)
IF(NEND.EQ.12) GO TO 40
WRITE(6,9010) I,NUMEL,(NODE(J),J=1,NEND)
9010 FORMAT(IX,14,11X,14,9X,20I5)
GO TO 35
40 WRITE(6,9010) I,NUMEL,(NODE(J),J=1,4),(NODE(J),J=9,12)
GO TO 35
2000 CONTINUE
C *** FOR OUTPUT OF DISPLACEMENT DATA
C
210 WRITE(6,210)
FORMAT(///, 5X,'DISPLACEMENTS TO BE PLOTTED',///)
WRITE(6,17)
DO 230 I=1,NNODE
U = 0.0
IF(NVDISP.NE.0) U = UPT(I)
V = 0.0
IF(NVDISP.NE.0) V = VPT(I)
W = 0.0
IF(NWDISP.NE.0) W = WPT(I)
WRITE(6,18) I,NUMPT(I),U,V,W
230 CONTINUE
999 RETURN
END
SUBROUTINE ELTYPE(MTYPE,AGEOM)

```

```

INIT11930
INIT11940
INIT11950
INIT11960
INIT11970
INIT11980
INIT11990
INIT12000
INIT12010
INIT12020
INIT12030
INIT12040
INIT12050
INIT12060
INIT12070
INIT12080
INIT12090
INIT12100
INIT12110
INIT12120
INIT12130
INIT12140
INIT12150
INIT12160
INIT12170
INIT12180
INIT12190
INIT12200
INIT12210
INIT12220
INIT12230
INIT12240
INIT12250
INIT12260
INIT12270
INIT12280
INIT12290
INIT12300
INIT12310
INIT12320
INIT12330
INIT12340
INIT12350
INIT12360
INIT12370
INIT12380
INIT12390
FLER0010

```

```

CCCCCCCCCCCC
*      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
*** THIS SUBROUTINE CALLS OTHER ROUTINES TO READ ELEMENT CONNECTIVITY
*** MTYPE = ELEMENT TYPE
*** KGEOM = 1 - ADINA ELEMENTS
*** KGEOM = 2 - NONSAP ELEMENTS
*** KGEOM = 9 - SAP IV ELEMENTS
*** CALLED BY GEOM1/GEOM2/GEOM9/
*      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
*      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
IF(KGEOM.EQ.1) GO TO 20
IF(KGEOM.EQ.2) GO TO 40
GO TO (1,2,3,4,5,6,7,8,9,10,11,12),MTYPE
1 CALL TRUSS
2 GO TO 900
2 CALL BEAM
2 GO TO 900
3 CALL PLANE
2 GO TO 900
4 CALL PLANE
2 GO TO 900
5 CALL THREEED
2 GO TO 900
6 CALL SHELL
2 GO TO 900
7 CALL BNDRY
2 GO TO 900
8 CALL SOL21
2 GO TO 900
9 CALL ERROR(1)
2 GO TO 900
10 CALL ERROR(1)
2 GO TO 900
11 CALL ERROR(1)
2 GO TO 900
12 CALL ERROR(1)
2 GO TO 900
20 CONTINUE
20 GO TO (21,22,23,24),MTYPE
21 CALL ADTRUS
2 GO TO 900
22 CALL ADPLAN
2 GO TO 900
23 CALL AD3DEE
ELER0020
ELER0030
ELER0040
ELER0050
ELER0060
ELER0070
ELER0080
ELER0090
ELER0100
ELER0110
ELER0120
ELER0130
ELER0140
ELER0150
ELER0160
ELER0170
ELER0180
ELER0190
ELER0200
ELER0210
ELER0220
ELER0230
ELER0240
ELER0250
ELER0260
ELER0270
ELER0280
ELER0290
ELER0300
ELER0310
ELER0320
ELER0330
ELER0340
ELER0350
ELER0360
ELER0370
ELER0380
ELER0390
ELER0400
ELER0410
ELER0420
ELER0430
ELER0440
ELER0450
ELER0460
ELER0470
ELER0480
ELER0490

```





```

6      GO TO 1000
      CONTINUE
9006   WRITE(6,90061)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN SCL21 ,ELEMENT CARD ERROR'//)
      GO TO 1000
7      CONTINUE
9007   WRITE(6,90071)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN ADTRUS,ELEMENT CARD ERROR'//)
      GO TO 1000
8      CONTINUE
9008   WRITE(6,90081)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN ADPLAN,ELEMENT CARD ERROR'//)
      GO TO 1000
9      CONTINUE
9009   WRITE(6,90091)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN AD3DEE,ELEMENT CARD ERROR'//)
      GO TO 1000
10     CONTINUE
9010   WRITE(6,90101)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN AD8EAM,ELEMENT CARD ERROR'//)
      GO TO 1000
11     CONTINUE
9011   WRITE(6,90111)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN NSTRUS,ELEMENT CARD ERROR'//)
      GO TO 1000
12     CONTINUE
9012   WRITE(6,90121)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN NSPLAN,ELEMENT CARD ERROR'//)
      GO TO 1000
13     CONTINUE
9013   WRITE(6,90131)
      FORMAT(/,1X,'A3NORMAL TERMINATION IN NS3DEE,ELEMENT CARD ERROR'//)
      GO TO 1000
14     CONTINUE
9014   WRITE(6,90141)
      FORMAT(/,1X,'A3NORMAL TERMINATION NONSAP MESH CANNOT BE PLOTTED'//)
      GO TO 1000
15     CONTINUE
16     CONTINUE
17     CONTINUE
18     CONTINUE
19     CONTINUE
20     CONTINUE

```

```

ELER0980
ELER0990
ELER1000
ELER1010
ELER1020
ELER1030
ELER1040
ELER1050
ELER1060
ELER1070
ELER1080
ELER1090
ELER1100
ELER1110
ELER1120
ELER1130
ELER1140
ELER1150
ELER1160
ELER1170
ELER1180
ELER1190
ELER1200
ELER1210
ELER1220
ELER1230
ELER1240
ELER1250
ELER1260
ELER1270
ELER1280
ELER1290
ELER1300
ELER1310
ELER1320
ELER1330
ELER1340
ELER1350
ELER1360
ELER1370
ELER1380
ELER1390
ELER1400
ELER1410
ELER1420
ELER1430
ELER1440
ELER1450

```

```

1000 CONTINUE
      CALL PSTOP
      RETURN
      END
      SUBROUTINE GEOM9(NUMPT,XPT,YPT,ZPT,UPT,VPT,WPT)
      * * * * *
      *** GEOM9 READS SAP IV GEOMETRY DATA
      *** CALLED BY PSAP1
      * * * * *
      COMMON/CTRL/ KGEOM,KDATA,KPLOT,KSYMXY,KSYMxz,KSYMZY,NOTAT,XLHT,
      1KHORZ,KVERT,PHI,THETA,PSI,NEWFR,ISCALE,PLOTSZ,XORGN,YORGN,
      2PSCALE,KDISP,DVAG,KODE
      COMMON/KOUNT/ NNODE,NNDIST,NUDISP,NWDISP
      COMMON/GCONT/NJMNP,NPAR(20),NELTYP,NJMEL
      DIMENSION NUMPT(1),XPT(1),YPT(1),ZPT(1),UPT(1),VPT(1),WPT(1)
      DATA CTEST/0/
      * * * * *
      *** INSERT ROUTINE HERE
      * * * * *
      100 READ(5,100) HED
      FORMAT(12A6)
      * * * * *
      *** READ MASTER CONTROL CARD
      *** NUMNP = TOTAL NUMBER OF NODE POINTS
      *** NELTYP = NUMBER OF ELEMENT GROUPS
      * * * * *
      200 READ(5,200) NUMNP,NELTYP
      FORMAT(2I5)
      NNODE=NUMNP
      * * * * *
      *** READ OR GENERATE NODAL POINT DATA
      * * * * *
      NOLD=0
      10 READ(5,9006) CT,N,XPT(N),YPT(N),ZPT(N),KN
      9006 FORMAT(A1,I4,30X,3F10.0,I5)
      * * * * *
      ***CHECK FOR CYLINDRICAL COORDINATES
      * * * * *
      IF(CT.NE.CTEST) GO TO 20
      R=XPT(N)
      XPT(N)=R*SIN(ZPT(N)/57.2958)
      ZPT(N)=R*COS(ZPT(N)/57.2958)

```

```

ELER1460
ELER1470
ELER1480
ELER1490
SAPF0010
SAPF0020
SAPF0030
SAPF0040
SAPF0050
SAPF0060
SAPF0070
SAPF0080
SAPF0090
SAPF0100
SAPF0110
SAPF0120
SAPF0130
SAPF0140
SAPF0150
SAPF0160
SAPF0170
SAPF0180
SAPF0190
SAPF0200
SAPF0210
SAPF0220
SAPF0230
SAPF0240
SAPF0250
SAPF0260
SAPF0270
SAPF0280
SAPF0290
SAPF0300
SAPF0310
SAPF0320
SAPF0330
SAPF0340
SAPF0350
SAPF0360
SAPF0370
SAPF0380
SAPF0390
SAPF0400
SAPF0410
SAPF0420
SAPF0430
SAPF0440

```





```

1010 READ(5,1010) MAT,NT
      FORMAT(2I5)
      IF(NT.EQ.0) NT=1
      DO 50 K=1,NTC
1005 READ(5,1005) DUMMY
      FORMAT(10A8)
      CONTINUE
      CONTINUE
C*** READ ELEMENT LOAD FACTORS
C
1002 READ(5,1002) ((EMUL(I,J),J=1,5),I=1,4)
      FORMAT(5F10.0)
C*** READ ELEMENT PROPERTIES
C
      IF(NPAR(14).EQ.0) NPAR(14) = 1
      N=NPAR(14)-1
130 READ(5,1003) M,((IE(I),I=1,4),KG
1003 FORMAT(5I5,30X,15)
      IF(KG.EQ.0) KG=1
140 N=N+1
      IF(M.EQ.N) GO TO 145
      DO 142 I=1,4
142 IX(I)=IX(I)+KG
      GO TO 150
145 DO 148 I=1,4
148 IX(I)=IE(I)
150 CONTINUE
      I = IX(1)
      J = IX(2)
      K = IX(3)
      L = IX(4)
      NUMEL=NUMEL+1
      WRITE(10) N4,N,I,J,K,L
      IF(N.EQ.NUMEL) RETURN
      IF(N.EQ.M) GO TO 130
      GO TO 140
      END
      SUBROUTINE BEAM
      * * * * *
      *** READS SAP IV BEAM ELEMENT CARDS (ELTYPE 2)
      *** CALLED BY ELTYPE
      * * * * *

```

PPF1410  
 SAPF1420  
 SAPF1430  
 SAPF1440  
 SAPF1450  
 SAPF1460  
 SAPF1470  
 SAPF1480  
 SAPF1490  
 SAPF1500  
 SAPF1510  
 SAPF1520  
 SAPF1530  
 SAPF1540  
 SAPF1550  
 SAPF1560  
 SAPF1570  
 SAPF1580  
 SAPF1590  
 SAPF1600  
 SAPF1610  
 SAPF1620  
 SAPF1630  
 SAPF1640  
 SAPF1650  
 SAPF1660  
 SAPF1670  
 SAPF1680  
 SAPF1690  
 SAPF1700  
 SAPF1710  
 SAPF1720  
 SAPF1730  
 SAPF1740  
 SAPF1750  
 SAPF1760  
 SAPF1770  
 SAPF1780  
 SAPF1790  
 SAPF1800  
 SAPF1810  
 SAPF1820  
 SAPF1830  
 SAPF1840  
 SAPF1850  
 SAPF1860  
 SAPF1870  
 SAPF1880

```

C
COMMON/GCONT/NUMNP,NPAR(20),NELTYP,NUMEL
N2=2
NUMEPC=NPARG(3)
NUMEPC=NPARG(4) * 2
NUMFEF=NPARG(5)
NUMMAT=NPARG(6)
READ MATERIAL PROPERTY CARDS (DUMMY)
DO 10 I=1,NUMMAT
  READ(5,1001) DUMMY
  FORMAT(10A8)
1001 CONTINUE
C **
DO 20 J=1,NUMEPC
  READ(5,1001) DUMMY1
  CONTINUE
20 CONTINUE
C **
DO 30 K=1,3
  READ(5,1001) DUMMY2
  CONTINUE
30 CONTINUE
C **
DO 40 L=1,NUMFEF
  READ(5,1001) DUMMY3
  CONTINUE
40 CONTINUE
C **
IF(NPAR(14).EQ.0) NPAR(14) = 1
N=NPARG(14)
READ(5,1002) N,N,I,J
FORMAT(3I5,4X,I8)
1002 IF (KK.EQ.0) KK=1
120 IF (M.NE.N) GO TO 200
  I = I
  J = J
  KKK = KK
  CONTINUE
200 NUMEL = NUMEL+1
WRITE(10) N2,N,I,J
IF (N.EQ.NUMEL) RETURN
N = N + 1
I = I + KKK
J = J + KKK
IF (N.GT.M) GO TO 100
GO TO 120
END
SUBROUTINE THREEED
C
C
C

```









SAPF3810  
SAPF3820  
SAPF3830  
SAPF3840  
SAPF3850  
SAPF3860  
SAPF3870  
SAPF3880  
SAPF3890  
SAPF3900  
SAPF3910  
SAPF3920  
SAPF3930  
SAPF3940  
SAPF3950  
SAPF3960  
SAPF3970  
SAPF3980  
SAPF3990  
SAPF4000  
SAPF4010  
SAPF4020  
SAPF4030  
SAPF4040  
SAPF4050  
SAPF4060  
SAPF4070  
SAPF4080  
SAPF4090  
SAPF4100  
SAPF4110  
SAPF4120  
SAPF4130  
SAPF4140  
SAPF4150  
SAPF4160  
SAPF4170  
SAPF4180  
SAPF4190  
SAPF4200  
SAPF4210  
SAPF4220  
SAPF4230  
SAPF4240  
SAPF4250  
SAPF4260  
SAPF4270  
SAPF4280

C  
DIMENSION NP(20), INP(20)  
COMMON/GCONT/NJMNP,NPAR(20),NELTYP,NJMEL  
N20=20  
NSOL21=NP(2)  
NUMMAT=NP(3)  
MAXTP=NP(4)  
IF(MAXTP.EQ.0) MAXTP=1  
NORTH0=NP(5)  
NDLS=NP(6)  
MAXNOD=NP(7)  
IF(MAXNOD.EQ.0) MAXNOD=21  
IF(MAXNOD.EQ.8) N20=8  
NOPSET=NP(8)  
READ THE MATERIAL PROPERTY CARDS  
DO 50 J=1,NUMMAT  
READ(5,9002) M,NTP  
9002 FORMAT(215)  
IF(NTP.EQ.0) NTP=1  
NTP2=2\*NTP  
DO 40 JJ=1,NTP2  
READ(5,9004) DUMMY  
9004 FORMAT(20A4)  
CONTINUE  
CONTINUE  
50  
C \*\*\*  
READ MATERIAL AXES ORIENTATION SETS  
IF(NORTH0.EQ.0) GO TO 61  
DO 60 J=1,NORTH0  
READ(5,9004) DUMMY  
CONTINUE  
60  
CONTINUE  
61  
C \*\*\*  
CONTINUE  
READ DISTRIBUTED SURFACE LOAD DATA  
IF(NDLS.EQ.0) GO TO 71  
NDLS2=NDLS\*2  
DO 70 J=1,NDLS2  
READ(5,9004) DUMMY  
CONTINUE  
70  
CONTINUE  
71  
C \*\*\*  
CONTINUE  
READ STRESS OUTPUT LOCATION SETS  
IF(NOPSET.EQ.0) GO TO 81  
DO 80 J=1,NOPSET  
READ(5,9004) DUMMY  
CONTINUE  
80  
CONTINUE  
81  
C \*\*\*  
CONTINUE  
READ ELEMENT LOAD CASE MULTIPLIERS  
DO 90 J=1,5  
READ(5,9004) DUMMY  
CONTINUE  
90



```

9000 NCARD=0
      READ(5,9000) DJMMY
      FORMAT(20A4)
      READ MASTER CONTROL CARDS
      C *** NUMNP = TOTAL NUMBER OF NODE POINTS
      C *** NELTYP = NUMBER OF ELEMENT GROUPS
      C ***
9001 READ(5,9001) NUMNP, (IDOF(I), I=1,6), NEGL, MODEX, NSTE
      FORMAT(15,6I1,14,3I5)
      NELTYP=NEGL+NEGNL
      NNODE=NUMNP
9002 READ(5,9002) IMASS, IDAMP, IMASSN, IDAMPN
      FORMAT(4I5)
      READ(5,9002) IEIG
      READ(5,9002) ISREF, NUMREF, IEQUIT, ITEMEX
      READ(5,9000) DJMMY
      READ(5,9000) DUMMY
      READ(5,9000) DUMMY
      READ(5,9000) DUMMY
      C *** READ OR GENERATE NODAL POINT DATA
      NOLD=0
      NEQ=0
10   READ(5,9006) CT, N, (ID(I), I=1,6), XPT(N), YPT(N), ZPT(N), KN
      FORMAT(A1,14,1X,14,5I5,3F10.0,15)
      C *** CHECK FOR CYLINDRICAL COORDINATES
      IF(CT.NE.CTEST) GO TO 12
      DUM=ZPT(N)/57.2958
      R=YPT(N)
      YPT(N)=R*COS(ZPT(N)/57.2958)
      ZPT(N)=R*SIN(ZPT(N)/57.2958)
12   CONTINUE
      NUMPT(N)=N
      IF(NOLD.EQ.0) GO TO 50
      C *** FOR GENERATION OF FIXED BOUNDARY CONDITIONS
      DO 15 I=1,6
      IF(IDOLD(I).EQ.-1.AND.ID(I).EQ.0) ID(I)=IDOLD(I)
      CONTINUE
15   IF(KNOLD.EQ.0) GO TO 50
      NUM=(N-NOLD)/KNOLD
      NUMN=NUM-1
      C *** IF(NUMV.LT.1) GO TO 50
      IF(CT.NE.CTEST) GO TO 21
      ROLD=YPT(NOLD)/COS(DUMOLD)
      RNEW=YPT(N)/COS(DUM)
      DR=(RNEW-ROLD)/NUM
      IF(IDOF(I).EQ.0.AND.IDOLD(I).EQ.0) NEQ=NEQ+NUMN
20   CONTINUE
      DX=(XPT(N)-XPT(NOLD))/NUM
      IF(CT.NE.CTEST) GO TO 21
      ROLD=YPT(NOLD)/COS(DUMOLD)
      RNEW=YPT(N)/COS(DUM)
      DR=(RNEW-ROLD)/NUM

```

ADNA0190  
ADNA0200  
ADNA0210  
ADNA0220  
ADNA0230  
ADNA0240  
ADNA0250  
ADNA0260  
ADNA0270  
ADNA0280  
ADNA0290  
ADNA0300  
ADNA0310  
ADNA0320  
ADNA0330  
ADNA0340  
ADNA0350  
ADNA0360  
ADNA0370  
ADNA0380  
ADNA0390  
ADNA0400  
ADNA0410  
ADNA0420  
ADNA0430  
ADNA0440  
ADNA0450  
ADNA0460  
ADNA0470  
ADNA0480  
ADNA0490  
ADNA0500  
ADNA0510  
ADNA0520  
ADNA0530  
ADNA0540  
ADNA0550  
ADNA0560  
ADNA0570  
ADNA0580  
ADNA0590  
ADNA0600  
ADNA0610  
ADNA0620  
ADNA0630  
ADNA0640  
ADNA0650  
ADNA0660

```

DT=(DUM-DUMOLD)/NUM
GO TO 22
21 CONTINUE
DY=(YPT(N)-YPT(NOLD))/NUM
DZ=(ZPT(N)-ZPT(NOLD))/NUM
22 CONTINUE
K=NOLD
DO 30 J=1,NUMN
  KK=K
  K=K+KNOLD
  XPT(K)=XPT(KK)+DX
  IF(CT-NE-CTEST) GO TO 26
  ROLD=ROLD+DR
  DUMOLD=DUMOLD+DT
  YPT(K)=ROLD*COS(DUMOLD)
  ZPT(K)=ROLD*SIN(DUMOLD)
  GO TO 28
26 CONTINUE
  YPT(K)=YPT(KK)+DY
  ZPT(K)=ZPT(KK)+DZ
  CONTINUE
  NUMPT(K)=K
  CONTINUE
30 NOLD=N
  KNOLD=KN
  DUMOLD=DUM
  TO COUNT DOFS TO DETERMINE NUMBER OF IC CARDS
  DO 55 I=1,6
    IF(IDOF(I).EQ.0.AND.ID(I).EQ.0) NEQ=NEQ+1
    IDOLD(I)=ID(I)
  CONTINUE
55 CONTINUE
  IF(N-NE-NUMNP) GO TO 10
  READ LOAD CONTROL CARDS
  READ(5,9000) DJMNY
  DO 80 I=1,IMASSN
    IF(IMASSN.EQ.0) GO TO 81
    READ(5,9000) DUMMY
  CONTINUE
80 CONTINUE
81 CONTINUE
  IF(IDAMPN.EQ.0) GO TO 91
  DO 90 I=1,IDAMPN
    READ(5,9000) DUMMY
  CONTINUE
90 CONTINUE
91 CONTINUE
  READ INITIAL CONDITIONS
  READ(5,9002) ICON
  IF(ICON.EQ.0) GO TO 100
  CARDNR=NEQ/6.0

```

ADNA0670  
ADNA0680  
ADNA0690  
ADNA0700  
ADNA0710  
ADNA0720  
ADNA0730  
ADNA0740  
ADNA0750  
ADNA0760  
ADNA0770  
ADNA0780  
ADNA0790  
ADNA0800  
ADNA0810  
ADNA0820  
ADNA0830  
ADNA0840  
ADNA0850  
ADNA0860  
ADNA0870  
ADNA0880  
ADNA0890  
ADNA0900  
ADNA0910  
ADNA0920  
ADNA0930  
ADNA0940  
ADNA0950  
ADNA0960  
ADNA0970  
ADNA0980  
ADNA0990  
ADNA1000  
ADNA1010  
ADNA1020  
ADNA1030  
ADNA1040  
ADNA1050  
ADNA1060  
ADNA1070  
ADNA1080  
ADNA1090  
ADNA1100  
ADNA1110  
ADNA1120  
ADNA1130  
ADNA1140







ADNA2110  
ADNA2120  
ADNA2130  
ADNA2140  
ADNA2150  
ADNA2160  
ADNA2170  
ADNA2180  
ADNA2190  
ADNA2200  
ADNA2210  
ADNA2220  
ADNA2230  
ADNA2240  
ADNA2250  
ADNA2260  
ADNA2270  
ADNA2280  
ADNA2290  
ADNA2300  
ADNA2310  
ADNA2320  
ADNA2330  
ADNA2340  
ADNA2350  
ADNA2360  
ADNA2370  
ADNA2380  
ADNA2390  
ADNA2400  
ADNA2410  
ADNA2420  
ADNA2430  
ADNA2440  
ADNA2450  
ADNA2460  
ADNA2470  
ADNA2480  
ADNA2490  
ADNA2500  
ADNA2510  
ADNA2520  
ADNA2530  
ADNA2540  
ADNA2550  
ADNA2560  
ADNA2570  
ADNA2580

```

C ***
NUMMAT=NPARG(16)
NSTRES=NPARG(13)
CALCULATE THE NUMBER OF MATERIAL CASE CARDS
IF(NPARG(15).EQ.1) NCARD=1
IF(NPARG(15).EQ.2) NCARD=2
IF(NPARG(15).EQ.3) NCARD=3
IF(NPARG(15).EQ.4) NCARD=4
IF(NPARG(15).EQ.5) NCARD=5
IF(NPARG(15).EQ.6) NCARD=6
IF(NPARG(15).EQ.7) NCARD=7
IF(NPARG(15).EQ.8) NCARD=8
IF(NPARG(15).EQ.9) NCARD=9
IF(NPARG(15).EQ.10) NCARD=10
IF(NPARG(15).EQ.11) NCARD=11
IF(NPARG(15).EQ.12) NCARD=12
IF(NPARG(15).EQ.13) NCARD=13
IF(NPARG(15).EQ.14) NCARD=14
GO TO 20
CARDNR=NPARG(17)/8.0
NCARD=INT(CARDNR)
TEST=CARDNR-NCARD
IF(TEST.GT.0.1) NCARD=NCARD+1
20 CONTINUE
C ***
N12=12
READ MATERIAL PROPERTIES
DO 50 J=1,NUMMAT
READ(5,9000) DUMMY
9000 FORMAT(20A4)
DO 45 I=1,NCARD
READ(5,9000) DUMMY
45 CONTINUE
C ***
CONTINUE OUTPUT TABLE CARDS
READ NSTRES NPARG(13).EQ.0) GO TO 61
DO 60 I=1,NSTRES
READ(5,9000) DUMMY
60 CONTINUE
C ***
CONTINUE
READ AND GENERATE ELEMENT DATA CARDS
IF(NPARG(14).EQ.0) NPARG(14)=1
NEL=NPARG(14)
READ(5,9002) I,NEL,IINC
130 IF(IINC.EQ.0) IINC=1
9002 FORMAT(15,15X,15)
9004 READ(5,9004) (INP(I),I=1,8)
140 FORMAT(8I5)
NEL=NEL+1
140 ML=INEL-NEL
IF(ML.LT.150) GO TO 155,160
150 CALL ERROR(8)
C *** NO GENERATION OF NODE POINTS REQUIRED

```



```

20 CONTINUE
N20=20
C *** READ MATERIAL PROPERTIES
DO 50 J=1,NUMMAT
  READ(5,9000) DUMMY
9000 FORMAT(20A4)
  DO 45 I=1,NCARD
    READ(5,9000) DUMMY
  CONTINUE
45 CONTINUE
50 READ STRESS OUTPUT TABLE CARDS
C *** IF(NPAR(13).EQ.0) GO TO 61
DO 60 I=1,NSTRES
  READ(5,9000) DUMMY
60 CONTINUE
61 IF(NPAR(14).EQ.0) NPAR(14)=1
  NEL=NPAR(14)-1
130 READ(5,9002) IVEL,IINC
9002 FORMAT(15,30X,15)
  IF(IINC.EQ.0) IINC=1
  READ(5,9004) (INP(I),I=1,8)
9004 READ(5,9004) (INP(I),I=9,N20)
140 FORMAT(12I5)
  NEL=NEL+1
  ML=INEL-NEL
  IF(ML) 150,155,160
150 CALL ERROR(9)
C *** NO GENERATION OF NODE POINTS REQUIRED
155 DO 156 I=1,N20
  NP(I)=INP(I)
156 CONTINUE
  GO TO 162
C *** GENERATION OF NODE POINTS REQUIRED
160 DO 161 I=1,N20
  IF(NP(I).EQ.0) GO TO 161
  NP(I)=NP(I)+KN
161 CONTINUE
162 NUMEL=NUMEL+1
  WRITE(10) N20,NEL,(NP(I),I=1,N20)
  IF(NEL.EQ.NPAR(2)) RETURN
  IF(NEL.LT.INEL) GO TO 140
  KN=IINC
  GO TO 130
END
SUBROUTINE ADSEAM

```

```

ADNA3070
ADNA3080
ADNA3090
ADNA3100
ADNA3110
ADNA3120
ADNA3130
ADNA3140
ADNA3150
ADNA3160
ADNA3170
ADNA3180
ADNA3190
ADNA3200
ADNA3210
ADNA3220
ADNA3230
ADNA3240
ADNA3250
ADNA3260
ADNA3270
ADNA3280
ADNA3290
ADNA3300
ADNA3310
ADNA3320
ADNA3330
ADNA3340
ADNA3350
ADNA3360
ADNA3370
ADNA3380
ADNA3390
ADNA3400
ADNA3410
ADNA3420
ADNA3430
ADNA3440
ADNA3450
ADNA3460
ADNA3470
ADNA3480
ADNA3490
ADNA3500
ADNA3510
ADNA3520
ADNA3530
ADNA3540

```



ADNA4030  
ADNA4040  
ADNA4050  
ADNA4060  
ADNA4070  
ADNA4080  
ADNA4090  
ADNA4100  
ADNA4110

162 J=J+KN  
CONTINUE  
NUMEL=NUMEL+1  
WRITE(101,N2,VEL(I,J) RETURN  
IF(VEL.EQ.NPAR(2))  
IF(VEL.LT.INEL) GO TO 140  
KN=IINC  
GO TO 130  
END



```
C ***** USER SUPPLIED DISPLACEMENT DATA SUBROUTINE CALLED BY PSAPI*****APRI981
C      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
C      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
C      RETURN                                     *      *      *      *      *      *      *      *      *
C      END                                         *      *      *      *      *      *      *      *      *
C
C      SUBROUTINE DATA5(NUMPT,XPT,YPT,ZPT,UPT,VPT,WPT)
C
C      ***SUBROUTINE TO READ EACH NODE'S DISPLACEMENT FROM DEVICE 58
C      *** CALLED BY PSAPI
C
C      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
C      REAL*8 D(6)
C      COMMON/CONTRL/ KGEOM,KDATA,KPLOT,KSYMXY,KSYMxz,KSYMZY,NOTAT,XLHT,
C      IKHORZ,KVERT,PHI,THETA,PSI,NEWFR,I$CALE,PLOTSZ,XORGN,YORGN,
C      2PSCALE,KDISP,DHAG,KODE
C      COMMON/KOUNT/ NNODE,NNDIST,NUDISP,NVDISP,NWDISP
C      COMMON/VALUES/ NVALUS
C      DIMENSION NUMPT(1),XPT(1),YPT(1),ZPT(1),UPT(1),VPT(1),WPT(1)
C      REWIND 58
C      WRITE(6,10)NVALUS
C      FORMAT('///5X,NUMBER OF DISPLACEMENT SETS TO BE PLOTTED =',I6//')
C      DO 20 I=1,NVALUS
C      READ(58)NODE,{D(I),I=1,6}
C      R1=SNGL(D{1})
C      R2=SNGL(D{2})
C      R3=SNGL(D{3})
C      UPT(II)=R1
C      VPT(II)=R2
C      WPT(II)=R3
C      CONTINUE
C      RETURN
C      END
C
C      SUBROUTINE DATA9(NUMPT,XPT,YPT,ZPT,UPT,VPT,WPT,DISPD,NON)
C
C      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
C      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
C      *** USER SUPPLIED DISPLACEMENT INPUT SUBROUTINE.
C      *** CALLED BY PSAPI
C
C      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *      *
C      COMMON/CDATA/NTIME,NTLC
C      COMMON/CONTRL/ KGEOM,KDATA,KPLOT,KSYMXY,KSYMxz,KSYMZY,NOTAT,XLHT,
C      IKHORZ,KVERT,PHI,THETA,PSI,NEWFR,I$CALE,PLOTSZ,XORGN,YORGN,
C      2PSCALE,KDISP,DHAG,KODE
C
C      10
C
C      20
```





```

END
SUBROUTINE CALPLT(X,Y,IPEN)
* * * * *
***** TO ADAPT FOR NPS VERSAPLOT CALLED BY PSAPI/PLOTX/GARROW/ERROR
BASIC PLOTTING CALL
* * * * *
INTEGER IPEN
REAL X,Y
CALL PLOT(Y,-X,IPEN)
RETURN
END
SUBROUTINE NOTATE (X,Y,HT,BCD,TH,N)
* * * * *
***** TO ADAPT FOR NPS VERSAPLOT CALLED BY PSAPI
FOR LETTERING ON PLOT
* * * * *
INTEGER N
REAL X,Y,HT,TH,THR
DIMENSION BCD(1)
THR=TH+270.0
CALL SYMBOL(Y,-X,HT,BCD,THR,N)
RETURN
END
SUBROUTINE CALNUM (X,Y,HT,FPN,PH,DEC)
* * * * *
***** TO ADAPT FOR NPS VERSAPLOT CALLED BY PLOTX
FOR NUMBERING ON PLOT
* * * * *
INTEGER DEC
REAL X,Y,HT,PH,FPN,PHR
PHR=PH+270.0
CALL NUMBER(Y,-X,HT,FPN,PHR,DEC)
RETURN
END
SUBROUTINE CALWH(X,Y)

```

```

ADPT0070
ADPT0080
APR1981
APR1981
APR1981
ADPT0090
APR1981
APR1981
*APR1981
APR1981
ADPT0100
APR1981
APR1981
ADPT0150
ADPT0160
ADPT0170
APR1981
*APR1981
APR1981
ADPT0180
APR1981
APR1981
APR1981
APR1981
ADPT0190
APR1981
ADPT0210
APR1981
APR1981
ADPT0260
ADPT0270
ADPT0280
APR1981
*APR1981
APR1981
ADPT0290
APR1981
APR1981
*APR1981
APR1981
ADPT0300
APR1981
APR1981
APR1981
ADPT0360
ADPT0370
ADPT0380
APR1981

```



## LIST OF REFERENCES

1. Stenstrom, F.E., Photoelastic Study of Elastic and Plastic Stress Fields in the Vicinity of a Notch, Master's Thesis, Naval Postgraduate School, Monterey, California, December 1980.
2. Engle, E.C., An Investigation of Residual Stresses in Simulated Wing Panels of 7075-T6 Aluminum, Master's Thesis, Naval Postgraduate School, Monterey, California, December 1979.
3. Stuart, G.L., An Investigation of Residual Stress Characterization of 7075-T6 Aluminum for Application in Fatigue Analysis, Master's Thesis, Naval Postgraduate School, Monterey, California, December 1978.
4. Massachusetts Institute of Technology Report 82448-1, A Finite Element Program for Automatic Dynamic Incremental Nonlinear Analysis (ADINA), by K. Bathe, September 1975 (revised May 1976).
5. Crandall, S.H., Engineering Analysis - A Survey of Numerical Procedures, pp. 171-173, McGraw-Hill, 1956.
6. Heise, U., "Combined Application of Finite Element Methods and Richardson Extrapolation to the Torsion Problem," Conference on the Mathematics of Finite Elements and Applications, ed. by J. R. Whiteman, V. 1., pp. 225-237, Academic Press, 1973.
7. Kibler, A.E., A Finite Element Preprocessor for SAPIV and ADINA, Master's Thesis, Naval Postgraduate School, Monterey, California, September 1977.
8. Howland, R.C.J., "Stress in a Strip of Finite Width, Which is Weakened by a Circular Hole," Philosophical Transactions of the Royal Society, Series A, V. 229, pp. 49-86, London, 1930.
9. ANSI/ASTM B557-79, Standard Methods of Tension Testing Wrought and Cast Aluminum and Magnesium Alloy Products, 1979.
10. Micro-Measurements, Transverse Sensitivity Errors TN-137, undated, Vishay Intertechnology, Inc.

11. Rivello, R.M., Theory and Analysis of Flight Structures, pp. 38-40, McGraw-Hill, 1969.
12. Garske, J.C., An Investigation of Methods for Determining Notch Root Stress from Far-Field Strain in Notched Flat Plates, Master's Thesis, Naval Postgraduate School, Monterey, California, September, 1977.
13. National Aeronautics and Space Administration (NASA) Report CR-1649, Finite Element Analysis of Structures in the Plastic Range, by H. Armen, Jr., A. Pifko, and H.S. Levine, February, 1971.
14. Segerlind, L.J., Applied Finite Element Analysis, p. 251, Wiley, 1976.
15. Zienkiewicz, O.C., The Finite Element Method, 3d Ed., McGraw-Hill Ltd, 1977.
16. David W. Taylor Naval Ship Research and Development Center Report, Accuracy Loss in Distorted Isoparametric Finite Elements, by D. A. Hopkins and L.N. Gifford, September 1978.
17. Cook, R.D., Concepts and Applications of Finite Element Analysis, pp. 87-92, Wiley, 1974.
18. Barlow, J., "Optimal Stress Locations in Finite Element Models," International Journal for Numerical Methods in Engineering, V. 10, pp. 243-251, 1976.
19. Hinton, E., and Campbell, J.S., "Local and Global Smoothing of Discontinuous Finite Element Functions Using a Least Squares Method," International Journal for Numerical Methods in Engineering, V. 8, pp. 461-480, 1974.
20. Peterson, R.E., Stress Concentration Factors, pp. 20-36 and 111-150, Wiley, 1974.
21. Frocht, M.M., "Photoelastic Studies in Stress Concentration," Mechanical Engineering, V. 58, August 1936.
22. Frocht, M.M., "Factors of Stress Concentration in Bars with Deep Sharp Grooves and Fillets in Tension," Proceedings of the Society of Experimental-Stress Analysis, V. 8, No. 2, 1951.
23. Naval Research Laboratory Report 7278, Finite Element Analysis of Notched Tensile Specimens in Plane Stress, by C. A. Griffis, 10 September 1971.

24. Clough, R.W., "The Finite Element Method in Structural Mechanics," Stress Analysis ed. by O. Zienkiewicz and G. Holister, pp. 85-119, Wiley, 1965.
25. Frocht, M.M., and Thomson, R.A., "Further Work on Plane Elastoplastic Stress Distributions," Proceedings of the International Symposium on Photoplasticity, Pergamon Press, 1963.
26. Advisory Group for Aerospace Research and Development Report AGARD-MAN-8-70, Manual on Fatigue of Structures, by W. G. Barrois, pp. 206-227, June 1970.
27. Lubahn, J.D., and Felgar, R.P., Plasticity and Creep of Metals, p. 510, Wiley, 1961.
28. Easterling, L.R., Stress Analysis of Ceramic Turbine Blades by Finite Element Method - Part I, Master's Thesis, Naval Postgraduate School, Monterey, California, March 1978.
29. Preisel, J.H., Stress Analysis of Ceramic Gas Turbine Blades by the Finite Element Method - Part II, Master's Thesis, Naval Postgraduate School, Monterey, California, March 1978.

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